

型号认可数据单 THE VALIDATION DATA SHEET

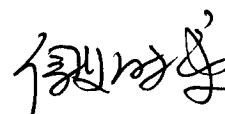
编号/NO: VTC0162A

版本/Revision: 2

型号/Type:

C90A, C90GT, C90GTi

批准人/Approved By:



日期/Date: 2012.3.31

本数据单是型号认可证（编号：VTC0162A）的组成部分，它规定了符合中国民用航空局的适航要求所颁发此型号认可证的产品状态和限制。

This data sheet, which is part of Validation of Type Certificate (No: VTC0162A), prescribes condition and limitation under the product for which the type certificate was issued meets the airworthiness requirements of the Chinese Civil Aviation Regulation.

型号认可证持有人/Validation of Type Certificate Holder:

Hawker Beechcraft Corporation

9709 East Central

Wichita, Kansas 67201

型号合格证持有人记录/ Type Certification Holder record:

1996 年 4 月 15 日 Beech Aircraft Corporation 转为 Raytheon Aircraft Company.

2007 年 3 月 26 日 Raytheon Aircraft Company 转为 Hawker Beechcraft Corporation.

Beech Aircraft Corporation transferred to Raytheon Aircraft Company on April 15, 1996
Raytheon Aircraft Company transferred to Hawker Beechcraft Corporation on March 26, 2007

有效页清单/List of effective pages:

页数/Pages	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	
版次/Revision	2	2	1	1	1	2	1	1	1	1	1	2	2	1	2	
页数/Pages	16	17	18	19	20	21	22	23	24							
版次/Revision	2	1	1	2	2	2	2	2	2							

第 1 部分 概述

SECTION 1 GENERAL

I. 概述/General

1. CAAC 认可数据单和型号合格证数据单

CAAC Validation Data Sheet and Type Certificate Data Sheet

	中国民用航空局认可数据单号、版次和颁发日期 CAAC Validation Data Sheet Number, Revision and issuance date	型号合格证数据单号、版次和颁发日期 Type Certificate Data Sheet Number, Revision and issuance date
当前 Current	编号/Number: VTC0162A 版次/Revision: Revision 2 日期/Date: 2012 年 3 月 31 日	编号/Number: 3A20 版次/Revision: REVISION 70 日期/Date: June 2, 2011
历史 History	编号/Number: VTC-162A	
	版次/Revision: Original Issue	
	日期/Date: 30 APRIL 2005	
	编号/Number: VTC0162A	编号/Number: 3A20
	版次/Revision: Revision 1	版次/Revision: REVISION 68
	日期/Date: 13 July 2009	日期/Date: 6 MAY 2009

2. 类别/Category: 正常类/Normal Category

3. 认可当局/Validation Authority: 中国民用航空局/CAAC

审定当局/Certifying Authority: 美国联邦航空局/FAA

4. FAA 合格证

型号合格证 Type Certificate

Number: 3A20

型号合格证持有人/Type Certificate Holder:

Hawker Beechcraft Corporation

9709 East Central

Wichita, Kansas 67201

5. 生产基础/Production Basis

依据生产许可证 8 号制造

Manufacture under Production Certificate No.8.

6. 申请 CAAC 认可审定的日期

Date of Application for CAAC Validation

7. CAAC 型号认可日期/CAAC Validation Date

30 April 2005

II. 认可审定基础/Validation Basis

CAAC 的认可审定基础由 FAA 审定基础中的适用要求和 CAAC 的附加技术条件组成

The CAAC validation basis is consists of the FAA certification basis as applicable and the CAAC additional technical conditions.

1. FAA 审定基础的适用要求/FAA certification basis as applicable

适用于 C90A

Applicable to Model C90A

CAR 3, Effective May 15, 1956, (Am. 3-1, 3-2, 3-8); CAR 3 (Amdt. 3-6); CAR3 §3.705(Amdt. 3-7); 14 CFR Part 23 §23.1385(c) (Amdt. Original); 14 CFR Part 23 §23.959, 23.1111 and 23.1583(a) (Amdt. 23-7). 14 CFR Part 23 §23.1387(a) and 23.1387(e) (Amdt. 23-12). Special Conditions outlined by FAA letters to Beech dated January 21, 1963, February 15, 1963 and February 27, 1963. Special Conditions outlined by FAA letters to Beech dated May 5, 1965. 14 CFR Part 23 § 23.143(a), 23.145(d), 23.153, 23.161(c)(3) and 23.173(a) (Amdt. 23-14); §23.175(Amdt. 23-17); §23.967(a)(5) (Amdt. 23-18); §23.1545(a) (Amdt. 23-23); §23.729 (Amdt. 23-26); 14 CFR Part 25 § 25.831(d) (Amdt. 25-41). 14 CFR Part 36, December 1, 1969 through (Amdt) 36-10; SFAR 27, February 1, 1974 through Amendment 27-4.

适用于拥有 EFIS 的 C90A

Applicable to Model C90A with EFIS:

CAR 3, Effective May 15, 1956, (Am. 3-1, 3-2, 3-8); CAR 3 (Amdt. 3-6); CAR3 §3.705(Amdt. 3-7); 14 CFR Part 23 §23.1385(c) (Amdt. Original); 14 CFR Part 23 §23.959, 23.1111 and 23.1583(a) (Amdt. 23-7). 14 CFR Part 23 §23.1387(a) and 23.1387(e) (Amdt. 23-12). Special Conditions outlined by FAA letters to Beech dated January 21, 1963, February 15, 1963 and February 27, 1963. Special Conditions outlined by FAA letters to Beech dated May 5, 1965. 14 CFR Part 23 § 23.143(a), 23.145(d), 23.153, 23.161(c)(3) and 23.173(a) (Amdt. 23-14); §23.175(Amdt. 23-17); §23.967(a)(5) (Amdt. 23-18); §23.1545(a) (Amdt. 23-23); §23.729 (Amdt. 23-26); Effective April 17, 1992, Electronics Flight Instrument Systems shall meet the requirements of §23.1301, 23.1309, 23.1311, 23.1321, 23.1322 and 23.1335 (Amdt. 23-41); Effective January 20, 1994, §23.1457(Amdt. 23-35); 14 CFR Part 25 § 25.831(d) (Amdt. 25-41). 14 CFR Part 36, December 1, 1969 through Amendment 36-10; SFAR 27, February 1, 1974 through Amendment 27-4.

适用于 C90GT

Applicable to Model C90GT

CAR 3, Effective May 15, 1956, (Am. 3-1, 3-2, 3-8); CAR 3 (Amdt. 3-6); CAR3 §3.705(Amdt. 3-7); 14 CFR Part 23 §23.1385(c) (Amdt. Original); 14 CFR Part 23 §23.959, 23.1111 and 23.1583(a) (Amdt. 23-7). 14 CFR Part 23 §23.1387(a) and 23.1387(e) (Amdt. 23-12). Special Conditions outlined by FAA letters to Beech dated January 21, 1963, February 15, 1963 and February 27, 1963. Special Conditions outlined by FAA letters to Beech dated May 5, 1965. 14 CFR Part 23 § 23.143(a), 23.145(d), 23.153, 23.161(c)(3) and 23.173(a) (Amdt. 23-14); §23.175(Amdt. 23-17); §23.967(a)(5) (Amdt. 23-18); §23.1545(a) (Amdt. 23-23); §23.729 and 23.1529(Amdt. 23-26); Effective April 17, 1992, Electronics Flight Instrument Systems shall meet the requirements of §23.1301, 23.1309, 23.1311, 23.1321, 23.1322 and 23.1335 (Amdt. 23-41); Effective January 20, 1994, §23.1457(Amdt. 23-35); §23.201, 23.203, and

23.207(Amdt.23-50); 14CFR Part 25 § 25.831(d) (Amdt.25-41). 14CFR Part 34, §34.11(a)(b)(c), 34.21(e), 34.71, 34.89(Amdt.34-3); 14CFR Part 36, December 1, 1969 through Amendment 36-25; SFAR 27, February 1, 1974 through Amendment 27-4.

