

型号认可证数据单  
THE VALIDATION DATA SHEET

编号/No: VTC0116A

版次/Revision: 7

型别/Type:

560XL

560

批准人/Approved By:



职务/Title: Director General

日期/Date: 2023-06-29

本数据单是型号认可证（编号 VTC0116A）的组成部分，它规定了符合中国民用航空局的适航要求所颁发此型号认可证的产品状态和限制。

This data sheet, which is a part of Validation Type Certificate (No: VTC0116A), prescribes conditions and limitations under which the product for which the Type Certificate was issued meets the airworthiness requirements of the Chinese Civil Aviation Regulation.

型号认可证持有人/ Validation of Type Certificate Holder:

德事隆航空公司，堪萨斯州威奇托塞斯纳大道 1 号。

Textron Aviation Inc. One Cessna Boulevard Wichita, Kansas 67215.

型号认可证持有人记录/Validation of Type Certificate Holder Record:

2018 年 7 月 30 日型号认可证持有人从 Cessna Aircraft Company 变更为 Textron Aviation Inc.

Cessna Aircraft Company transferred to Textron Aviation Inc. on July 30, 2018.

## 第 1 部分 概述

## SECTION 1 GENERAL

## 1. 型号认可证数据单版次信息以及修订记录/ Revision and Change Record of the Validation of Type Certificate Data Sheet

## 数据单版次信息/Revision of the Type Certificate Data Sheet

|               | 型号认可证数据单编号、版次和颁发日期<br>Validation of Type Certificate Data Sheet<br>Number, Revision and issuance date | 型号合格证数据单编号、版次和颁发日期<br>Type Certificate Data Sheet Number, Revision and<br>issuance date |
|---------------|-------------------------------------------------------------------------------------------------------|-----------------------------------------------------------------------------------------|
| 当前<br>Current | No.VTC0116A, 版次7/Rev 7<br>2023年6月29日<br>No. VTC0116A; Revision 7, Date of Issue:<br>2023-06-29        | 型别 560XL<br>对应 FAA 数据单:<br>FAA TCDS A22CE, Issue: 73<br>April 27, 2022                  |
| 历史<br>History | No.VTC0116A, 版次6/Rev 6<br>2023年03月21日<br>No. VTC0116A; Revision 6, Date of Issue:<br>2023-3-21        | FAA TCDS A22CE, Issue: 72<br>20 April 2021                                              |
| 历史<br>History | No.VTC0116A, 版次5/Rev 5<br>2021年1月11日<br>No. VTC0116A; Revision 5, Date of Issue:<br>2021-1-11         | FAA TCDS A22CE, Issue: 70<br>15 May 2019                                                |
| 历史<br>History | No.VTC0116A, 版次4/Rev 4<br>2018年7月30日<br>No. VTC0116A; Revision 4, Date of Issue:<br>2018-7-30         | FAA TCDS A22CE, Issue: 68<br>04 May 2018                                                |
| 历史<br>History | No.VTC0116A, 版次3/Rev 3<br>2014年3月27日<br>No. VTC0116A; Revision 3, Date of Issue:<br>2014-3-27         | FAA TCDS A22CE, Issue: 65<br>04 February 2011                                           |
| 历史<br>History | No.VTC0116A, 版次2/Rev 2<br>2011年8月29日<br>No. VTC0116A; Revision 2, Date of Issue:<br>2011-8-29         | FAA TCDS A22CE, Issue: 64<br>01 October 2008                                            |
| 历史<br>History | No.VTC0116A, 版次1/Rev 1<br>2004年11月8日<br>No. VTC0116A; Revision 1, Date of Issue:<br>2004-11-08        | FAA TCDS A22CE, Issue: 59<br>22 March 2004                                              |
| 历史<br>History | No.VTC-116A, 版次0/Rev 0<br>2001年5月28日<br>No. VTC-116A; Revision 0, Date of Issue:<br>2001-5-28         | FAA TCDS A22CE, Issue: 53<br>26 June 2000                                               |

注：上表中出口国局方型号合格证数据单信息仅供参考

Note: Above TCDS information of Certification Authority is only for reference

## 认可数据单修订记录/ Change Record of the VTCDS

| 版次/ Revision | 相对上一版的修订记录<br>Change Record                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                      | 备注<br>Remark |
|--------------|----------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------|--------------|
| 版次 7/Rev 7   | <p>修订第 3 部分第 2 节/ Revise Section 3 Part 2:</p> <p>1. 更新 XLS+审定基础, 增加以前遗漏的 FAR §§ 25.1457(a)(6).<br/>Update Certification Basis of XLS+ to add previously omitted FAR §§ 25.1457(a)(6).</p> <p>2.更新 XLS+审定基础, 针对 XLS+序列号 560-6318 和 560-6322 及以后审定基础为第 36 部第 1 至 31 号修正案 (第 5 阶段噪声要求)。<br/>Update Certification Basis of XLS+ to amend by Amendments 36-1 thru 36-31 (Stage 5) for S/N 560-6318 and 560-6322 and on.</p> <p>修订第 3 部分第 3 节/ Revise Section 3 Part 3:</p> <p>3. 增加了 XLS+序列号 560-6318 和 560-6322 及以后的重量信息。<br/>适应性调整其它各重量信息对应的序列号。<br/>Add weight info for XLS+ S/N 560-6318, S/N 560-6322 and on. Adjust the serial number corresponding to other weight infos.</p> |              |
| 版次 6/Rev 6   | <p>修订第 4 部分/Revise Section 4:</p> <p>1. 增加注释 15, 更换环保手提式灭火器。<br/>Add note 15 for usage hand fire extinguisher with Halotron (Non-Halon) agent.</p> <p>2. 更新注释 2, 将标牌参考文件更改为飞机装配图。<br/>Update note 2 to reference the airplane assembly drawing as the primary reference for required placards.</p>                                                                                                                                                                                                                                                                                                                                                                               |              |
| 版次 5/Rev 5   | <p>1. 更新 FAA 审定基础<br/>Update FAA Certification Basis</p> <p>2. 修订中华人民共和国国家标准“3 号喷气燃料”版次<br/>Revise the version of National Standard of the People's Republic of China “No.3 Jet Fuel”</p> <p>3. 针对 560XL (EXCEL) 序列号为 560-5002 的飞机, 取消发动机高压转子转速 N2 限制<br/>Remove N2 engine rotor operating speed limit for serial number 560-5002 of 560XL (EXCEL)</p> <p>4. 修订 560XL (XLS) 噪声修正案水平<br/>Revise 560XL (XLS) Amendment Level of Noise standard</p> <p>5. 注 6 文字勘误<br/>Correction on Note 6</p>                                                                                                                                                                                     |              |

2. 认可当局/Validation Authority:

中国民用航空局/CAAC

审定当局/Certifying Authority:

美国联邦航空管理局/FAA

3. 型号合格证持有人/Type Certificate Holder:

德事隆航空公司,  
堪萨斯州威奇托塞斯纳大道 1 号。

Textron Aviation Inc.

One Cessna Boulevard

Wichita, Kansas 67215

**型号合格证持有人记录/Type Certificate Holder Record:**

|                                                                        |                                      |
|------------------------------------------------------------------------|--------------------------------------|
| Cessna Aircraft Company<br>P.O. Box 7704<br>Wichita, Kansas 67277      | 至 2015- 07-29<br>Until July 29, 2015 |
| Textron Aviation Inc.<br>One Cessna Boulevard<br>Wichita, Kansas 67215 | 自 2015-07-29<br>Since July 29, 2015  |

**4. 生产制造商/Manufacturer**

德事隆航空公司，

堪萨斯州威奇托塞斯纳大道 1 号。

Textron Aviation Inc.

One Cessna Boulevard, Wichita, Kansas 67215

**生产制造商记录/Manufacturer Record:**

|                                                                        |                                                                        |
|------------------------------------------------------------------------|------------------------------------------------------------------------|
| Cessna Aircraft Company<br>P.O. Box 7704<br>Wichita, Kansas 67277      | 以下系列飞机是以塞斯纳飞机公司的名称生产的：<br>560XL: 560-5001 至 560-6191。                  |
| Textron Aviation Inc.<br>One Cessna Boulevard<br>Wichita, Kansas 67215 | 公司名称变更于 2015 年 7 月 29 日生效。以下系列产品以德事隆航空公司的名称生产：<br>560XL: 560-6192 及以后。 |

**5. 双发延程飞行/ETOPS: 不适用/Not applicable****6. CAAC 专用要求/Special Requirements:**

|                                                                                                                                                                                                         |
|---------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------|
| 交付到中国的每一架特定的航空器必须满足下述特殊要求：<br>Each of specific aircraft delivered to China shall meet the following Special Requirements:                                                                               |
| (1) 必须满足中国民用航空局的运行要求（例如标记标牌的中文要求）。<br>The operational requirements of CAAC must be met (e.g., Chinese language requirements for markings and placards).                                                 |
| (2) 燃油符合中华人民共和国国家标准“3 号喷气燃料”（GB 6537）的要求。<br>Fuel conforming to National Standard of the People's Republic of China “No.3 Jet Fuel” (GB 6537).                                                          |
| (3) 在每一航空器上必须安装快速存储器(QAR)或具有快速存取记录器(QAR)功能的其它设备（参照中国适航指令 CAD-97-MULT-38）。<br>Quick Access Recorder (QAR) or other equivalent equipment must be installed on each aircraft (reference to CAD97-MULT-38). |

## 第 2 部分 数据 (型别: 560)

## SECTION 2 DATA (Models: 560)

## 1. 产品型别和批准日期/Model and Approval Date

| 型别 MODEL     | 批准日期<br>Approval Date  | 类别<br>category                        |
|--------------|------------------------|---------------------------------------|
| 560(Encore+) | 2011年8月29日/29 Aug 2011 | 运输类航空器/Transport<br>Category Aircraft |

## 2. 认可审定基础和生产依据/ Certification Basis and Production Basis

CAAC的认可审定基础由FAA的审定基础 (详见FAA型号合格证数据单: 编号A22CE, Rev 73) 和 CAAC的附加技术条件组成。

The CAAC validation certification basis consists of the FAA certification basis (refer to FAA Type Certificate Data Sheet No. A22CE, Rev 73) and the CAAC Additional Technical Conditions (ATCs).

## FAA 审定基础//FAA certification basis

详见FAA型号合格证数据单A22CE (版本73) 中的560(Encore+)型别的审定基础/ Refer to FAA Type Certificate Data Sheet No. A22CE Revision 73.