在 C90A/C90GT 安装 IHAS 800 附加要求

Additional Requirements for IHAS 800 Installation on Model C90A/C90GT:

14 CFR Part 23 §23.301(a), 23.303, 23.305(a)(b), 23.307(a), 23.321, 23.331, 23.333, 23.335, 23.337, 23.341, 23.347, 23.349, 23.351, 23.365(a)(b)(c), 23.425, 23.441, 23.471, 23.473, 23.571, 23.573, 23.609(b), 23.613(a)(b), 23.615, 23.619, 23.621, 23.623, 23.625, 23.627, 23.629, 23.1331(a)(b), 23.1365(a)(d)(e), 23.1367(a)(b)(c)(d) (Amdt.Original); §23.611, 23.617, 23.777(a)(b), 23.867(a)(b), 23.561(a)(b), 23.1351(b) (Amdt. 23-7); §23.1581(Amdt. 23-13); §23.1322(a)(b)(c)(d) (Amdt. 23-17); 23.1301(a)(b)(d), 23.1321(a)(b)(c)(d)(e), 23.1357(a)(b)(c)(d) (Amdt.23-20); 23.1541(Amdt.23-21); §23.1307(b) (Amdt.23-23); §23.1529(Amdt.23-26); §23.1583(m), 23.1587(Amdt.23-34); §23.1311(a)(b), 23.1331(c), 23.1359(c) (Amdt.23-35); §23.1309(a)(b), (Amdt.23-41)

在 C90A/C90GT 安装 RDR2100/KDM850 附加要求

Additional Requirement for RDR2100/KDM850 Installation on Model C90A/C90GT

14 CFR Part 23 §23.301(a), 23.303, 23.305(a)(b), 23.307(a), 23.321, 23.331, 23.333, 23.335, 23.337, 23.341, 23.347, 23.349, 23.351, 23.365(a)(b)(c), 23.421, 23.423, 23.425, 23.441, 23.471, 23.473, 23.571, 23.573, 23.601, 23.603(a)(b), 23.605(a), 23.607, 23.609(a)(b), 23.613(a)(b), 23.615, 23.619, 23.621, 23.623, 23.625, 23.627, 23.629, 23.1331(a)(b), 23.1365(a)(d)(e), 23.1367(a)(b)(c)(d) (Amdt. Original); §23.611, 23.617, 23.777(a)(b), 23.867(a)(b), 23.561(a)(b), 23.959, 23.1111, 23.1351(a) (Amdt. 23-7); §23.1581(Amdt. 23-13); §23.1322(a)(b)(c)(d) (Amdt. 23-17); 23.1301(a)(b)(d), 23.1321(a)(b)(c)(d)(e), 23.1357(a)(b)(c)(d) (Amdt.23-20); §23.1307(b), (Amdt.23-23); 23.1529(Amdt.23.26), §23.1583(m), 23.1585(j), 23.1587(Amdt.23-34); §23.1311(a)(b), 23.1331(c), 23.1359(c) (Amdt.23-35); §23.1309(a)(b), (Amdt.23-41)

在 C90A/C90GT 安装 GPS 400 附加要求

Additional Requirements for GPS 400 Installation on Model C90A/C90GT:

14 CFR Part 23 §23.301(a), 23.303, 23.305(a)(b), 23.307(a), 23.321, 23.331, 23.333, 23.335, 23.337, 23.341, 23.347, 23.349, 23.351, 23.365(a)(b)(c), 23.421, 23.423, 23.425, 23.441, 23.471, 23.473, 23.571, 23.573, 23.601, 23.603(a)(b), 23.605(a), 23.607, 23.609(a)(b), 23.613(a)(b), 23.615, 23.619, 23.621, 23.623, 23.625, 23.627, 23.1331(a)(b), 23.1365(a)(d)(e), 23.1367(a)(b)(c)(d), 23.1431(a)(b)(c) (Amdt.Original); §23.611, 23.617, 23.777(a)(b), 23.867(a)(b), 23.561(a)(b), 23.1351(b) (Amdt. 23-7); §23.1581 (Amdt. 23-13); §23.1322(a)(b)(c)(d) (Amdt. 23-17); §23.1301(a)(b)(c)(d), 23.1321(a)(b)(c)(d)(e), 23.1357(a)(b)(c)(d) (Amdt. 23-20); §23.1307(b) (Amdt. 23-23); §23.1529 (Amdt. 23.26); §23.1583(m), 23-1585(j), 23.1587 (Amdt. 23-34); §23.1329(h), 23.1331(c), 23.1359(c) (Amdt. 23-35); §23.1309(a)(b) (Amdt. 23-41).

适用于 C90GTi

Applicable to Model C90GTi:

CAR 3, Effective May 15, 1956, (Am.3-1, 3-2, 3-8); CAR 3 (Amdt.3-6); CAR3 §3.705(Amdt.3-7); 14 CFR Part 23 §23.601; 23 §23.1385(c) (Amdt.Original); 23 §23.959, 23.1111 and 23.1583(a) (Amdt.23-7). 14CFR Part 23 § 23.1387(a) and 23.1387(e) (Amdt.23-12). Special Conditions outlined by FAA letters to Beech dated January 21,

1963, February 15, 1963 and February 27, 1963. Special Conditions outlined by FAA letters to Beech dated May 5, 1965. 14CFR Part 23 § 23.143(a), 23.145(d), 23.153, 23.161(c)(3) and 23.173(a) (Amdt.23-14); §23.175(Amdt.23-17); §23.967(a)(5) (Amdt.23-18); §23.1545(a) (Amdt.23-23); §23.729 (Amdt.23-26) and 23.1529(Amdt.23-26). §23.201; 23.203; 23.207; (Amdt.23-50). 14CFR Part 25 § 25.831(d) (Amdt.25-41). 14CFR Part 34, §34.11(a)(b)(c), 34.21(e), 34.71, 34.89(Amdt.34-3); SFAR 27, February 1, 1974 through Amendment 27-4.

安装 Rockwell Collins Pro Line 21 航电设备的附加要求

Additional Requirement for Rockwell Collins Pro Line 21 Avionics Installation:

23.1367(a)(b)(c)(d); 23.1381(a)(b)(c) (Amit. Original). §23.1301(a)(b)(c)(d); 23.1335(Amit.23-20); 23.1501(a) (Amdt.23-21). §23.1457(a)(c)(Amdt.23-35). §23.1322(a)(b)(c)(d); 23.1357; (Amdt.23-43). §23.1549(Amdt.23-45). §23.1309(a)(b)(c)(d)(e); 23.1311(a)(b)(c); 23.1321(a)(b)(c)(d)(e); 23.1329(a)(b)(c)(d)(e)(f); 23.1359(c); 23.1365(a)(b)(c)(d)(e); 23.1431(a)(b)(c)(Amdt.23-49). 23-1521(b)(c); 23.1543(c); 23.1545(a)(b)(c); 23.1555(a)(Amdt.23-0). §23.1305(a)(2)(3), (c)(1)(2)(5), (e)(1)(Amdt.23-52). §23.901(e)(1)(Amdt.23-53). 14 CFR Part 36, through Amendment 36-28.

专用条件	23-108-SC	在 HIRF 条件下的系统保护
Special Condition	23-108-SC	"Protection of Systems for High Intensity Radiated Field(HIRF)"
等效安全	ACE-07-06	电子发动机指示系统 (EIS) 安装
Equivalent Level of Safety	ACE-07-06	"Installing Electronic Engine Indicating System(EIS)"

1958 年 5 月 1 日申请型号合格证

1959 年 2 月 4 日颁发型号合格证 No.3A20, 依据委托授权方式由制造人获得。

Application for Type Certificate dated May 1, 1958

Type Certificate No.3A20 Issued February 4, 1959, obtained by the manufacture under delegation option Procedures.

生产基础: 生产许可证 No. 8, 授权委任制造人 CE-2 根据 FAR 21 部有关委任单位的规定颁发适航证。

Production Basis: Production Certificate No.8. Delegation Option Manufacture No.CE-2 authorized to issue airworthiness certificates under delegation option provision of Part 21 of the Federal Aviation Regulations.

2. CAAC 附加技术条件/CAAC additional technical conditions

2.1 CCAR21.29, Amendment Level: R3

2.2 环保标准/Environmental Standards Noise: CCAR-36-R1, dated 15 April 2007

2.3 用于告知乘客或地面救援人员舱门位置和打开方法的标记标牌必须满足中文要求

All literal markings and placards on an airplane used for indicating passengers or outboard rescue personnel of the location of doors and their means of opening must be at least in Chinese language.

2.4 燃油符合中国航空燃油 3 号喷气燃料 (GB -6537-2006) ;

Fuel conforms to China Aviation No.3 JET FUEL (PRC National Standard GB-6537-2006) ;

2.5 进口要求/Import Requirements

为具备在中国运行的资格, 按本型号认可证制造的每架航空器必须具有一份出口适航证, 其上签注有出口国适航当

局的审定声明（用英文），即：本架航空器符合 CAAC 所批准的型号设计(VTC0207A)并处于可安全运行的状态。

To be considered eligible for operation in China, each aircraft manufactured under this Validation of Type Certificate must be accompanied by a certificate of airworthiness for export endorsed by FAA or FAA_authorized person which states (in English language): This aircraft conforms to its CAAC type design (VTC0207A) and is in a condition for safe operation .

2.6 产品有效性/Series Nos. Eligible (Note 1)

对于申请 CAAC 审定的每一架飞机，其出口适航证都必须由 FAA 按上述“进口要求”予以签注。

A FAA Certificate of Airworthiness for export endorsed as noted below under “Import Requirements” must be submitted for each individual airplane for which application for the CAAC certification.

2.7 资料批准/Service Information

CAAC 接受的服务通告、结构修理手册、航空器飞行手册、供应商手册和航空器维护手册等使用维护资料应经 FAA 批准，FAA 的这些批准代表了 CAAC 的批准，且仅限于本型号设计。

Service bulletins, structural repair manuals, aircraft flight manuals, vender manuals, and aircraft maintenance manuals which contain a statement that the document is FAA approved, are accepted by the CAAC and are considered CAAC-approved. These approvals pertain to the type design only.

3 技术特性/Technical Characteristic

3.1 CAAC 批准的型号设计的定义文件 CAAC Approved Type Design Definition (applicant) document

- a. C90A AIRCRAFT GENERAL ASSY-MODEL C90A DWG NO. 90-000007
- b. C90GT AIRCRAFT GENERAL ASSY-MODEL C90GT DWG NO. 90-000008
- c. C90GTi AIRCRAFT GENERAL ASSY-MODEL C90GTi DWG NO. 90-000009
- d. PLACARDS, INTERIOR—CHINA DWG NO.002-380011
- f. PLACARD INSTL—INTERIOR AND EXTERIOR CHINA DWG NO. 90-005130

第 2 部分

SECTION 2

型号 C90A, 空中国王, (正常类), 1983 年 12 月 1 日获 FAA 批准。CAAC 于 2005 年 4 月 30 日认可。

Model C90A, King Air, (Normal Category), Approved December 1, 1983 by FAA. Validated April 30, 2005 by CAAC.

<u>发动机</u>	2 台 Pratt & Whitney Aircraft of Canada, Ltd. 的 PT6A-21(涡轮螺旋桨)
<u>Engines</u>	2 Pratt & Whitney Aircraft of Canada, Ltd. PT6A-21(Turboprop)
<u>燃油</u>	JP-4, JP-5(MIL-T-5624); JP-8(MIL-T-83133); Jet A, Jet A-1, 和 Jet B 符合 P&WC S.B. 1244 或 ASTM Spec D1655. 三号喷气燃料 (GB-6537-2006) 应急燃油见注 1
<u>Fuel</u>	JP-4, JP-5(MIL-T-5624); JP-8(MIL-T-83133); Jet A, Jet A-1, and Jet B conforming to P&WC S.B. 1244 or ASTM Spec D1655. No.3 JET(GB-6537-2006) See NOTE1 for Emergency fuels.
<u>滑油 (发动机和减速器)</u>	P&WC PT6 发动机 No. 1 号服务通报列表中允许的滑油牌号。
<u>Oil (Engine and Gearbox)</u>	P&WC PT6 Engine Service Bulletin No. 1 list approved brand oils

<u>发动机限制</u>	轴马力	N1 燃气涡轮转速	螺旋桨轴转速	最大许用涡轮间温度°C
起飞 (5 分钟)	550	101.5	2200*	695
最大连续	550	101.5	2200*	695
启动瞬间 (2 秒)		102.6		1090
最大反桨	300	88.0	2100	695

* 见注 2

在低高度及低环境温度下, 发动机起飞功率要比飞机审定时的功率大。在这种条件下, 扭矩表上的扭矩限制不得超越。

滑油温度	-40°F	最低启动温度
	-40°F to 210°F	低慢车
	50°F to 210°F	最大连续

<u>Engine Limits</u>	Max. Permissible			
	N1 Gas	Prop	Turbine Inter	
	Shaft	Generator	Shaft	stage Temp.
	horsepower	Speed	Speed	(Deg. C)

Takeoff(5 minutes)	550	101.5	2200*	695
Max Continuous	550	101.5	2200*	695
Start Transient(2 seconds)		102.6		1090
Max Reverse	300	88.0	2100	695

See NOTE 2

At low altitude and low ambient temperature the engines may produce more power at takeoff than that for which the airplane has been certificated. Under these conditions, the placarded torque meter limit shall not be exceeded.

Oil temperature: -40°F minimum starting
-40°F to 210°F low idle
50°F to 210°F max. continuous

螺旋桨和螺旋桨限制

具有反桨功能的四片桨叶螺旋桨（见注 3）

两个 Hartzell 公司带 D8990K 螺旋桨叶的 HC-E4N-3N 桨毂

依据 Hawker Beechcraft 公司的规范 BS186497.

直径: 90.00 in. (明义尺寸) 最小允许修复至 89.00 in.

(不允许进一步减少)

在 30in 站位的桨距设置

飞行慢车止动点 (见注 4)

辅助飞行慢车止位 (见注 4)

反桨 -11°

顺桨 87°

Propeller and Propeller Limits

Reversing Four Bladed Propeller (See NOTE 3)

2 Hartzell HC-E4N-3N hubs with D8990K blades per

Hawker Beechcraft Corporation Specification BS186497.

Diameter: 90.00 in. (Normal) Minimum allowable for repair 89.00 in.