- (A) FAR 25 部 (生效日期: 1965 年 2 月 1 日) 25-1 修正案号至 25-17 号修正案, 除了适用于发动机和 FADEC 的段落, 以及下列条款:

Part 25 of the Federal Aviation Regulations effective February 1, 1965, as amended by Amendments 25-1 through 25-17, except for paragraphs applicable for engines and FADEC and the following:

- (a) 附加条款/Addition:

FAR §§ 25.625 as amended by Amendments 25-1 through 25-17, and; § 25.1401 as amended by Amendments 25-1 thru 25-27, 25.787, 25.789, 25.791, 25.853, 25.855, 25.857, and 25.1359 as amended by Amendments 25-1 thru 25-32, § 25.1303(a)(2) as amended by Amendments 25-1 thru 25-38, 25.305 as amended by Amendments 25-1 thru 25-54, 25.119, 25.121, 25.125, 25.143, 25.145, 25.149, 25.201, 25.203, 25.251, 25.253, 25.337, 25.361, 25.363, 25.371, 25.471, 25.473, 25.479, 25.481, 25.483, 25.485, 25.489, 25.491, 25.493, 25.509, 25.611, 25.721, 25.723, 25.725, 25.727, 25.731, 25.733, 25.735, 25.863, 25.865, 25.867, 25.869, 25.901, 25.933, 25.934, 25.939, 25.943, 25.951, 25.952, 25.959, 25.961, 25.965, 25.977, 25.979, 25.994, 25.995, 25.997, 25.999, 25.1001, 25.1041, 25.1043, 25.1045, 25.1091, 25.1093, 25.1103, 25.1121, 25.1123, 25.1141, 25.1143, 25.1145, 25.1163, 25.1165, 25.1181, 25.1183, 25.1185, 25.1189, 25.1195, 25.1197, 25.1199; 25.1203, 25.1205(revoked), 25.1207, 25.1305, 25.1316, 25.1322, 25.1326, 25.1337, 25.1351, 25.1385(c), 25.1387, 25.1419, 25.1438, 25.1521, 25.1529, 25.1549, and 25.1551 as amended by Amendments 25-1 thru 25-91, 25.901, 25.903(a)(b)(c)(e)(f), 25.933, 25.934, 25.939, 25.943 as amended by Amendments 25-1 through 25-117.

- (b) 仅对于电子飞行仪表系统, 增加下列条款/Additions for the Electronic Flight Instrument Systems only:  
14 CFR §§ 25.1301, 25.1303(b), 25.1322 as amended by Amendments 25-1 thru 25-38; 25.1309, 25.1321(a)(b)(d)(e), 25.1331, 25.1333, and 25.1335 as amended by Amendments 25-1 thru 25-41.

- (c) 仅增加如下限制/Addition limited ONLY to:

- (i) 增压系统/Pressurization System; 电子控制和排水系统/Digital Controller and Outflow System
- (ii) 防滑系统/Anti-skid System; 独立机轮数字防滑控制器/Individual Wheel Digital Anti-skid Controller
- (iii) 防冰系统/Anti-ice System; 尾翼除冰套控制和监控及机翼外缘引气控制和故障显示/Electronic Tail Boot Control and Monitoring and Outboard Wing Leading Edge Bleed Air Control and Fault Annunciation
- (iv) 仅对于内部控制逻辑和显示功能的综合警告、警戒及咨询通知显示系统限制/Integrated Warning, Caution, and Advisory Annunciation System limited to the Internal Control Logic and Display Functions Only.

14 CFR 第 25.1309 条第 25-1 到 25-91 号修正案

14 CFR 25.1309 as amended by Amendments 25-1 thru 25-91.

- (d) 仅增加对于 FADEC 发动机控制系统的限制: 25.1309 条第 25-1 到 25-117 号修正案。  
Addition limited only to FADEC engine control system: 25.1309 as amended by Amendments 25-1 thru 25-117.
- (B) 14 CFR 第 36 部, 1969 年 12 月 1 日生效, 噪声标准, 36-1 号修正案。  
14 CFR Part 36 effective December 1, 1969, Noise Standards, as amended by Amendments 36-1 thru 36-21.
- (C) 14 CFR 第 34 部, 1990 年 9 月 10 日生效, 涡轮发动机燃油排泄和排气排出物要求, 第 34-1 到 34-3 号修正案。  
14 CFR Part 34 effective September 10, 1990, Fuel Venting and Exhaust Emission Requirements for Turbine Engine Powered Airplanes, as amended by Amendments 34-1 thru 34-3.
- (D) 以下的专用条件/Special Conditions as follows:
- (a) 25-ANM-21, 高高度 (45000 英尺) 运行附加要求  
25-ANM-21, additional requirements for High Altitude Operation (45,000 feet).
  - (b) 25-ANM-79, 仅针对高强度辐射场的附加要求  
25-ANM-79, additional requirements for High Intensity Radiated Fields (HIRF) only.
  - (c) 25-25-CE-4, 涡轮发动机附加要求 (第 3 段, 飞行中重新启动功能; 第 6 段, 涡轮发动机安装; 第 7 段, 发动机点火系统)  
25-25-CE-4, additional requirements for Turbine engine powerplant installation [Paragraphs 3, Inflight restart capability; 6, Turbine engine powerplant installation; and 7, Engine ignition system only].
- (E) 以下的等效安全/Equivalent levels of safety as follows:
- (a) 14 CFR 第 25.815 条, 过道宽度  
14 CFR 25.815, Passenger Cabin Aisle Width.
  - (b) 14 CFR 第 25.773(b)(2)条, 风挡玻璃清晰视野的使用  
14 CFR 25.773(b)(2), Use of clear vision area of windshield.
  - (c) 等效安全(ELOS)备忘录 # AT4267WI-T-P-1, 14 CFR 第 25.1549(a)到(c)数字显示仅显示 N2, 发动机燃油流量(Wf), 及备用 N1, N2 及 ITT  
ELOS Memo # AT4267WI-T-P-1, 14 CFR 25.1549(a) thru (c) digital only displays N2, engine fuel flow (Wf), and Standby N1, N2, and ITT.
  - (d) 14 CFR 第 25.813(e)条, 易破门  
14 CFR 25.813(e), Frangible door.
  - (e) 14 CFR 第 25.807(d)应急出口水上迫降  
14 CFR 25.807(d), Emergency exits ditching.
  - (f) 14 CFR 第 25.841(b)(6)条, 25-87 号修正案, 客舱增压-高高度起飞及着陆运行  
14 CFR 25.841 (b)(6), Cabin Pressurization – High Altitude Takeoff and Landing Operation, through Amendment 25-87.
  - (g) ELOS 备忘录 # ST4383WI-T-SE-1, 14CFR 第 25.1309(a)(9)条及 25.1547 电子备用方向指示器  
ELOS Memo # ST4383WI-T-SE-1, 14 CFR 25.1303(a)(9), 25.1547 electronic standby direction indicator.
- (F) 不符合 14 CFR 第 25.801 条水上迫降  
14 CFR 25.801 ditching not complied with.
- (G) 符合结冰保护已按照 14 CFR 第 25.1419 条进行了证明  
Compliance with ice protection has been demonstrated in accordance with 14 CFR 25.1419.

#### 中国民用航空局附加技术条件/CAAC Additional Technical Conditions

交付到中国的每一架 560 飞机须满足下述附加技术条件:

Each of Model 560 aircraft delivered to China shall meet the following Additional Technical Conditions:

- (1) 必须符合 FAA 颁布的所有适用的适航指令;  
All applicable airworthiness directives (ADs) issued by the FAA shall be complied with.
- (2) 燃油符合中华人民共和国国家标准 “3 号喷气燃料” (GB 6537) 的要求;  
Fuel conforming to National Standard of the People’s Republic of China “No.3 Jet Fuel” (GB 6537).
- (3) 根据 CCAR 91.435 应安装应急定位发射机;  
Per CCAR 91.435, Emergency Locator Transmitter shall be installed.

(4) 标牌要求/ Placard Requirements:

- (a) 根据 CCAR 21.29(5), 机上所有对旅客进行的提示、警告和通知的文字标记和标牌; 机上所有向旅客或机外营救人员指示应急出口和门的位置以及开启方法的文字和标牌; 旅客可能使用的机上所有应急设备的操作、使用说明应当至少有中文表述;  
Per CCAR 21.29(5), all on-board placard and marking for awareness, warning, and note to the passenger, all on-board placard and marking for opening method and location indication of the emergency exit and door to passenger and door to passenger and outside rescuer, and all operational instructions for on board emergency equipment possible operated by passengers shall be presented in Chinese at least.
- (b) 中文字符的尺寸需符合适用的 CCAR 25.811 条的要求;  
The Dimension of the Chinese characters shall comply with the provision of CCAR 25.811 as applicable.
- (c) 如有其他中文标牌和标记, 应当在交付前或者交付时由民航局接受或者批准。  
Other placard and marking in Chinese, if any, shall be accepted or approved by the CAAC before or at the moment of the delivery.

(5) 根据 CCAR91.433 安装飞行数据记录仪。

注: 如果飞机预期按照 CCAR 第 121 部运行, 则必须按照 CAD1997-MULT-38 安装快速存储器(QAR)。

Per CCAR91.433 Flight Data Recorder should be installed.

Note:

If Aircraft is intended to be operated under CCAR Part 121, A Quick Access Recorder (QAR) must be installed per CAD1997-MULT-38.

生产依据/ Production Basis:

FAA 1998 年 5 月 7 日颁发的生产许可证 No.PC-4 适用于 560 ( Encore+) 序列号 560-0751 到 560-5000。

Production Certificate No. PC-4 issued May 7, 1998. Applies to 560 (Encore+) S/N 560-0751 through 560-5000.

### 3. 技术特征和使用限制/Technical Characteristics and Operational Limitations

发动机/Engines:

2 台加拿大普惠公司型号为: PW535B 的涡轮风扇发动机 (发动机型号合格证编号 E-27, 型号认可证编号 VTC0101E, 版次 2)

Two Pratt & Whitney Canada Corporation PW535B Turbofans (Type Certificate No. E-27, Validation Type Certificate No. VTC0101E, Rev.2)

燃油/Fuel:

商用煤油 Jet A, Jet A-1, JP-5, JP-8, 3 号喷气燃料 (GB 6537)

Commercial Kerosene JET A, JET A-1, JP-5, JP-8, No.3 Jet Fuel (GB 6537)

发动机限制/Engine Limits:

标准日海平面推力/Static thrust, sea level, standard day

起飞 (5分钟, 全发正常工作) /

Takeoff (5min., Normal ALL Engines Operating)

3400磅/lbs

起飞 (10分钟, 单发不工作) /Takeoff (10min., One Engine Inoperative)

3400磅/lbs

最大连续/Maximum Continuous

3400磅/lbs

最大允许发动机转子转速/ Max. permissible engine rotor operating speed

N1 (风扇) /N1 (Fan)

100%

15850rpm

N2 (燃气发生器) /N2 (Gas Gen)

100%

33970rpm

最大允许的涡轮内部燃气温度/Max. permissible interturbine temperature

起飞/Takeoff

700°C

最大连续/Maximum Continuous

700°C

启动/Starting

740°C

瞬时 (20秒) /Transient (20 seconds)

740°C

## 空速限制/Airspeed Limits:

|                                                               |                                                    |
|---------------------------------------------------------------|----------------------------------------------------|
| V <sub>MO</sub> (最大使用) /V <sub>MO</sub> (Maximum Operating)   |                                                    |
| 海平面到8000英尺/Sea Level to 8000 ft.                              | 260KCAS (262KIAS)                                  |
| 8000英尺到28907英尺/ 8000 to 28907 ft.                             | 290KCAS (292KIAS)                                  |
| M <sub>MO</sub> (28907英尺以上) /M <sub>MO</sub> Above 28907 ft   | 0.75马赫(0.755MIAS)                                  |
| V <sub>A</sub> (海平面) /V <sub>A</sub> (Sea Level)              |                                                    |
| 16830 lb                                                      | 193KCAS (194KIAS)                                  |
| 随重量和高度的变化见AFM/See AFM for variation with weight and altitude. |                                                    |
| V <sub>FE</sub> (襟翼展开/Flaps extended)                         |                                                    |
| 35° (着陆/Landing)                                              | 173KCAS(173KIAS)                                   |
| 15° (起飞和进近/Takeoff and approach)                              | 199KCAS(200KIAS)                                   |
| 7° (起飞/Takeoff)                                               | 199KCAS(200KIAS)                                   |
| V <sub>MCA</sub> (空中最小操纵速度/Minimum control speed Air)         | 84KCAS(86KIAS)                                     |
| V <sub>MCG</sub> (地面最小操纵速度/Minimum control speed Ground)      |                                                    |
| 15° (起飞) / 15° Takeoff                                        | 95KCAS(92KIAS)                                     |
| 7° (起飞) /7° Takeoff)                                          | 99KCAS(99KIAS)                                     |
| V <sub>LO</sub> (起落架放下操作/Landing gear operating extend)       | 249KCAS(250KIAS)                                   |
| V <sub>LO</sub> (起落架收起操作/Landing gear operating retract)      | 199KCAS(200KIAS)                                   |
| V <sub>LE</sub> (起落架放下/ Landing gear extend)                  | 249KCAS(250KIAS)                                   |
| V <sub>SB</sub> (减速板打开/Speed brakes extended)                 | 任何速度(无论襟翼放下或收起)<br>Any speed with or without flaps |

## 轮胎/Tire

|                              |            |
|------------------------------|------------|
| 最大地面速度/Maximum ground speed: | 165节/knots |
|------------------------------|------------|

## 重心范围 (起落架放下) / C.G. Range (Landing Gear Extended)

## 重心前限/Forward Limits

从17030磅时的基准线后方299.43英寸 (22.07% MAC) 线性变化到12400磅时的基准线后方296.14英寸(18.0% MAC); 等于或小于12400磅时, 在基准线后方296.14英寸(18.0% MAC)。  
Linear variation from 299.43 in. aft of datum (22.07% MAC) at 17,030 lb. to 296.14 in. aft of datum (18.0% MAC) at 12,400 lb.; 296.14 in. aft of datum (18.0% MAC) at 12,400lb. or less.

## 重心后限/Aft Limits

等于或小于17030磅时, 在基准线后方304.23英寸 (28.0% MAC)。  
304.23 in. aft of datum (28.0 % MAC) at 17,030lb. or less.

## 空机重心范围/ Empty Wt. C.G. Range:

无/None

## 基准/Datum

前增压舱前端面之前 94 英寸  
94.0 in. forward of the front face of the forward pressure bulkhead.

## 平均空气动力弦/MAC

80.98 英寸 (L.E. of MAC Sta.+281.56)  
注意: 这是对没有前缘翻边和小翼的基本翼型的参考。  
Note: This is reference MAC for basic wing without leading edge cuff and tip.

## 水平测量方法/Leveling Means

机组座椅导轨 (横向水平); 客舱门台阶铰链支架 (纵向水平)。/  
Crew Seat Rails (Lateral level); Cabin Door Step Hinge Brackets (Longitudinal level).

## 最大重量/Maximum

|               |            |
|---------------|------------|
| 起飞/Takeoff    | 16830磅/lb. |
| 着陆/Landing    | 15200磅/lb. |
| 零燃油/Zero fuel | 12850磅/lb. |
| 停机坪/Ramp      | 17030磅/lb. |

## 最小机组/Minimum Crew

所有飞行-两人（驾驶和副驾驶）/ For all flights: 2 persons (pilot and co-pilot)

## 座位数/No. of Seats

最多 10 人(2 名驾驶员，8 名乘客) /Up to 10 (2 Pilots, up to 8 Passengers)

## 最大行李/Maximum Baggage

|                      |                                                     |
|----------------------|-----------------------------------------------------|
| 前货舱/Nose Compartment | 310磅/lb.(at Sta.+74.0)                              |
| 后客舱/Aft cabin        | 600磅/lb.(at Sta.+348.0)                             |
| 尾锥/Tailcone          | 300磅/lb (at Sta.+434.0)和<br>200磅/lb (at Sta.+462.0) |

## 燃油量（加仑）/Fuel Capacity (Gal.)

两个大翼油箱/ Two wing tanks

每个总量为406.4; 可用量为403  
Total 406.4 each; usable 403 each  
+303.5英寸/in.

ARM

不可用燃油数据见注1/See NOTE 1 for data on unusable fuel

## 滑油容量（夸脱）/Oil Capacity (Quarts)

两个装于发动机的滑油箱挂箱/Two engine mounted tanks

每个总量为8.6; 不可用量为0.6  
Total 8.6 each; usable 0.6 each  
+387.0英寸/in.

ARM

## 风挡玻璃防冰液/Windshield Anti-Ice

容量/Capacity:

2.0夸脱/quarts

ARM

+91.5英寸/in.

## 液体/Fluid

批准的防冰液: TT-I-735 异丙基醇/Approved Anti-Ice Fluids: TT-I-735 Isopropyl alcohol

## 最大运行高度/Maximum Operating

45000 英尺/ft.

## 操纵面运动/Control Surfaces Movements

|                           |          |                  |         |                 |
|---------------------------|----------|------------------|---------|-----------------|
| 升降舵/ Elevator             | 向上/Up    | 18°+1°, -1/2°    | 向下/Down | 15°±1°          |
| 升降舵调整片/ Elevator trim tab | 向上/Up    | 4 1/2°+0°, -1/2° | 向下/Down | 11°+/-1/2°, -0° |
| 方向舵/Rudder                | 向右/Right | 22°+1°, -0°      | 向左/Left | 22°+1°, -0°     |
| 方向舵调整片/ Rudder trim tab   | 向右/Right | 10°±1°           | 向左/Left | 10°±1°          |
| 副翼/Aileron                | 向上/Up    | 19°±1°           | 向下/Down | 15°±1°          |
| 副翼调整片/ Aileron trim tab   | 向上/Up    | 20°±1°           | 向下/Down | 20°±1°          |
| 机翼襟翼/Wing flap            | 向下/Down  | 0°到35°±1°        |         |                 |
| 减速板/Speed brake           | 向上/Upper | 0°到58°+2°, -0°   |         |                 |

调整指南见飞机维修手册/See Airplane Maintenance Manual for rigging instructions.

## 制造商序列号/Manufacturer Serial No. Eligible:

560-0751 到 560-5000 (Encore +)。/560-0751 through 560-5000 (Encore +).

## 型号合格证信息/Type Certificate Information

1968 年 7 月 16 日申请型号合格证。

1971 年 9 月 9 日颁发型号合格证 A22CE。

Application for Type Certificate dated July 16, 1968.

Type Certificate No. A22CE issued September 9, 1971.

## 设备/ Equipment:

适用的适航规章里所要求的基本设备（见审定基础）必须安装在飞机上。

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

## 第 3 部分 数据 (型别: 560XL)

## SECTION 3 DATA (Models: 560XL)

## 1. 产品型别和批准日期/Model and Approval Date

| 型别 MODEL      | 批准日期<br>Approval Date  | 类别<br>category                     |
|---------------|------------------------|------------------------------------|
| 560XL (Excel) | 2004年11月8日/08 Nov 2004 | 运输类航空器/Transport Category Aircraft |
| 560 XL-(XLS)  | 2004年11月8日/08 Nov 2004 | 运输类航空器/Transport Category Aircraft |
| 560 XL-(XLS+) | 2014年3月27日/27 Mar 2014 | 运输类航空器/Transport Category Aircraft |

## 2. 认可审定基础和生产依据/ Certification Basis and Production Basis

CAAC的认可审定基础由FAA的审定基础 (详见FAA型号合格证数据单: 编号A22CE, Rev 73) 和 CAAC的附加技术条件组成。

The CAAC validation certification basis consists of the FAA certification basis (refer to FAA Type Certificate Data Sheet No. A22CE, Rev 73) and the CAAC Additional Technical Conditions (ATCs).

## FAA 审定基础//FAA certification basis

详见FAA型号合格证数据单A22CE (版本73) 中的560XL型别的审定基础/ Refer to FAA Type Certificate Data Sheet No. A22CE Revision 73.

序列号 560-5001 到 560-5500/ S/N 560-5001 through 560-5500 (EXCEL)

序列号 560-5501 到 560-6000/ S/N 560-5501 through 560-6000 (XLS)

- (A) FAR 25 部 (生效日期:1965 年 2 月 1 日) 修正案 25-1 至修正案 25-82, 以及下列附加条款及豁免:  
Part 25 of the Federal Aviation Regulations effective February 1, 1965, as amended by Amendments 25-1 through 25-82, with additions and exceptions as follows:

## (a) 附加条款/Addition:

FAR §§ 25.305, 25.321, 25.331, 25.333, 25.335, 25.341, 25.343, 25.345, 25.349, 25.371, 25.373, 25.391, 25.427, 25.1517 为修正案 25-86; 25.351 为修正案 25-91.

FAR §§ 25.305, 25.321, 25.331, 25.333, 25.335, 25.341, 25.343, 25.345, 25.349, 25.371, 25.373, 25.391, 25.427, 25.1517 as amended by Amendment 25-86; 25.351 as amended by Amendment 25-91.