(no further reduction permitted)

Pitch Settings at 30 in. Sta:

Flight Idle Stop (See NOTE 4)

Secondary flight idle stop (See NOTE4)

Reverse -11°

Feather 87°

螺旋桨和螺旋桨限制

非反桨螺旋桨

2 副哈泽尔 HC-B3TN-2(B)/T10173B-8

S/N LJ-1063 至 LJ-1287, LJ-1288
至 LJ-1294, LJ-1296 至 LJ-1299

直径: 93-3/8 英寸 (名义), 允许修理的最小值 90-3/8 英寸。

(不允许进一步降低)

在 30 英寸站位的桨距设定: 低 19°, 顺桨 87°

Propeller and Propeller Limits
S/N LJ-1063 through LJ-1287,
LJ-1288 through LJ-1294, LJ-1296
through LJ-1299

螺旋桨和螺旋桨限制

S/N LJ-1288, LJ-1295, LJ-1302,
LJ-1303, LJ-1305 至 LJ-1308,
LJ-1311, LJ-1312, LJ-1314 至
LJ-1316, LJ-1318, LJ-1320 至
LJ-1366, LJ-1368 至 LJ-1372,
LJ-1374, LJ-1376, LJ-1378,
LJ-1383, LJ-1385, LJ-1387,
LJ-1388, LJ-1390 至 LJ-1393,
LJ-1395, LJ-1396, LJ-1398 至
LJ-1402, LJ-1404 至 LJ-1410,
LJ-1412 至 LJ-1424, LJ-1426 至
LJ-1430, LJ-1432 至 LJ-1434,
LJ-1436 至 LJ-1726, LJ-1728 至
LJ-1753, LJ-1755

Propeller and Propeller Limits

S/N LJ-1288, LJ-1295
LJ-1302, LJ-1303, LJ-1305 thru
LJ-1308, LJ-1311, LJ-1312 LJ-1314
thru LJ-1316, LJ-1318 LJ-1320 thru
LJ-1366, LJ-1368 thru LJ-1372,
LJ-1374 LJ-1376, LJ-1378 LJ-1383,
LJ-1385, LJ-1387, LJ-1388,
LJ-1390 thru LJ-1393, LJ-1395,
LJ-1396, LJ-1398 thru LJ-1402,
LJ-1404 thru LJ-1410,
LJ-1412 thru LJ-1424, LJ-1426 thru
LJ-1430, LJ-1432 thru LJ-1434,
LJ-1436 thru LJ-1729, LJ-1728 thru
LJ-1753, LJ-1755

Non Reversing Propeller

2 Hartzell HC-B3TN-2(B)/T10173B-8

Diameter: 93-3/8 in. (Nominal) Minimum allowable for repair 90-3/8 in.
(no further reduction permitted)

Pitch settings at 30 in. Sta.: Low 19°, Feather 87°

反桨螺旋桨

2 副 McCauley 4HFR34C768 桨毂, 带 94LMA-4 桨叶

直径: 90 英寸 (名义), 允许修理最小值 89 英寸。(不允许进一步降低)
30 英寸站位的桨距设置:

飞行慢车止位

见(注 5)

反桨

-10° ± 2°

顺桨

85.8° ± 2°

最小慢车速度

1100 rpm(注 6)

Reversing Propeller

2 McCauley 4HFR34C768 hubs with 94LMA-4 blades

Diameter: 90 in. (Nominal) Minimum allowable for repair
89 in. (no further reduction permitted)

Pitch settings at 30 in. sta.:

Flight idle stop

(See NOTE 5)

Reverse

-10° ± 2°

Feather

85.8° ± 2°

minimum idle speed

1100 rpm((See NOTE6)

空速限制

S/N LJ-1063 至 LJ-1137 和 LJ-1146

最大使用速度

CAS

260mph

CAS

(226 knots)

IAS

226 knots

机动速度

195mph

(169 knots)

169 knots

S/N LJ-1138 至 LJ-1145	襟翼放下速度（完全放下）	161mph	(140 knots)	148 knots
	最大起落架操作速度			
	放下	209mph	(182 knots)	182 knots
	收起	189mph	(164 knots)	163 knots
	最大起落架放下速度	209mph	(182 knots)	182 knots
	最大使用速度	260mph	(226 knots)	226 knots
	机动速度	195mph	(169 knots)	169 knots
LJ-1147 至 LJ-1726, LJ-1728 至 LJ-1753, LJ-1755	襟翼放下速度（完全放下）	161mph	(140 knots)	148 knots
	最大起落架操作速度			
	放下	209mph	(182 knots)	182 knots
	收起	189mph	(164 knots)	163 knots
	最大起落架放下速度	209mph	(182 knots)	182 knots
<u>Airspeed Limits</u>		<u>CAS</u>	<u>CAS</u>	<u>IAS</u>
S/N LJ-1063 through	Maximum operating speed	260mph	(226 knots)	226 knots
LJ-1137 and LJ-1146	Maneuvering	195mph	(169 knots)	169 knots
	Flap extended speed (full down)	161mph	(140 knots)	148 knots
	Maximum landing gear operating speed			
S/N LJ-1138 through LJ-1145	Extension	209mph	(182 knots)	182 knots
	Retraction	189mph	(164 knots)	163 knots
	Maximum landing gear extended speed	209mph	(182 knots)	182 knots
	Maximum operating speed	260mph	(226 knots)	226 knots
	Maneuvering	195mph	(169 knots)	169 knots
LJ-1147 through LJ-1726, LJ-1728 through LJ-1753, LJ-1755	Flap extended speed (full down)	161mph	(140 knots)	148 knots
	Maximum landing gear operating speed			
	Extension	209mph	(182 knots)	182 knots
	Retraction	189mph	(164 knots)	163 knots
	Maximum landing gear extended speed	209mph	(182 knots)	182 knots
<u>重心范围（起落架放下）</u>		9650lb 时(+152.0)		
S/N LJ-1063 至 LJ-1137 和 LJ-1146	7400 lb 时或更小(+144.7)至 (+160.0) 在 在给定的两点线性变化 由于起落架收起导致的力矩变化-3825in.-lb。			
<u>重心范围（起落架放下）</u>		10,100 磅时从(+152.0)到(+160.0)		
系列号从 LJ-1138 到 LJ-1145, 从 LJ-1147 到 LJ-1726, 从 LJ-1728 到 LJ-1753, LJ-1755（见注 7）	9999 磅时为(+151.7)（注释 7） 7850 lb 或更小时从(+144.7)到(+160.0) 给定各点间线性变化 收上起落架引起地力矩变化为-3825 in. -lb.			

<u>C.G. Range(Landing Gear Extended)</u>	(+152.0) to (+160.0) at 10,100lb			
S/N LJ-1063 through LJ-1137 and LJ-1146	(+144.7) to (+160.0) at 7850 lb or less			
	Straight line variation between points given			
	Moment change due to retracting landing gear -3825in.-lb.			
<u>C.G. Range (Landing Gear Extended)</u>	(+152.0) to (+160.0) at 10,100 lb.			
	(+151.7) at 9999 lbs. (Note 7)			
S/N LJ-1138 through LJ-1145,	(+144.7) to (+160.0) at 7850 lb or less			
LJ-1147 through LJ-1726,	Straight line variation between points given			
LJ-1728 through LJ-1753, LJ-1755	Moment change due to retracting landing gear -3825 in. –lb.			
(See Note 7)				
<u>空机重心范围</u>	无			
<u>Empty Wt. C.G. Range</u>	None			
<u>最大重量</u>	机坪	9710lb.		
S/N LJ-1063 至 LJ-1137 和 LJ-1146	起飞	9168lb		
	着陆	9650lb		
<u>最大重量</u>	机坪	10,160lb.		
S/N LJ-1138 至 LJ-1145, LJ-1147 至 LJ-1726,LJ-1728 至 LJ-1753,LJ-1755	起飞	10,100lb.		
	着陆	9,600lb		
<u>Maximum Weight</u>	Ramp:	9710lb.		
S/N LJ-1063 through LJ-1137 and LJ-1146	Takeoff	9168lb		
	Landing	9650lb.		
<u>Maximum Weight</u>	Ramp:	10,160lb.		
S/N LJ-1138 through LJ-1145,	Takeoff	10,100lb.		
LJ-1147LJ-1726,LJ-1728 through LJ-1753,LJ-1755	Landing	9,600lb		
<u>座位数</u>	最大 13 人（包括在+129 处的 2 个人）。对乘客就坐见装载说明			
<u>No. of Seats</u>	Maximum 13 (including 2 at +129). See loading instructions for passenger loading			
<u>最大行李载重</u>	350lb.(+275)			
<u>(结构限制)</u>	350lb.(+70)（行李及电子设备）			
<u>Maximum Baggage</u>	350lb.(+275)			
<u>(Structural Limit)</u>	350lb.(+70)(Baggage and Avionics)			
<u>燃油容量</u>	油箱	容积 Gal	可用 Gal	力臂
	左右吊舱	61ea	61 ea	+131
	左右机翼	131 ea.	131 ea	+167
	不可用燃油数据见注 8。			