## (b) 豁免 (如下表所示) / Exceptions (as shown in table):

| 条款号    | 标题       | 有效修正案    | 例外<br>[非审定基础的一部分]   |
|--------|----------|----------|---------------------|
| 25.562 | 应急着陆动态条件 | 25-82 适用 | 25.562(c)(5)和(c)(6) |

|         |                                                                                                                                                           |                                  |                                   |
|---------|-----------------------------------------------------------------------------------------------------------------------------------------------------------|----------------------------------|-----------------------------------|
| 25.571  | 结构损伤容限和疲劳评定                                                                                                                                               | 25-82 适用                         | 25.571(e)(1)                      |
| 25.631  | 鸟击损伤                                                                                                                                                      | 无, 此条款并非审定基础的一部分                 | 25.631 不适用                        |
| 25.671  | 操作系统-总体 <ul style="list-style-type: none"> <li>适用于 2 位置的水平安定面。</li> <li>所有其他飞机操纵系统。</li> </ul>                                                            | 25-82<br><br>原版适用(25-1 至 25-17)  | 无<br><br>25.671 修正案 25-23 及其以后不适用 |
| 25.677  | 配平系统 <ul style="list-style-type: none"> <li>适用于 2 位置的水平安定面。</li> <li>适用于其他飞机配平系统, 包括升降舵配平。</li> </ul>                                                     | 25-82<br><br>原版适用 (25-1 至 25-17) | 无<br><br>25.677 修正案 25-23 及以后不适用  |
| 25.1309 | 设备、系统和安装 <ul style="list-style-type: none"> <li>只适用于电子飞行仪表系统 (Honeywell Primus 1000 驾驶舱显示), 飞行指引, 液压, 电气, 增压系统, 和 2-位置水平安定面。</li> <li>所有其他飞机系统</li> </ul> | 25-82<br><br>原版适用 (25-1 至 25-17) | 无<br><br>25.1309 修正案 25-23 及以后不适用 |

| SECTION NO. | TITLE                                                                                                                                                                                     | EFFECTIVE AMENDMENT                                         | EXCEPTIONS [Not Part of Cert. Basis]                                      |
|-------------|-------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------|-------------------------------------------------------------|---------------------------------------------------------------------------|
| 25.562      | Emergency landing dynamic conditions.                                                                                                                                                     | 25-82 Applicable                                            | §25.562(c)(5)(6)                                                          |
| 25.571      | Damage-tolerance and fatigue evaluation of structure.                                                                                                                                     | 25-82 Applicable                                            | §25.571(e)(1)                                                             |
| 25.631      | Bird strike damage                                                                                                                                                                        | None, this section is not part of cert. basis.              | §25.631 not applicable                                                    |
| 25.671      | Control Systems – General. <ul style="list-style-type: none"> <li>Applicable to the 2-position horizontal stabilizer</li> <li>All other airplane control systems.</li> </ul>              | 25-82<br><br>Original Issue Applicable (25-1 through 25-17) | None<br><br>§ 25.671 as amended by Amdts. 25-23 and later, not applicable |
| 25.677      | Trim Systems <ul style="list-style-type: none"> <li>Applicable to the 2-position horizontal stabilizer.</li> <li>All other airplane trim systems, including the elevator trim.</li> </ul> | 25-82<br><br>Original Issue Applicable (25-1 through 25-17) | None<br><br>§25.677 as amended by Amdts. 25-23 and later, not applicable  |
| 25.1309     | Equipment, systems, and installations. <ul style="list-style-type: none"> <li>Applicable to Electronic Flight</li> </ul>                                                                  |                                                             |                                                                           |

|  |                                                                                                                                                                       |                                                      |                                                                 |
|--|-----------------------------------------------------------------------------------------------------------------------------------------------------------------------|------------------------------------------------------|-----------------------------------------------------------------|
|  | Instrument systems (Honeywell Primus 1000 Cockpit Display), Flight Guidance, hydraulic, electrical, pressurization system, and 2-position horizontal stabilizer only. | 25-82                                                | None                                                            |
|  | <ul style="list-style-type: none"> <li>All other airplane systems.</li> </ul>                                                                                         | Original Issue<br>Applicable<br>(25-1 through 25-17) | § 25.1309 as amended by Amdts. 25-23 and later, not applicable. |

(B) 对于 EXCEL: FAR 第 36 部, 1969 年 12 月 1 日生效, 噪声标准, 36-1 至 36-21 修正案。

对于 XLS: FAR 第 36 部, 1969 年 12 月 1 日生效, 噪声标准, 36-1 至 36-28 修正案。

For EXCEL: FAR Part 36 effective December 1, 1969, as amended by Amendments 36-1 through 36-21.

For XLS: FAR Part 36 effective December 1, 1969, as amended by Amendments 36-1 through 36-28.

(C) FAR 第 34 部, 1990 年 9 月 10 日生效, 第 34-1 修正案。

FAR Part 34 effective September 10, 1990 as amended by Amendments 34-1.

(D) 以下的专用条件/Special Conditions as follows:

(a) 25-ANM-79, 高强度辐射场的影响。与系统闪电防护有关的部分不适用, 以及  
25-ANM-79, effects for High Intensity Radiated Fields (HIRF). The portions associated with System Lightning Protection do not apply and:

(b) 25-ANM-21, 高高度运行(45000 英尺), 见注 7。  
25-ANM-21, High Altitude Operation (45,000 feet). See Note 7.

(E) 豁免/Exemption:

(a) 豁免号 6706 被授予。560XL 型 对 FAR 25.677(b)条有关水平安定面位置指示器的要求豁免  
Exemption number 6706 granted. Model 560XL exempt from requirements of FAR § 25.677(b) for horizontal stabilizer position indicator.

(b) 豁免号 8621 被授予。560XL 型 对 FAR 25.813(e)条有关客舱间舱门安装的要求豁免  
Exemption 8621 granted. Model 560XL exempt from requirements of FAR § 25.813(e) for installation of interior doors between passenger compartments.

(F) 等效安全水平如下/Equivalent levels of safety as follows:

(a) FAR 25.807(e)应急出口水上迫降 (有关水障碍)

FAR § 25.807(e), Emergency exits ditching;

(b) FAR 25.815 条, 乘客客舱过道宽度, 见注 10

FAR § 25.815, Passenger Cabin Aisle Width(See Note 10);

(c) FAR 25.813(e)条, 乘客舱之间的厕所门安装

FAR § 25.813(e), Lavatory door installation between passenger compartments;

(d) FAR 25.811(d)(1)条及 25.812(b)(1)(i)条, 应急出口标识及位置标记

FAR §§ 25.811(d)(1); 25.812(b)(1)(i), Emergency exit markings and locator signs;

(e) FAR 25.841(b)(6)条, 高高度机场起飞和着陆运行

FAR § 25.841(b)(6), Takeoff and landing operations at high elevation airports;

(f) FAR 25.1549(a)和(b)涡轮发动机 N2 只有数字显示

FAR § 25.1549(a) and (b), Digital only display of turbine engine N2.

- (g) FAR 1.1, 1.2; 25.101, 25.105, 25.109, 25.113, 25.115, 25.735, 25.1587, 终止起飞距离及着陆性能标准（包括磨损刹车标准）；和  
FAR 1.1; 1.2; 25.101, 25.105, 25.109, 25.113, 25.115, 25.735, 25.1587, Rejected takeoff distance and landing performance criteria (includes worn brake criteria), and;
- (h) FAR 25.13055(a)(4) (5) (6), (c)(1) (3), 25.1549(a)到(d), APU 发动机转子速度、排气温度只有数字显示，滑油压力或滑油温度没有指示。  
FAR 25.1305(a)(4) (5) (6), (c)(1) (3), and 25.1549(a) thru (d), Digital only display of APU engine rotor speed, exhaust gas temperature and no indication of oil pressure or oil temperature.
- (i) ELOS #SP2095WI-T-AG-8, 颁布于 2002 年 8 月 23 日, FAR 25.815 客舱过道宽度  
ELOS #SP2095WI-T-AG-8, dated August 23, 2002, FAR §25.815, Passenger Cabin Aisle Width.
- (j) FAR 25.1301(a)(d), 25.1303(a)(3), 25.1309(a)(b)(d)(e), 25.1316, 25.1327, 25.1331, 25.1333, 25.1547, 25.1351(d), 25.1353(a)(c)和 25.1459(e), 电动备用方向指示器（罗盘）（适用于 XLS）。  
FAR 25.1301(a)(d), 25.1303(a)(3), 25.1309(a)(b)(d)(e), 25.1316, 25.1327, 25.1331, 25.1333, 25.1547, 25.1351(d), 25.1353(a)(c), and 25.1459(e), Electric Standby Direction Indicator (Compass) (Applies to XLS).
- (G) FAR 25.801 水上迫降没有符合  
FAR § 25.801 ditching not complied with.
- (H) 已按照 FAR 25.1419 条演示符合防冰要求  
Compliance with ice protection has been demonstrated in accordance with FAR § 25.1419.

#### 序列号 S/N 560-6001 及之后/ S/N 560-6001 and on (XLS+)

- (A) FAR 25 部（生效日期：1965 年 2 月 1 日）修正案 25-1 至修正案 25-82，以及下列附加条款及豁免：  
Part 25 of the Federal Aviation Regulations effective February 1, 1965, as amended by Amendments 25-1 through 25-82, with additions and exceptions as follows:
- (a) 附加条款/Additions:  
FAR §§ 25.305, 25.321, 25.331, 25.333, 25.335, 25.341, 25.343, 25.345, 25.349, 25.371, 25.373, 25.391, 25.427, 25.1517 为修正案 25-86；25.351 为修正案 25-91。  
FAR §§ 25.305, 25.321, 25.331, 25.333, 25.335, 25.341, 25.343, 25.345, 25.349, 25.371, 25.373, 25.391, 25.427, 25.1517 as amended by Amendment 25-86; 25.351 as amended by Amendment 25-91.
- 对于主发动机安装，FAR §§25.901, 25.903, 25.933, 25.934, 25.939, 25.943 按修正案 25-1 至 25-117。  
FAR §§25.901, 25.903, 25.933, 25.934, 25.939, and 25.943 as amended by Amendments 25-1 through 25-117 for main engine installations.
- 对于序列号 560-6247 及之后飞机的全球定位系统传感器，FAR §§25.1317(b)按 25-1 至 25-122 号修正案。  
FAR §§25.1317(b) as amended by Amendment 25-1 through 25-122, is applicable to Global Positioning System Sensor for S/N 560-6247 and on.
- 对于安装可选 ATN-B1 数据链的驾驶舱语音记录器安装，FAR §§25.1457(a)(6)按 25-1 至 25-124 号修正案。  
FAR §§ 25.1457(a)(6) as amended by Amendment 25-1 thru 25-124, is applicable to the Cockpit Voice Recorder Installation when the optional ATN-B1 Datalink Installation is installed.