<u>Fuel Capacity</u>	Tank	Cap Gal	Usable Gal	Arm
	L & R Nacelle	61ea	61 ea	+131
	L & R Wing	131 ea.	131 ea	+167
	See NOTE 8 for data on unusable fuel.			
<u>滑油容量</u>	共 22.4 qt. (+101)			
	不可用滑油数据见注 8			
<u>Oil Capacity</u>	22.4 qt.Total (+101)			
	See NOTE 8 for data on unusable oil			
<u>最大使用限制</u>	30,000ft 压力高度。			
<u>Max. Oper. Limit</u>	30,000ft. pressure Altitude.			
<u>操纵面使用范围</u>	襟翼	最大	44°	
	副翼调整片（只在左侧）	上	15°	下 15°
	副翼	上	20°	下 20°
	升降舵调整片	上	10°	下 21°
	升降舵反伺服调整片	上	12°	下 8°
	升降舵	上	25°	下 15°
	方向舵调整片	右	30°	左 30°
	方向舵	右	24°	左 19°
<u>Control Surface Movement</u>	Wing flaps	Maximum	44°	
	Aileron tab(left only)	Up	15°	Down 15°
	Aileron	Up	20°	Down 20°
	Elevator tab	Up	10°	Down 21°
	Elevator tab anti-servo	Up	12°	Down 8°
	Elevator	Up	25°	Down 15°
	Rudder tab	Right	30°	Left 30°
	Rudder	Right	24°	Left 19°
<u>适用的系列号</u>	LJ-1063 至 LJ-1726, LJ-1728 至 LJ-1753 和 LJ-1755。			
	见注 10			
<u>Serial Nos. Eligible</u>	LJ-1063 through LJ-1726, LJ-1728 through LJ-1753 and LJ-1755。			
	See NOTE 10			

型号 C90GT, 空中国王,(正常类), 2005 年 12 月 16 日获 FAA 批准。

Model C90GT, King Air, (Normal Category), Approved December 16, 2005 by FAA

发动机 2 台 Pratt & Whitney Aircraft of Canada, Ltd.的 PT6A-135A(涡轮螺旋桨)
依据 Hawker Beechcraft 公司的规范 BS184061.

Engines 2 Pratt & Whitney Aircraft of Canada, Ltd. PT6A-135A(Turboprop)
Per Hawker Beechcraft Corporation Specification BS184061

燃油

JP-4, JP-5(MIL-T-5624); JP-8(MIL-T-83133); Jet A, Jet A-1, 和 Jet B 符合 P&WC S.B. 1244 或 ASTM Spec D1655.
中国二号喷气燃料
应急燃油见注 1

Fuel

JP-4, JP-5(MIL-T-5624); JP-8(MIL-T-83133); Jet A, Jet A-1, and Jet B conforming to P&WC S.B. 1244 or ASTM Spec D1655;and Chinese Jet Fuel No.3.
See NOTE 1 for Emergency fuels.

滑油 (发动机和减速器)

P&WC PT6 发动机 No. 1001 号服务通报列表中允许的滑油牌号。

Oil (Engine and Gearbox)

P&WC PT6 Engine Service Bulletin No. 1001 list approved brand oils

发动机限制

	轴马力	N1 燃气涡轮转速	螺旋桨轴转速	最大许用涡轮间温度°C
起飞 (5 分钟)	550	101.5	1900*	805
最大连续	550	101.5	1900*	805
启动瞬间 (2 秒)		102.6		1090
最大反桨	300	88.0	1825	805

*见注 2

在低高度及低环境温度下, 发动机起飞功率要比飞机审定时的功率大。在这种条件下, 扭矩表上的扭矩限制不得超越。

滑油温度	-40°F	最低启动温度
	-40°F to 210°F	低慢车
	50°F to 210°F	最大连续

Engine Limits

		N1 Gas	Prop	Max. Permissible
	Shaft	Generator	Shaft	Turbine Inter
	horsepower	Speed	Speed	stage Temp.
				(Deg. C)
Takeoff(5 minutes)	550	101.5	1900*	805
Max Continuous	550	101.5	1900*	805
Start Transient(2 seconds)		102.6		1090
Max Reverse	300	88.0	1825	805

See NOTE 2

At low altitude and low ambient temperature the engines may produce more power at takeoff than that for which the airplane has been certificated. Under these conditions, the placarded torque meter limit shall not be exceeded.

Oil temperature:	-40°F	minimum starting
	-40°F to 210°F	low idle
	50°F to 210°F	max. continuous

螺旋桨和螺旋桨限制

具有反桨功能的四片桨叶螺旋桨（见注 3）

两个 Hartzell 公司带 D8990SK 螺旋桨叶的 HC-E4N-3N 桨毂

依据 Hawker Beechcraft 公司的规范 BS186497.

直径: 90.00 in. (明义尺寸) 最小允许可修复至 89.00 in.

(不允许进一步减少)

在 30in 站位的桨距设置

飞行慢车止动点

(见注 9)

反桨

-10 ±.5

顺桨

85.8 ±.5

最小慢车速度

1100 RPM (见注 6)

Propeller and Propeller Limits

Reversing Four Bladed Propeller (See NOTE 3)

2 Hartzell HC-E4N-3N hubs with D8990SK blades per

Hawker Beechcraft Corporation Specification BS186497.

Diameter: 90.00 in. (Normal) Minimum allowable for repair 89.00 in.