## (b) 豁免（如下表所示）/ Exceptions (as shown in table):

| 条款号     | 标题                                                                                                                        | 有效修正案                                             | 例外<br>[非审定基础的一部分]                  |
|---------|---------------------------------------------------------------------------------------------------------------------------|---------------------------------------------------|------------------------------------|
| 25.562  | 应急着陆动态条件                                                                                                                  | 25-82 适用                                          | 25.562(c)(5)和(c)(6)                |
| 25.571  | 结构损伤容限和疲劳评定                                                                                                               | 25-82 适用                                          | 25.571(e)(1)                       |
| 25.631  | 鸟击损伤                                                                                                                      | 无，此条款并非审定基础的一部分                                   | 25.631 不适用                         |
| 25.671  | 操作系统-总体<br>• 适用于 2 位置的水平安定面。<br><br>• 所有其他飞机操纵系统。                                                                         | 25-82<br><br>原版适用(25-1 至 25-17)                   | 无<br><br>25.671 修正案 25-23 及其以后不适用  |
| 25.677  | 配平系统<br>• 适用于 2 位置的水平安定面。<br><br>• 适用于其他飞机配平系统，包括升降舵配平。                                                                   | 25-82<br><br>原版适用<br>(25-1 至 25-17)               | 无<br><br>25.677 修正案 25-23 及其以后不适用  |
| 25.1309 | 设备、系统和安装<br>• 只适用于电子飞行仪表系统 (Rockwell Collins ProLine 21)，飞行指引，液压，电气，增压系统，和 2-位置水平安定面。<br><br>• 所有其他飞机系统<br><br>• FADEC 安装 | 25-82<br><br>原版适用<br>(25-1 至 25-17)<br><br>25-117 | 无<br><br>25.1309 修正案 25-23 及其以后不适用 |

| SECTION NO. | TITLE                                                                                                                           | EFFECTIVE AMENDMENT                                            | EXCEPTIONS<br>[Not Part of Cert. Basis]                                   |
|-------------|---------------------------------------------------------------------------------------------------------------------------------|----------------------------------------------------------------|---------------------------------------------------------------------------|
| 25.562      | Emergency landing dynamic conditions.                                                                                           | 25-82 Applicable                                               | §25.562(c)(5)(6)                                                          |
| 25.571      | Damage-tolerance and fatigue evaluation of structure.                                                                           | 25-82 Applicable                                               | §25.571(e)(1)                                                             |
| 25.631      | Bird strike damage                                                                                                              | None, this section is not part of cert. basis.                 | §25.631 not applicable                                                    |
| 25.671      | Control Systems – General.<br>• Applicable to the 2-position horizontal stabilizer<br><br>• All other airplane control systems. | 25-82<br><br>Original Issue Applicable<br>(25-1 through 25-17) | None<br><br>§ 25.671 as amended by Amdts. 25-23 and later, not applicable |
| 25.677      | Trim Systems                                                                                                                    | 25-82                                                          | None                                                                      |

|         |                                                                                                                                                                                                                                                                                                                                                        |                                                                                 |                                                                             |
|---------|--------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------|---------------------------------------------------------------------------------|-----------------------------------------------------------------------------|
|         | <ul style="list-style-type: none"> <li>Applicable to the 2-position horizontal stabilizer.</li> <li>All other airplane trim systems, including the elevator trim.</li> </ul>                                                                                                                                                                           | Original Issue<br>Applicable<br>(25-1 through 25-17)                            | §25.677 as amended by Amdts. 25-23 and later, not applicable                |
| 25.1309 | Equipment, systems, and installations. <ul style="list-style-type: none"> <li>Applicable to Electronic Flight Instrument systems (Honeywell Primus 1000 Cockpit Display), Flight Guidance, hydraulic, electrical, pressurization system, and 2-position horizontal stabilizer only.</li> <li>All other airplane systems.</li> </ul> FADEC Installation | 25-82<br><br>Original Issue<br>Applicable<br>(25-1 through 25-17)<br><br>25-117 | None<br><br>§ 25.1309 as amended by Amdts. 25-23 and later, not applicable. |

(B) 对于 XLS+ (S/N 560-6001 到 560-6317 和 560-6319 到 560-6321) : FAR 第 36 部, 1969 年 12 月 1 日生效, 噪声标准, 36-1 至 36-28 修正案。对于 XLS+ (S/N 560-6318 和 560-6322 及以后) : FAR 第 36 部, 1969 年 12 月 1 日生效, 噪声标准, 36-1 至 36-31 修正案。

FAR Part 36 effective December 1, 1969, as amended by Amendments 36-1 thru 36-28 for S/N 560-6001 to 560-6317 and 560-6319 to 560-6321. FAR Part 36 effective December 1, 1969, as amended by Amendments 36-1 thru 36-31 for S/N 560-6318 and 560-6322 and on.

(C) FAR 第 34 部, 1990 年 9 月 10 日生效, 第 34-1 修正案。

FAR Part 34 effective September 10, 1990, as amended by Amendments 34-1.

(D) 专用条件如下/Special Conditions as follows:

- 25-ANM-79, 高强度辐射场的影响。与系统闪电防护有关的部分不适用, 以及 25-ANM-79, effects for High Intensity Radiated Fields (HIRF). The portions associated with System Lightning Protection do not apply; and
- 25-ANM-21, 高高度运行(45000 英尺), 见注 7。  
High Altitude Operation (45,000 feet). See Note 7.

(E) 豁免/Exemption:

- 豁免号 6706 被授予。560XL 型对 FAR 25.677(b)条有关水平安定面位置指示器的要求豁免  
Exemption number 6706 granted. Model 560XL exempt from requirements of FAR § 25.677(b) for horizontal stabilizer position indicator.
- 豁免号 8621 被授予。560XL 型对 FAR 25.813(3)条有关乘客舱之间内部门安装的要求豁免  
Exemption number 8621 granted. Model 560XL exempt from requirements of FAR § 25.813(3) for installation of interior doors between passenger compartments.

(F) 等效安全水平如下/Equivalent levels of safety as follows:

- FAR 25.807(e)应急出口水上迫降 (有关水障碍)  
FAR § 25.807(e), Emergency exits ditching (involves water barrier).

- (b) FAR 25.815 条, 乘客客舱过道宽度, 见注 10  
FAR § 25.815, Passenger Cabin Aisle Width(See Note 10).
- (c) FAR 25.813(e)条, 乘客舱之间的厕所门安装  
FAR § 25.813(e), Lavatory door installation between passenger compartments.
- (d) FAR 25.811(d)(1)条及 25.812(b)(1)(i)条, 应急出口标识及位置标记  
FAR §§ 25.811(d)(1); 25.812(b) (1) (i), Emergency exit markings and locator signs.
- (e) FAR 25.841(b)(6)条, 高高度机场起飞和着陆运行  
FAR § 25.841(b)(6), Takeoff and landing operations at high elevation airports.
- (f) FAR 25.1549(a)和(b)涡轮发动机 N2 只有数字显示  
FAR § 25.1549(a) and (b), Digital only display of turbine engine N2.
- (g) FAR 1.1, 1.2; 25.101, 25.105, 25.109, 25.113, 25.115, 25.735, 25.1587, 终止起飞距离及着陆性能标准 (包括磨损刹车标准)。  
FAR §§ 1.1; 1.2; 25.101, 25.105, 25.109, 25.113, 25.115, 25.735, 25.1587, Rejected takeoff distance and landing performance criteria (includes worn brake criteria).
- (h) FAR §§ 25.1305(a)(4)(5)(6), (c)(1)(3), 25.1549(a)到(d), APU 发动机转子速度、排气温度只有数字显示, 滑油压力或滑油温度没有指示。  
FAR §§ 25.1305(a)(4)(5)(6), (c)(1) (3), and 25.1549(a) through (d), Digital only display of APU engine rotor speed, exhaust gas temperature and no indication of oil pressure or oil temperature.
- (i) ELOS #SP2095WI-T-AG-8, 2002 年 8 月 23 日, FAR 25.815, 客舱通道宽度。  
ELOS #SP2095WI-T-AG-8, dated August 23, 2002, FAR §25.815, Passenger Cabin Aisle Width.
- (j) ELOS #AT4531 WI-T-SE-1 FAR 25.1301(a)(d), 25.1303(a)(3), 25.1309(a)(b)(d)(e), 25.1316, 25.1324, 25.1331, 25.1333, 25.1547, 25.1351(d), 25.1353(a)(c), 25.1459(e), 电子备用方向指示器 (罗盘)。  
ELOS #AT4531 WI-T-SE-1 FAR §§ 25.1301(a)(d), 25.1303(a)(3), 25.1309(a)(b)(d)(e), 25.1316, 25.1327, 25.1331, 25.1333, 25.1547, 25.1351(d), 25.1353(a)(c), and 25.1459(e), Electric Standby Direction Indicator (Compass).
- (k) ELOS #AT4531WI-T-P-1, 2008 年 3 月 20 日, 21.21(b)(1), 25.1549, 数字显示仅显示发动机高压涡轮转速(N2)及发动机燃油流量(Wf), 和备用 N1, N2 及 ITT。  
ELOS #AT4531WI-T-P-1, dated March 20, 2008, §§21.21(b)(1), 25.1549, Use of Numeric Digital Only Display for Engine High-Pressure Turbine Speed (N2) and Engine Fuel Flow (Wf), and Standby N1, N2, and ITT.
- (l) ELOS #AT4531 WI-T-SE-2, 2008 年 5 月 30 日, 25.1397(c), 25.1401(d), 外部灯光色度要求。  
ELOS #AT4531WI-T-SE-2, dated May 30, 2008, §§25.1397(c), 25.1401(d), Exterior Lighting Chromaticity Requirements.
- (G) FAR 25.801 水上迫降没有符合  
FAR § 25.801 ditching not complied with.
- (H) 已按照 FAR 25.1419 条表明符合防冰要求  
Compliance with ice protection has been demonstrated in accordance with FAR § 25.1419.

中国民用航空局附加技术条件/CAAC Additional Technical Conditions

无。CAAC 接受 FAA 审定基础作为认可审定基础。

None.CAAC accepts FAA Certification Basis as Validation Basis.

## 生产依据/ Production Basis:

生产许可证 PC-4 于 1985 年 6 月 1 日生效, 或以后颁发。生产许可证 PC-312 于 1971 年 9 月 9 日至 1985 年 5 月 31 日有效。1985 年 2 月 15 日及以后有效的生产许可证 PC-4 适用于所有备件生产。PC-4 的最新颁发日期和对飞机型号和序列号的集体有效性见注 6。

Production Certification No. PC-4 effective June 1, 1985, or later issue Production Certification No. PC-312 effective September 9, 1971, through May 31, 1985. Effective February 15, 1985, and on, Production Certificate No. PC-4 is applicable to all spare production. See Note 6 for latest issue date of PC-4 and specific affectivity on airplane models and serial number.