(no further reduction permitted)

Pitch Settings at 30 in. Sta:

Flight Idle Stop

(See **NOTE 9**)

Reverse

-10°±.5°

Feather

85.8°±.5

Minimum idle speed

1100 RPM (See **NOTE 6**)

空速限制

	<u>CAS</u>	<u>CAS</u>	<u>IAS</u>
最大使用速度	260mph	(226 knots)	226 knots
机动速度	195mph	(169 knots)	169 knots
襟翼放下速度（完全放下）	161mph	(140 knots)	148 knots
最大起落架操作速度			
放下	209mph	(182 knots)	182 knots
收起	189mph	(164 knots)	163 knots
最大起落架放下速度	209mph	(182 knots)	182 knots

Airspeed Limits

	<u>CAS</u>	<u>CAS</u>	<u>IAS</u>
Maximum operating speed	260mph	(226 knots)	226 knots
Maneuvering	195mph	(169 knots)	169 knots
Flap extended speed (full down)	161mph	(140 knots)	148 knots
Maximum landing gear operating speed			
Extension	209mph	(182 knots)	182 knots
Retraction	189mph	(164 knots)	163 knots
Maximum landing gear extended speed	209mph	(182 knots)	182 knots

<u>重心范围 (起落架伸出)</u>	(+152.0) 至 (+160.0) 在 10,100lb (+144.7)至 (+160.0) 在 7850 lb 或更小 在给定的两点线性变化 由于起落架收起导致的力矩变化-3825in.-lb。			
<u>C.G. Range(Landing Gear Extended)</u>	(+152.0) to (+160.0) at 10,100lb (+144.7) to (+160.0) at 7850 lb or less Straight line variation between points given Moment change due to retracting landing gear -3825in.-lb.			
<u>空机重心范围</u>	无			
<u>Empty Wt. C.G. Range</u>	None			
<u>最大重量</u>	机坪	10,160lb.		
	起飞	10,100lb.		
	着陆	9,600lb		
<u>Maximum Weight</u>	Ramp:	10,160lb.		
	Takeoff	10,100lb.		
	Landing	9,600lb		
<u>座位数</u>	最大 13 人 (包括在+129 处的 2 个人)。对乘客就坐见装载说明			
<u>No. of Seats</u>	Maximum 13 (including 2 at +129). See loading instructions for passenger loading			
<u>最大行李载重</u>	350lb.(+275)			
<u>(结构限制)</u>	350lb.(+70) (行李及电子设备)			
<u>Maximum Baggage</u>	350lb.(+275)			
<u>(Structural Limit)</u>	350lb.(+70)(Baggage and Avionics)			
<u>燃油容量</u>	油箱	容积 Gal	可用 Gal	力臂
	左右吊舱	61ea	61 ea	+131
	左右机翼	131 ea.	131 ea	+167
	不可用燃油数据见注 8。			
<u>Fuel Capacity</u>	Tank	Cap Gal	Usable Gal	Arm
	L & R Nacelle	61ea	61 ea	+131
	L & R Wing	131 ea.	131 ea	+167
	See NOTE 6 for data on unusable fuel.			
<u>滑油容量</u>	共 18.4 qt. (+101)			
	不可用滑油数据见注 8			
<u>Oil Capacity</u>	18.4 qt.total (+101)			
	See NOTE 6 for data on unusable oil			
<u>最大使用限制</u>	30,000ft 压力高度。			

<u>Max. Oper. Limit</u>	30,000ft. pressure Altitude.				
<u>操纵面使用范围</u>	襟翼	最大	44 °		
	副翼调整片 (只在左侧)	上	15 °	下	15 °
	副翼	上	20 °	下	20 °
	升降舵调整片	上	10 °	下	21 °
	升降舵反伺服调整片	上	12 °	下	8 °
	升降舵	上	25 °	下	15 °
	方向舵调整片	右	30 °	左	30 °
	方向舵	右	24 °	左	19 °
<u>Control Surface Movement</u>	Wing flaps	Maximum	44 °		
	Aileron tab(left only)	Up	15 °	Down	15 °
	Aileron	Up	20 °	Down	20 °
	Elevator tab	Up	10 °	Down	21 °
	Elevator tab anti-servo	Up	12 °	Down	8 °
	Elevator	Up	25 °	Down	15 °
	Rudder tab	Right	30 °	Left	30 °
	Rudder	Right	24 °	Left	19 °
<u>适用的系列号</u>	LJ-1727, LJ-1754, 从 LJ-1756 到 LJ-1846, 和从 LJ-1848 到 LJ-1852。 见注 9				
<u>Serial Nos. Eligible</u>	LJ-1727, LJ-1754, LJ-1756 thru LJ-1846, and LJ-1848 thru LJ-1852. See NOTE 9				

型号 C90GTi, 空中国王,(正常类), 2007 年 12 月 13 日获 FAA 批准。

Model C90GTi, King Air, (Normal Category), Approved December 13, 2007 by FAA

<u>发动机</u>	2 台 Pratt & Whitney Aircraft of Canada, Ltd. 的 PT6A-135A(涡轮螺旋桨) 依据 Hawker Beechcraft 公司的规范 BS184061.
<u>Engines</u>	2 Pratt & Whitney Aircraft of Canada, Ltd. PT6A-135A(Turboprop) Per Hawker Beechcraft Corporation Specification BS184061
<u>燃油</u>	JP-4, JP-5(MIL-T-5624); JP-8(MIL-T-83133); Jet A, Jet A-1, 和 Jet B 符合 P&WC S.B. 1244 或 ASTM Spec D1655. 中国二号喷气燃料 应急燃油见注 1
<u>Fuel</u>	JP-4, JP-5(MIL-T-5624); JP-8(MIL-T-83133); Jet A, Jet A-1, and Jet B conforming to P&WC S.B. 1244 or ASTM Spec D1655; and Chinese Jet Fuel No.3. See NOTE 1 for Emergency fuels.
<u>滑油 (发动机和减速器)</u>	P&WC PT6 发动机 No. 1001 号服务通报列表中允许的滑油牌号。

Oil (Engine and Gearbox)

P&WC PT6 Engine Service Bulletin No. 1001 list approved brand oils

发动机限制

	轴马力	N1 燃气涡轮转速	螺旋桨轴转速	最大许用涡轮间温 度℃
起飞 (5 分钟)	550	101.5	1900*	805
最大连续	550	101.5	1900*	805
启动瞬间 (2 秒)		102.6		1090
最大反桨	300	88.0	1825	805

* 见注 2

在低高度及低环境温度下，发动机起飞功率要比飞机审定时功率大。在这种条件下，扭矩表上的扭矩限制不得超越。

滑油温度	-40°F	最低启动温度
	-40°F to 210°F	低慢车
	50°F to 210°F	最大连续

Engine Limits

	Shaft horsepower	N1 Gas Generator Speed	Prop Shaft Speed	Max. Permissible Turbine Inter stage Temp. (Deg. C)
Takeoff(5 minutes)	550	101.5	1900*	805
Max Continuous	550	101.5	1900*	805
Start Transient(2 seconds)		102.6		1090
Max Reverse	300	88.0	1825	805

See NOTE 2

At low altitude and low ambient temperature the engines may produce more power at takeoff than that for which the airplane has been certificated. Under these conditions, the placarded torque meter limit shall not be exceeded.

Oil temperature:	-40°F	minimum starting
	-40°F to 210°F	low idle
	50°F to 210°F	max. continuous

螺旋桨和螺旋桨限制

具有反桨功能的四片桨叶螺旋桨（见注 3）

两个 Hartzell 公司带 D8990SK 螺旋桨叶的 HC-E4N-3N 桨毂

依据 Hawker Beechcraft 公司的规范 BS186497.

直径: 90.00 in. (明义尺寸) 最小允许可修复至 89.00 in.

(不允许进一步减少)

在 30in 站位的桨距设置

飞行慢车止动点	(见注 9)
反桨	-10° ±.5°
顺桨	85.8° ±.5
最小慢车速度	1100 RPM (见注 6)

Propeller and Propeller Limits

Reversing Four Bladed Propeller (See NOTE 3)
 2 Hartzell HC-E4N-3N hubs with D8990SK blades per
 Hawker Beechcraft Corporation Specification BS186497.
 Diameter: 90.00 in. (Normal) Minimum allowable for repair 89.00 in.
 (no further reduction permitted)
 Pitch Settings at 30 in. Sta:

Flight Idle Stop	(See NOTE 9)
Reverse	-10°±.5°
Feather	85.8°±.5
Minimum idle speed	1100 RPM (See NOTE6)

空速限制

	<u>CAS</u>	<u>CAS</u>	<u>IAS</u>
最大使用速度	260mph	(226 knots)	226 knots
机动速度	195mph	(169 knots)	169 knots
襟翼放下速度 (完全放下)	161mph	(140 knots)	148 knots
最大起落架操作速度			
放下	209mph	(182 knots)	182 knots
收起	189mph	(164 knots)	163 knots
最大起落架放下速度	209mph	(182 knots)	182 knots

Airspeed Limits

	<u>CAS</u>	<u>CAS</u>	<u>IAS</u>
Maximum operating speed	260mph	(226 knots)	226 knots
Maneuvering	195mph	(169 knots)	169 knots
Flap extended speed (full down)	161mph	(140 knots)	148 knots
Maximum landing gear operating speed			
Extension	209mph	(182 knots)	182 knots
Retraction	189mph	(164 knots)	163 knots
Maximum landing gear extended speed	209mph	(182 knots)	182 knots

重心范围 (起落架伸出)

(+152.0) 至 (+160.0) 在 10,100lb
 (+144.7)至 (+160.0) 在 7850 lb 或更小
 在给定的两点线性变化
 由于起落架收起导致的力矩变化-3825in.-lb。

<u>C.G. Range(Landing Gear Extended)</u>	(+152.0) to (+160.0) at 10,100lb (+144.7) to (+160.0) at 7850 lb or less Straight line variation between points given Moment change due to retracting landing gear -3825in.-lb.			
<u>空机重心范围</u>	无			
<u>Empty Wt. C.G. Range</u>	None			
<u>最大重量</u>	机坪	10,160lb.		
	起飞	10,100lb.		
	着陆	9,600lb		
<u>Maximum Weight</u>	Ramp:	10,160lb.		
	Takeoff	10,100lb.		
	Landing	9,600lb		
<u>座位数</u>	最大 13 人（包括在+129 处的 2 个人）。对乘客就坐见装载说明			
<u>No. of Seats</u>	Maximum 13 (including 2 at +129). See loading instructions for passenger loading			
<u>最大行李载重</u>	350lb.(+275)			
<u>(结构限制)</u>	350lb.(+70)（行李及电子设备）			
<u>Maximum Baggage</u>	350lb.(+275)			
<u>(Structural Limit)</u>	350lb.(+70)(Baggage and Avionics)			
<u>燃油容量</u>	油箱	容积 Gal	可用 Gal	力臂
	左右吊舱	61ea	61 ea	+131
	左右机翼	131 ea.	131 ea	+167
	不可用燃油数据见注 8。			
<u>Fuel Capacity</u>	Tank	Cap Gal	Usable Gal	Arm
	L & R Nacelle	61ea	61 ea	+131
	L & R Wing	131 ea.	131 ea	+167
	See NOTE 8 for data on unusable fuel.			
<u>滑油容量</u>	共 18.4 qt. (+101)			
	不可用滑油数据见注 8			
<u>Oil Capacity</u>	18.4 qt. total (+101)			
	See NOTE 8 for data on unusable oil			
<u>最大使用限制</u>	30,000ft 压力高度。			
<u>Max. Oper. Limit</u>	30,000ft. pressure Altitude.			
<u>操纵面使用范围</u>	襟翼	最大	44 °	
	副翼调整片（只在左侧）	上	15 °	下 15 °
	副翼	上	20 °	下 20 °

	升降舵调整片	上	10°	下	21°
	升降舵反伺服调整片	上	12°	下	8°
	升降舵	上	25°	下	15°
	方向舵调整片	右	30°	左	30°
	方向舵	右	24°	左	19°
<u>Control Surface Movement</u>	Wing flaps	Maximum	44°		
	Aileron tab(left only)	Up	15°	Down	15°
	Aileron	Up	20°	Down	20°
	Elevator tab	Up	10°	Down	21°
	Elevator tab anti-servo	Up	12°	Down	8°
	Elevator	Up	25°	Down	15°
	Rudder tab	Right	30°	Left	30°
	Rudder	Right	24°	Left	19°
<u>适用的系列号</u>	LJ-1847,LJ-1853 及以后。				
<u>Serial Nos. Eligible</u>	LJ-1847,LJ-1853 and after.				
<u>基准</u>	在机翼主（前）梁中心线前方 160 英寸处。				
<u>Datum</u>	Located 160 in. forward of wing main (forward) spar centerline.				
<u>调平方法</u>	机身左侧登机门前方或后方的两个外部螺钉。				
<u>Leveling Means</u>	2 external screws on left side of fuselage forward or aft of entrance door.				

设备 按适航规章（见合格审定基础）中所规定的基本设备必须安装在型号合格审定的飞机上，另外附加以下列表设备。

1. 如果依据 Beech 图纸 50-369100 和 50-361100 安装除冰设施的飞机，将安装 Safe Flight Corp., 的 P/N 168-2 或 P/N 168-3 的失速警告指示器。
2. C90 型安装了 Safe Flight Instrument Corp., 的 P/N 795-1 or P/N 795-9, 失速警告指示器。C90A, 和 C90GT, C90GTi 型安装了 Safe Flight Instrument Corp., 的 P/N 795-13 失速警告指示器。
3. FAA 认可的飞行手册和飞行员操作手册

Model C90A	P/N 90-590024-5	系列号从 LJ-1063 到 LJ-1137 和 LJ-1146
Model C90A	P/N 90-590024-23	系列号从 LJ-1138 到 LJ-1145, 从 LJ-1147 到 LJ-1287, 从 LJ-1289 到 LJ-1294, 从 LJ-1296 到 LJ-1299
Model C90A	P/N 90-590024-35	系列号 LJ-1288, LJ-1295, LJ-1302, 系列号 LJ-1303, 从 LJ-1305 到 LJ-1308, LJ-1311, LJ-1312, 从 LJ-1314 到 LJ-1316, LJ-1318, 从 LJ-1320 到 LJ-1352
Model C90A	P/N 90-590024-43	系列号 LJ-1300, LJ-1301, LJ-1304, LJ-1309,

Model C90A	P/N 90-590024-61	LJ-1310, LJ-1313, LJ-1317 和 LJ-1319 系列号 LJ-1367, LJ-1373, LJ-1377, LJ-1384, LJ-1386, LJ-1389, LJ-1394 和 LJ-1397, LJ-1403, LJ-1411, LJ-1425, LJ-1431 和 LJ-1435
Model C90A	P/N 90-590024-69	系列号从 LJ-1353 到 LJ-1726, 从 LJ-1728 到 LJ-1753 和 LJ-1755,除了 61 手册系列号 I.
Model C90GT	P/N 90-590024-111	系列号 LJ-1727, LJ-1754, 从 LJ-1756 到 LJ-1846 和从 LJ-1848 到 LJ-1852.
Model C90GTi	P/N 90-590024-163	系列号 LJ-1847, 从 LJ-1853 到 LJ-1963, LJ-1965, LJ-1967,从 LJ-1969 到 LJ-1971 和从 LJ-1973 到 LJ-1976.
Model C90GTi	P/N 90-590024-217	系列号 LJ-1964, LJ-1966, LJ-1968, LJ-1972, LJ-1977 及以后。

Equipment

The basic required equipment as prescribed in the applicable airworthiness regulation (see Certification Basis) must be installed in the aircraft for certification. In addition the following items of equipment are required;

1. Pre-stall warning indicator, Safe Flight Corp., P/N 168-2 or P/N 168-3, to be used if deicing equipment is installed per Beech Dwg. 50-369100 and 50-361100.
2. Pre-stall warning indicator, Safe Flight Instrument Corp., P/N 795-1 or P/N 795-9, for Model C90, P/N 795-13 for Models C90A, and C90GT, C90GTi.
3. FAA Approved Airplane Flight Manual or Pilot's Operating Handbook