## 3. 技术特征和使用限制/Technical Characteristics and Operational Limitations

## 发动机/Engines:

**序列号 560-5001 到 560-5500/ S/N 560-5001 through 560-5500**

2 台加拿大普惠公司型号为: PW545A 的涡轮风扇发动机(发动机型号合格证编号 E-28, 型号认可证编号 VTC0160E, 版次 1)

Two Pratt & Whitney Canada Corporation PW545A Turbofans (Type Certificate No. E-27, Validation Type Certificate No. VTC0160E, Rev.1)

**序列号 560-5501 到 560-6000/ S/N 560-5501 through 560-6000**

2 台加拿大普惠公司型号为: PW545B 的涡轮风扇发动机(发动机型号合格证编号 E-28, 型号认可证编号 VTC0160E, 版次 1)

Two Pratt & Whitney Canada Corporation PW545B Turbofans (Type Certificate No. E-27, Validation Type Certificate No. VTC0160E, Rev.1)

**序列号 560-6000 及以后/ S/N 560-6001 and on**

2 台加拿大普惠公司型号为: PW545C 的涡轮风扇发动机(发动机型号合格证编号 E-28, 型号认可证编号 VTC0160E, 版次 1)

Two Pratt & Whitney Canada Corporation PW545C Turbofans (Type Certificate No. E-27, Validation Type Certificate No. VTC0160E, Rev.1)

## 燃油/Fuel:

**序列号 560-5001 到 560-5500/ S/N 560-5001 through 560-5500**

Jet A, Jet A-1, Jet B, JP-4, JP-5, JP-8, 3 号喷气燃料(GB 6537)。对防冰添加剂的使用, 参考 FAA 批准的飞机飞行手册。

Jet A, Jet A-1, Jet B, JP-4, JP-5, JP-8, No.3 Jet Fuel (GB 6537). For use of anti-icing additives, refer to the FAA Approved Airplane Flight Manual.

**序列号 560-5501 到 560-6000/ S/N 560-5501 through 560-6000**

Jet A, Jet A-1, Jet B, JP-4, JP-5, JP-8, 3 号喷气燃料(GB 6537)。对防冰添加剂的使用, 参考 FAA 批准的飞机飞行手册。

Jet A, Jet A-1, Jet B, JP-4, JP-5, JP-8, No.3 Jet Fuel (GB 6537). For use of anti-icing additives, refer to the FAA Approved Airplane Flight Manual.

**序列号 560-6000 及以后/ S/N 560-6001 and on**

Jet A, Jet A-1, JP-5, JP-8, 3 号喷气燃料(GB 6537)。对防冰添加剂的使用, 参考 FAA 批准

的飞机飞行手册。

Jet A, Jet A-1, JP-5, JP-8, No.3 Jet Fuel (GB 6537). For use of anti-icing additives, refer to the FAA Approved Airplane Flight Manual.

#### 发动机限制/Engine Limits:

##### 序列号 560-5001 到 560-5500 / S/N 560-5001 through 560-5500

##### PW545A 涡轮风扇发动机/ PW545A Turbofans

|                                                                             |        |           |
|-----------------------------------------------------------------------------|--------|-----------|
| 标准日海平面推力/Static thrust, sea level, standard day                             |        |           |
| 起飞（5分钟）/Takeoff (5min)                                                      |        | 3952磅/lbs |
| 最大连续/Maximum Continuous                                                     |        | 3372磅/lbs |
| 最大允许发动机转子转速(PW545A)/ Max. permissible engine rotor operating speed (PW545A) |        |           |
| N1（风扇）/N1 (Fan)                                                             | 100%   | 13034rpm  |
| N2（燃气发生器）/N2( Gas Gen.)                                                     | 101.8% | 33289rpm  |
| 最大允许的涡轮内部燃气温度/Max. permissible interturbine temperature                     |        |           |
| 起飞/Takeoff                                                                  |        | 720°C     |
| 最大连续/ Maximum Continuous                                                    |        | 720°C     |
| 启动/Starting                                                                 |        | 720°C     |
| 瞬时（20秒）/Transient (20 seconds)                                              |        | 760°C     |

##### 序列号 560-5501 到 560-6000 / S/N 560-5501 through 560-6000

##### PW545B 涡轮风扇发动机/ PW545B Turbofans

|                                                                             |        |           |
|-----------------------------------------------------------------------------|--------|-----------|
| 标准日海平面推力/Static thrust, sea level, standard day                             |        |           |
| 起飞（5分钟）/Takeoff (5min)                                                      |        | 4119磅/lbs |
| 最大连续/Maximum Continuous                                                     |        | 3372磅/lbs |
| 最大允许发动机转子转速(PW545B)/ Max. permissible engine rotor operating speed (PW545B) |        |           |
| N1（风扇）/N1 (Fan)                                                             | 100%   | 13034rpm  |
| N2（燃气发生器）/N2(Gas Gen)                                                       | 102.8% | 33622rpm  |
| 最大允许的涡轮内部燃气温度/Max. permissible interturbine temperature                     |        |           |
| 起飞/Takeoff                                                                  |        | 740°C     |
| 最大连续/ Maximum Continuous                                                    |        | 720°C     |
| 启动/Starting                                                                 |        | 740°C     |
| 瞬时（20秒）/Transient (20 seconds)                                              |        | 780°C     |

##### 序列号 560-6001 及之后 / S/N 560-6001 and on

##### PW545C 涡轮风扇发动机/ PW545C Turbofans

|                                                 |  |           |
|-------------------------------------------------|--|-----------|
| 标准日海平面推力/Static thrust, sea level, standard day |  |           |
| 起飞（5分钟）/Takeoff (5min)                          |  | 4119磅/lbs |
| 最大连续/Maximum Continuous                         |  | 3372磅/lbs |

最大允许发动机转子转速(PW545C)/ Max. permissible engine rotor  
operating speed(PW545C):

|                         |        |          |
|-------------------------|--------|----------|
| N1 (风扇) /N1 (Fan)       | 100%   | 13034rpm |
| N2 (气体发生器) /N2(Gas Gen) | 102.8% | 33622rpm |

最大允许的涡轮内部燃气温度/Max. permissible interturbine temperature

|                                  |       |
|----------------------------------|-------|
| 起飞/Takeoff                       | 740°C |
| 最大连续/ Maximum Continuous         | 720°C |
| 启动/Starting                      | 740°C |
| 瞬时 (20秒) /Transient (20 seconds) | 780°C |

#### 空速限制/Airspeed Limits:

V<sub>MO</sub> (最大使用) /V<sub>MO</sub> (Maximum Operating)

海平面到8000英尺/Sea Level to 8000 ft. 261KCAS (260KIAS)

8000英尺到26515英尺/8000 to 26515 ft. 306KCAS (305KIAS)

M<sub>MO</sub> (26515英尺以上) /M<sub>MO</sub> Above 26515 ft 0.752马赫(0.750MIAS)

V<sub>A</sub> (海平面) /V<sub>A</sub> (Sea Level)

20000 lb 196KCAS (195KIAS)

随重量和高度的变化见AFM/See AFM for variation with weight and altitude.

V<sub>B</sub> (最大阵风强度的速度) /V<sub>B</sub>(Speed for max. gust intensity)

211KCAS(210KIAS)

V<sub>FE</sub> (襟翼展开/Flaps extended)

35° (着陆/Landing) 175KCAS(173KIAS)

15° (起飞和进近/Takeoff and approach) 201KCAS(200KIAS)

7° (起飞/Takeoff) 201KCAS(200KIAS)

V<sub>MCA</sub> (空中最小操纵速度(起飞)) / Air (Takeoff) [Min control speed] 90KCAS(90KIAS)

V<sub>MCL</sub> (空中最小操纵速度(着陆)) / Air (Landing) [Min control speed] 92KCAS(92KIAS)

V<sub>MCG</sub> (地面最小操纵速度) /(Minimum control speed) Ground: 98KCAS (98KIAS)

V<sub>LO</sub> (起落架放下操作) /Landing gear operating extend): 251KCAS (250KIAS)

#### 轮胎限制/Tire limit

最大地面速度/Maximum ground speed:

165节/knots

#### \*重心范围/C.G. Range

起落架放下/Landing Gear Extended 序列号/ S/N 560-5001 through 560-5500

\*见注 11

重心前限/Forward Limits

从20200磅时的基准线后方324.18英寸(21.39% MAC)线性变化到11500磅时的基准线后方318.92英寸(15.00% MAC)。

Linear variation from 324.18 in. aft of datum (21.39% MAC) at 20,200 lb. to 318.92 in. aft of datum (15.00 % MAC) at 11,500 lb.

## 重心后限/Aft Limits

从15000磅到11500磅为基准线后方331.26英寸(30.0% MAC)。

从17800磅时的基准线后方330.74英寸(29.37% MAC)线性变化到15000磅时的基准线后方331.26(30.0% MAC)。

从17800磅到20200磅为基准线后方330.74英寸(29.37% MAC)。

331.26 in. aft of datum (30.0% MAC) from 15,000 lb. through 11,500 lb.

Linear variation from 330.74 in. aft of datum (29.37% MAC) at 17,800 lb. to 331.26 in. aft of datum (30.0% MAC) at 15,000 lb.

330.74 in. aft of datum (29.37% MAC) from 17,800 lb. through 20,200 lb.

## 重心范围/C.G. Range

## 起落架放下/Landing Gear Extended    序列号/ S/N 560-5501 through 560-6000

## 重心前限/Forward Limits

从20400磅时的基准线后方324.29英寸(21.52% MAC)线性变化到11500磅时的基准线后方318.92英寸(15.00% MAC)。

Linear variation from 324.29 in. aft of datum (21.52% MAC) at 20,400 lb. to 318.92 in. aft of datum (15.00 % MAC) at 11,500 lb.

## 重心后限/Aft Limits

从15000磅到11500磅为基准线后方331.26英寸(30.0% MAC)。

从15000磅时的基准线后方331.26英寸(30.0% MAC)线性变化到17800磅时的基准线后方330.74(29.37% MAC)。

从17800磅到20400磅为基准线后方330.74英寸(29.37% MAC)。

331.26 in. aft of datum (30.0% MAC) from 15,000 lb. through 11,500 lb.

Linear variation from 331.26 in. aft of datum (30.0% MAC) at 15,000 lb. to 330.74 in. aft of datum (29.37% MAC) at 17,800 lb.

330.74 in. aft of datum (29.37% MAC) from 17,800 lb. through 20,400 lb.

## 重心范围/C.G. Range

## 起落架放下/Landing Gear Extended    序列号560-6001及以后/ S/N 560-6001 and on

## 重心前限/Forward Limits

从20400磅时的基准线后方324.29英寸(21.52% MAC)线性变化到11500磅时的基准线后方318.92英寸(15.00% MAC)。

Linear variation from 324.29 in. aft of datum (21.52% MAC) at 20,400 lb. to 318.92 in. aft of datum (15.00 % MAC) at 11,500 lb.

## 重心后限/Aft Limits

从15000磅到11500磅为基准线后方331.26英寸(30.0% MAC)。

从15000磅时的基准线后方331.26英寸(30.0% MAC)线性变化到17800磅时的基准线后方330.74(29.37% MAC)。

从17800磅到20400磅为基准线后方330.74英寸(29.37% MAC)。

331.26 in. aft of datum (30.0% MAC) from 15,000 lb. through 11,500 lb.

Linear variation from 331.26 in. aft of datum (30.0% MAC) at 15,000 lb. to 330.74 in. aft of datum (29.37% MAC) at 17,800 lb.  
330.74 in. aft of datum (29.37% MAC) from 17,800 lb. through 20,400 lb.

空机重心范围/ Empty Wt. C.G. Range:

无/None

基准/Datum

零参考基准线是紧靠着水平线 127.25 处客舱门后方的配平螺钉前面 221.0 处。  
Zero reference datum is 221.0 inches forward of the leveling screw just aft of the cabin door on W.L. 127.25

平均空气动力弦/MAC

82.231 英寸（基准线后部 306.593 英寸的 MAC 前缘）。  
注意：这是对不带翼尖的基本机翼的参考弦长。  
82.231 in. (Leading edge of MAC 306.593 in. aft of datum)  
Note: This is reference MAC for basic wing without and tip.

水平测量方法/Leveling Means

舱门内侧地板面板平行于 B.L. 13.00。  
Outboard floor panel inside of door parallel to B.L. 13.00.

**\*序列号 560-5001 到 560-5500 / S/N 560-5001 through 560-5500**

\*见注 11/See Note 11

最大重量/Maximum

|               |            |
|---------------|------------|
| 起飞/Takeoff    | 20000磅/lb. |
| 着陆/Landing    | 18700磅/lb. |
| 零燃油/Zero fuel | 15000磅/lb. |
| 停机坪/Ramp      | 20000磅/lb. |

最小重量/Minimum Weight

|              |           |
|--------------|-----------|
| 飞行中/Inflight | 12400磅/lb |
|--------------|-----------|

**序列号 560-5501 到 560-6000 / S/N 560-5501 through 560-6000**

最大重量/Maximum

|               |            |
|---------------|------------|
| 起飞/Takeoff    | 20200磅/lb. |
| 着陆/Landing    | 18700磅/lb. |
| 零燃油/Zero fuel | 15100磅/lb. |
| 停机坪/Ramp      | 20400磅/lb. |

最小重量/Minimum Weight

|              |           |
|--------------|-----------|
| 飞行中/Inflight | 12400磅/lb |
|--------------|-----------|

**序列号 560-6001 到 560-6317 和 560-6319 到 560-6321 / S/N 560-6001 thru 560-6317 and 560-6319 thru 560-6321****最大重量/Maximum**

|               |            |
|---------------|------------|
| 起飞/Takeoff    | 20200磅/lb. |
| 着陆/Landing    | 18700磅/lb. |
| 零燃油/Zero fuel | 15100磅/lb. |
| 停机坪/Ramp      | 20400磅/lb. |

**最小重量/Minimum Weight**

|              |           |
|--------------|-----------|
| 飞行中/Inflight | 12400磅/lb |
|--------------|-----------|

**序列号 560-6318 和 560-6322 及以后 / S/N 560-6318 and 560-6322 and on****最大重量/Maximum**

|               |            |
|---------------|------------|
| 起飞/Takeoff    | 20330磅/lb. |
| 着陆/Landing    | 18700磅/lb. |
| 零燃油/Zero fuel | 15360磅/lb. |
| 停机坪/Ramp      | 20530磅/lb. |

**最小重量/Minimum Weight**

|              |           |
|--------------|-----------|
| 飞行中/Inflight | 12400磅/lb |
|--------------|-----------|

**最小机组/Minimum Crew**

所有飞行-两人（驾驶和副驾驶）/ For all flights: 2 persons (pilot and co-pilot)

**座位数/No. of Seats**

2 到 14 人（两名驾驶员，12 名乘客）/ 2 to 14 (2 Pilots, 12 Passengers)

见注 10/See Note 10

**最大行李/Maximum Baggage**

尾锥/Tailcone 基准线后方 431 英尺处 700 磅/ 700 lb. at 431.0 in. aft of datum

**燃油量（加仑）/Fuel Capacity (Gal.)**

两个大翼油箱/ Two wing tanks 每个总量为505.8; 可用量为503.0  
Total 505.8 each; usable 503.0 each

ARM基准线后部328.8英寸/ARM 328.8 in. aft of datum

不可用燃油数据见注1/See NOTE 1 for data on unusable fuel

**滑油容量（夸脱）/Oil Capacity (Quarts)**

两个装于发动机的滑油箱挂箱/Two engine mounted tanks 每个总量为7.5; 不可用量为0.6  
Total 7.5 each; usable 0.6 each

ARM 基准线后部 433.9 英寸/ARM +433.9 in. aft of datum

**最大运行高度/Maximum Operating**

45000 英尺/ft.

## 操纵面运动/Control Surfaces Movements

|                                                                                      |                                                          |               |         |               |
|--------------------------------------------------------------------------------------|----------------------------------------------------------|---------------|---------|---------------|
| 升降舵（安定面在+1°）/Elevator(With stabilizer +1°)                                           | 向上/Up                                                    | 19°+1°, -0°   | 向下/Down | 15°±1°        |
| 升降舵调整片（安定面在+1°）/Elevator trim tab (With stabilizer +1°)                              | 向上/Up                                                    | 5°±1°         | 向下/Down | 15°±1°        |
| 方向舵/Rudder                                                                           | 向右/Right                                                 | 22°+1°, -0°   | 向左/Left | 22°+1°, -0°   |
| 对于安装了方向舵偏移系统的飞机/For Aircraft with rudder bias system installed                       |                                                          |               |         |               |
| 升降舵（垂直于铰链）/Rudder (perpendicular to hinge)                                           | 向右/Right                                                 | 28°+0.5°, -0° | 向左/Left | 28°+0.5°, -0° |
| 方向舵配平片（方向舵定中时垂直于铰链）<br>Rudder trim tab (perpendicular to hinge with Rudder centered) | 向右/Right                                                 | 11.5°±0.5°    | 向左/Left | 11.5°±0.5°    |
| 副翼/Aileron                                                                           | 上/Up                                                     | 19°±1°        | 下/Down  | 15°±1°        |
| 副翼调整片/Aileron trim tab                                                               | 上/Up                                                     | 20°±1°        | 下/Down  | 20°±2°        |
| 机翼襟翼/Wing flap                                                                       | 0° 和 7°, 15°, 35° ±1°<br>0°, and extend 7°, 15°, 35° ±1° |               |         |               |
| 减速板-上面/Speed brake-Upper                                                             | 向上/Up                                                    | 60°+2°        |         |               |
| -下面/-Lower                                                                           | 向上/Up                                                    | 65°+2°        |         |               |
| 2-位置水平安定面/ 2-position Horizontal stabilizer                                          |                                                          |               |         |               |
| 起飞和着陆/T/O & Landing                                                                  |                                                          | -2°+0°, -0.1° |         |               |
| 巡航/Cruise                                                                            |                                                          | +1°+0.1°, -0° |         |               |
| 调整指南见飞机维修手册/See Airplane Maintenance Manual for rigging instructions.                |                                                          |               |         |               |

## 经批准的中文标记标牌/The Approved Chinese Placards

The Chinese Placards involved in the following data have been accepted by CAAC and approved by FAA

| 标题/Title                                                | 编号/No.    | 版次/Version                                                                 |
|---------------------------------------------------------|-----------|----------------------------------------------------------------------------|
| 内部中文信息标记<br>CHINESE INFORMATION<br>MARKINGS - INTERIORS | C-GEN-290 | B 版, 含后续批准版次<br>Rev.B, including the<br>subsequent approved<br>version(s). |

## 制造商序列号/Manufacturer Serial No. Eligible:

560-5001 到 560-5500 (Excel)/560-5001 through 560-5500 (Excel)  
 560-5501 到 560-6000 (XLS)/560-5501 through 560-6000 (XLS)  
 560-6001 及以后。(XLS+)/560-6001 and on (XLS+)

## 型号合格证信息/Type Certificate Information

1968 年 7 月 16 日申请型号合格证。

1971 年 9 月 9 日颁发型号合格证 A22CE。

Application for Type Certificate dated July 16, 1968.

Type Certificate No. A22CE issued September 9, 1971.

## 设备/ Equipment:

适用的适航规章里所要求的基本设备（见审定基础）必须安装在飞机上。

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

## 第 4 部分 注释（型别：560/560XL）

## SECTION 4 NOTE (Models: 560/560XL)

注释/NOTE 1: 审定的空机重量包括了设备清单中的设备重量，现行有效的重量与平衡报告应包含设备清单及加载说明。如需要，在每一架飞机初始审定时应当提供。

Current weight and balance report including list of equipment included in certificated empty weight, and loading instructions when necessary must be provided for each aircraft at the time of original certification.

审定的空机重量及对应的重心位置应该包括：

The certified empty weight and corresponding center of gravity location must include:

|                     |                        |                                             |
|---------------------|------------------------|---------------------------------------------|
| 不可用燃油/Unusable fuel | 47.6 lb. at +300.1 in. | (560 Encore+, S/N 560-0751 through 560-5000 |
|                     | 37.8 lb. at +335.6 in. | 560XL S/N 560-5001 and on                   |

|              |                        |                                             |
|--------------|------------------------|---------------------------------------------|
| 满滑油/Full Oil | 34.1 lb. at +387.0 in. | (560 Encore+, S/N 560-0751 through 560-5000 |
|              | 23.7 lb. at +433.9 in. | 560XL S/N 560-5001 and on                   |

|                      |                         |                                             |
|----------------------|-------------------------|---------------------------------------------|
| 液压油/ Hydraulic fluid | 15.4 lb. at +228.02 in. | (560 Encore+, S/N 560-0751 through 560-5000 |
|                      | 34.2 lb. at +354.0 in.  | 560XL S/N 560-5001 and on                   |

|                                         |                      |                                             |
|-----------------------------------------|----------------------|---------------------------------------------|
| 防冰液/Anti-Ice fluid<br>(风挡玻璃/Windshield) | 3.4 lb. at +91.5 in. | (560 Encore+, S/N 560-0751 through 560-5000 |
|-----------------------------------------|----------------------|---------------------------------------------|

注释/NOTE 2: 飞机必须按照 FAA 批准的下表所列或后续批准版本的飞机飞行手册 (AFM) 运行。所有 FAA 批准的 AFM、适用的运行规则或审定基础要求的标牌，都必须按本型号合格证规定的飞机装配图零件清单 xxxxxxxx 进行安装。标牌也可以参考德事隆图解零件目录 (IPC)。如果在 IPC 和正在检查的飞机之间发现任何差异，则需要使用先前所述的装配图零件清单进行协调。

Airplanes must be operated according to the FAA Approved AFM, Part Number as listed in the table below (or later FAA approved revision). All placards required by either the FAA Approved AFM, the applicable operating rules, or the certification basis, must be installed as specified for this Type Certificate via Parts List xxxxxxxx, Airplane Assembly. A useful placarding reference is the Textron Aviation Illustrated Parts Catalogue (IPC). Any discrepancies identified between the IPC and an aircraft under inspection needs to be reconciled using the previously stated parts list.

| 型别/Model      | 飞行手册件号（或后续批准版本）/AFM P/N<br>(Or later approved revision) | 飞机装配图号 /Airplane Assembly |
|---------------|---------------------------------------------------------|---------------------------|
| 560XL (Excel) | 56XFM-03                                                | 6600000                   |
| 560 XL-(XLS)  | 56XFMA-00                                               | 6600000                   |
| 60 XL-(XLS+)  | 56XFMB-00                                               | 6600000                   |

|            | 560 Encore+                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                         | 56FMC-00 | 6500560 |
|------------|---------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------|----------|---------|
| 注释/NOTE 3: | <p>检查、强制报废寿命信息以及其它持续适航性要求见维修手册第 4 章“适航性限制”。</p> <p>See Maintenance Manual, Chapter 4, "Airworthiness Limitations" for inspections, mandatory retirement life information, and other requirements for continued airworthiness.</p>                                                                                                                                                                                                                                                                                                                                                                                                                                                                                   |          |         |
| 注释/NOTE 4: | <p>替换的座椅（机组和乘客），即使符合 TSO C39，也必须证明符合 CCAR 25.785 条的要求。所有型号的 560 XL 替换座椅必须符合 CCAR25.562 的要求，应急着陆动态条件，见审定基础中。</p> <p>All Model 500, 550, 552, S550 and 560 replacement seats (crew and passenger), although they may comply with TSO C39, must also be demonstrated to comply with FAR 25.785. All Model 560XL replacement seats must comply with FAR § 25.562, Emergency landing dynamic conditions, as shown in the certification basis.</p>                                                                                                                                                                                                                                                                        |          |         |
| 注释/NOTE 5: | <p>批准的前起落架轮胎限制列在 FAA 批准的飞行手册的限制章节中。</p> <p>Approved nose gear tires are limited to those listed in the Limitations Section of the FAA Approved Airplane Flight Manual.</p>                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          |          |         |
| 注释/NOTE 6: | <p>1998 年 5 月 7 日颁发的生产许可证 No.PC-4 适用于 560 (Encore+)型飞机，序列号 560-0751 到 560-5000；以及 560XL，序列号 560-5000 及以后。</p> <p>Production Certificate No. PC-4 issued May 7, 1998. Applies to 560 (Encore+) S/N 560-0751 through 560-5000; And 560XL beginning at S/N 560-5000 and on.</p>                                                                                                                                                                                                                                                                                                                                                                                                                                        |          |         |
| 注释/NOTE7:  | <p>一些型别已通过专用条件或符合某些 25 部条款被批准高高度运行（高度超过 41000 英尺）。对压力舱所做的任何改装必须按照相关审定基础中的相关要求得到批准。这包括无论是因裂纹增长或天线丢失可能导致压力舱开口的改装，对于 560 (Encore+)，其开口尺寸大于 4.00 平方英尺；对于 560XL，其开口尺寸不大于 3.98 平方英寸。</p> <p>Certain models have been approved for high altitude operations (altitudes above 41000 feet), either by Special Conditions or compliance with Part 25 sections. Any modifications to the pressure vessel must be approved in accordance with the requirements as shown in the appropriate certification basis. This includes modifications which could result in a pressure vessel opening, either through crack-growth or antenna loss, for 560 (Encore +), greater than 4.00 sq. in, for 560XL, greater than 3.98 sq.in.</p> |          |         |
| 注释/NOTE 8: | <p>560 (Encore+)及 560XL 型飞机，满足在减少的垂直最小间隔(RVSM)初始适航要求。</p> <p>Model 560 (Encore+) and 560XL meet the initial airworthiness requirements for operation in Reduced Vertical Separation Minimum (RVSM) airspace.</p>                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    |          |         |
| 注释/NOTE 9: | <p>560XL (S/N 560-5001 到 560-5500)，当左后厕所或左后侧向座椅被安装并被批准在起飞和着陆过程中坐人时，左分割组件（件号：6679017-1）或等同物必须一直被装上。结构分割件是座椅约束系统的整体部分。</p> <p>Model 560XL (s/n 560-5001 through 560-5500) Left divider assembly (part no. 6679017-1) or equivalent must always be installed when the LH aft toilet or LH aft side-facing seat installed and approved for occupancy during takeoff and landing. The structural divider is an integral part of the seat restraint system.</p> <p>560XL (S/N 560-5501 through 560-6000)，当左后厕所或左后侧向座椅被安装并被批准在起飞</p>                                                                                                                                                                                       |          |         |

和着陆过程中坐人时，左分割组件（件号：4589029-1）或等同物必须一直被装上。结构分割件是座椅约束系统的整体部分。

Model 560XL. (s/n 560-5501 through 560-6000) Left divider assembly (part no. 4589029-1) or equivalent must always be installed when the LH aft toilet or LH aft side-facing seat installed and approved for occupancy during takeoff and landing. The structural divider is an integral part of the seat restraint system.

560XL (S/N 560-6001 及之后)，当左后厕所或左后侧向座椅被安装并被批准在起飞和着陆过程中坐人时，左分割组件（件号：4569035-1）或等同物必须一直被装上。结构分割件是座椅约束系统的整体部分。

Model 560XL. (s/n 560-6001 and on) Left divider assembly (part no. 4569035-1) or equivalent must always be installed when the LH aft toilet or LH aft side-facing seat installed and approved for occupancy during takeoff and landing. The structural divider is an integral part of the seat restraint system.

注释/NOTE10: 560XL 通道宽度 ELOS 适用于 7 座到 12 座乘客座椅布局，当从通道地板测量的距离在 25 英寸到 27.5 英寸之间时，允许的最小通道宽度为 15 英寸。任何对过道宽度的进一步减小要求 FAA 进一步评估，并且没有包括在此次给予的等效安全水平中。

Model 560XL width of aisle equivalent level of safety applies to passenger seating arrangements from 7 to 12 passengers, and allows a minimum aisle width of 15 inches when measured from 25 inches to 27.5 inches from the dropped aisle floor. Any further reduction in aisle width requires further FAA evaluation and is not included in this grant of equivalent level of safety.

注释/NOTE 11: 560XL型，序列号 560-5001到560-5500，按照 SB 560XL-32-28进行过改装的飞机：

重心范围（起落架放下）：

重心前限：从20400磅时的基准线后方324.30英寸(21.54% MAC)线性变化到12400磅时的基准线后方319.47英寸(15.81% MAC)。

重心后限：从15000磅到12400磅为基准线后方后331.26英寸(30.0% MAC)；

从17800磅时的基准线后方331.26英寸(29.37% MAC)线性变化到15000磅时的基准线后方330.47英寸(30.0% MAC)。

在17800磅到20400磅为基准线后方330.74英寸(29.37% MAC)

|       |     |            |
|-------|-----|------------|
| 最大重量： | 起飞  | 20200磅/lb. |
|       | 着陆  | 18700磅/lb. |
|       | 零燃油 | 15100磅/lb. |
|       | 停机坪 | 20400磅/lb. |

|       |     |            |
|-------|-----|------------|
| 最小重量： | 飞行中 | 12400磅/lb. |
|-------|-----|------------|

Model 560XL aircraft, S/N 560-5001 through 560-5500, modified per SB560XL-32-28:

C.G. Range (Landing Gear Extended):

Forward Limits: Linear variation from 324.30 in. aft of datum (21.54% MAC) at 20,400 lb. to 319.47in. aft of datum (15.81 % MAC) at 12,400 lb.

Aft Limits: 331.26 in. aft of datum (30.0% MAC) from 15,000 lb. through 12,400 lb.

Linear variation from 331.26 in. aft of datum (29.37% MAC) at 17,800 lb. to 330.74 in. aft of datum (30.0% MAC) at 15,000 lb.

330.74 in. aft of datum (29.37% MAC) from 17,800 lb. through 20,400 lb.

|                 |            |
|-----------------|------------|
| Maximum Weight: |            |
| Takeoff         | 20,200 lb. |
| Landing         | 18,700 lb  |
| Zero fuel       | 15,100 lb. |
| Ramp            | 20,400 lb. |
| Minimum Weight: |            |
| Inflight        | 12,400 lb. |

注释/NOTE 12: 从使用的尾锥设备出来的可燃蒸汽通风已经通过试飞演示尾锥内部和外部压差来被满意的验证。此次验证只是在一定设备安装和使用情况下完成。尾锥内部设备安装或其他安装额外点火源可能影响尾锥压差的改装必须与 Wichita 航空器审定中心协调。

The venting of flammable vapors away from operating tailcone equipment has been satisfactorily demonstrated by flight test demonstration of the differential between tailcone internal and external area pressure. This demonstration was accomplished only with certain equipment installed and operating. Equipment installations or other modifications to the tailcone which add additional ignition sources or possibly affect tailcone differential pressures must be coordinated with the Wichita Aircraft Certification Office.

注释/NOTE 13: 以下系列产品以 Cessna Aircraft Company 的公司名制造, 560XL: 从序列号 560-5001 至 560-6191。

The following serials are manufactured under the name Cessna Aircraft Company: 560XL: 560-5001 thru 560-6191.

注释/NOTE 14: 公司名变更自 2015 年 7 月 29 日开始生效。以下系列产品以 Textron Aviation Inc. 的公司名制造, 560XL: 从序列号 560-6192 往后。

Company name change effective 7/29/15. The following serials are manufactured under the name Textron Aviation Inc.: 560XL: 560-6192 and on.

注释/NOTE 15: 560XL: 从序列号 560-5001 至 560-5500, 560-5501 至 560-6000, 560-6001 及以后。

设备要求: 适用的适航规章里所要求的基本设备(见审定基础)必须安装在飞机上。只有使用水剂和 U.S. – UL 5B:C – Halotron BrX (2-BTP), C3H2BrF3, CAS 编号 1514 82 5 的手提式灭火器才被批准使用。

Model 560XL, S/N 560-5001 thru 560-5500, 560-5501 thru 560-6000, 560-6001 and on.

Required Equipment: The basic required equipment prescribed in the applicable airworthiness requirements (see certification basis) must be installed in the aircraft. Only hand fire extinguishers that use water and U.S. – UL 5B:C – Halotron BrX (2-BTP), C3H2BrF3, CAS Number 1514 82 5 are approved for use.

----完----  
----END----