Model C90A	P/N 90-590024-5	S/N LJ-1063 through LJ-1137 and LJ-1146
Model C90A	P/N 90-590024-23	S/N LJ-1138 through LJ-1145, LJ-1147 through LJ-1287, LJ-1289 through LJ-1294, LJ-1296 through LJ-1299
Model C90A	P/N 90-590024-35	S/N LJ-1288, LJ-1295, LJ-1302, S/N LJ-1303, LJ-1305 thru LJ-1308, LJ-1311, LJ-1312, LJ-1314 thru LJ-1316, LJ-1318, LJ-1320 thru LJ-1352
Model C90A	P/N 90-590024-43	S/N LJ-1300, LJ-1301, LJ-1304, LJ-1309, LJ-1310, LJ-1313, LJ-1317 and LJ-1319
Model C90A	P/N 90-590024-61	S/N LJ-1367, LJ-1373, LJ-1377, LJ-1384, LJ-1386, LJ-1389, LJ-1394 and LJ-1397, LJ-1403, LJ-1411, LJ-1425, LJ-1431 and LJ-1435
Model C90A	P/N 90-590024-69	S/N LJ-1353 through LJ-1726, LJ-1728 through LJ-1753 and LJ-1755, except 61 manual serial.
Model C90GT	P/N 90-590024-111	S/N LJ-1727, LJ-1754, LJ-1756 thru LJ-1846 and

		LJ-1848 thru LJ-1852.
Model C90GTi	P/N 90-590024-163	S/N LJ-1847, LJ-1853 thru LJ-1963, LJ-1965, LJ-1967, LJ-1969 thru LJ-1971 and LJ-1973 thru LJ-1976.
Model C90GTi	P/N 90-590024-217	S/N LJ-1964, LJ-1966, LJ-1968, LJ-1972, LJ-1977 and after.

- 注 1 应急状态下可使用 MIL-G-5572
在 65-90, 65-A90, 65-A90-1, 65-A90-2, 65-A90-3, 65-A90-4, B90, C90, C90A, C90GT, E90, H90 (T-44A) 和 GTi 型号使用 80/87, 91/98, 100/130 和 115/145 等级燃油时, 在每次大修间隔不超过 150 小时。在更换燃油型号时, 无需对系统中的剩余燃油进行清洗。
- NOTE1 Emergency use of MIL-G-5572
Grades 80/87, 91/98, 100/130 and 115/145 are permitted on Models 65-90, 65-A90, 65-A90-1, 65-A90-2, 65-A90-3, 65-A90-4, B90, C90, C90A, C90GT, E90, H90 (T-44A) and GTi for a total time period not to exceed 150 hours during any overhaul period. It is not necessary to purge the unused fuel from the system when switching fuel types.
- 注 2 对于 C90GT 和 C90GTi, 其最大螺旋桨轴超转限制是 110% (2090rpm)。1900rpm 定义为 100%螺旋桨转速, 该转速是稳定的工作转速限制。对燃气涡轮, 102.6%只允许 2 秒钟, 101.5%针对适用的环境温度和其他限制, 无限制时间, 100%对应的燃气涡轮转速为 37,500rpm。
- NOTE2 The maximum propeller shaft overspeed limit for the Model C90GT and C90GTi is 110 percent (2090rpm). 100 percent propeller shaft speed is defined as 1900rpm and is the normal steady state operating limit. Gas generator speeds up to 102.6 percent are permissible for 2 second and to 101.5 percent for unlimited periods subject to applicable temperature and other limits, 100 percent gas generator speed is defined as 37,500rpm.
- 注 3 依据型号, 在列表中的反桨将毂与螺旋桨的搭配是可接受的。在同一将毂上混用不同的螺旋桨件好事允许的。
- NOTE3 By model, any combination of reversing hub and blade part numbers listed is acceptable. It is permissible to mix blade part numbers on the same hub.
- 注 4 在螺旋桨转速 2000rpm 时的飞行慢车指示扭矩应修正为海平面标准日 600±60 英尺磅。当装有辅助飞行慢车止动位, 辅助飞行慢车止动位应比飞行慢车位的螺旋桨转速高 210±40 rpm, 此时燃气发生器速度为 70%。
- NOTE4 Flight idle at 2000 propeller rpm shall be an indicated 600±60 ft. -lb. torque corrected for sea level standard day. Secondary flight idle stop when installed shall be 210±40 propeller rpm higher than flight idle stop with a gas generator speed of 70%.
- 注 5 飞行慢车螺旋桨低桨距止动器设置为使得在2000rpm时发动机扭矩为608±40英尺磅。扭矩修正到海平面标准日状况。地面慢车低桨距止动位设置为使得在58%到60% N1, 螺旋桨转速不少于1100 rpm。

- NOTE5 Flight idle propeller low pitch stop is set so that at 2000 rpm the engine torque is 608±40 ft. lb. Torque corrected to sea level standard day conditions. Ground idle low pitch stop is set so that at 58% to 60% N1, prop rpm is not less than 1100 rpm.
- 注6 在飞行手册和维护手册，对螺旋桨操作提到以下警告：
“地面，在受限制的转速范围长时间工作，将对螺旋桨产生高的螺旋桨载荷，对螺旋桨产生疲劳损伤。这种损伤将降低螺旋桨的疲劳寿命，并产生螺旋桨故障，并使飞机失去控制。”
- NOTE6 The following Warning concerning propeller operation is in both the AFM and Maintenance Manual:
“Stabilized ground operation within the propeller restricted RPM range can generate high propeller stresses and result in fatigue damage to the propeller. This damage can lead to a reduced propeller fatigue life, propeller, propeller failure and loss of control of the aircraft.”
- 注7 合并了图纸更改单 MOD005147-1 的 C90A 型飞机，他的最大停机坪重量是 10,059lbs., 最大起飞重量 9,999lbs., 最大着陆重量 9,600lbs. 在飞行手册和飞行员操作手册中图纸更改单 MOD005147-1 要求加上补充页 PN 90-590024-81 和一个重量使用限制卡 MOD005147-3, 符合上述要求的飞机系列号是从 LJ-1469 到 LJ-1726, 从 LJ-1728 到 LJ-1753, 和 LJ-1755。
- NOTE7 Model C90A Airplane which incorporate MOD Drawing MOD005147-1 are limited to a maximum ramp weight of 10,059lbs., a maximum takeoff weight of 9,999 lbs., and a maximum landing weight of 9,600 lbs. MOD Drawing MOD005147-1 requires an AFM and POH supplement PN 90-590024-81 and an operating weight limitation placard, MOD005147-3. Eligible Serial Numbers are LJ-1469 through LJ-1726, LJ-1728 through LJ-1753, and LJ-1755.
- 注8 在审定之处，在每架飞机上必须提供：当前重量和平衡数据；载重信息：包括在空机重量中的设备清单。型号认可的空机重量必须包含不可用燃油和不可用滑油。
基本空机重量必须包含不可用燃油和发动机滑油（包括滑油量和无法放掉的燃油，如果适用的化）如以下列表。
对于 65-B80(LD-468 以上)型号是 96 lbs.燃油 (+168 in.) 和 81 lbs.滑油(+131 in.)，对于 C90, C90A, C90GT, H90(T-44A)和 C90GTi. 是 40 lbs.燃油(+140 in.) 56 lbs.的滑油(+101 in.) 对于 E90 型号是 24 lbs.燃油(+140 in.) 和 16 lbs.滑油(+101 in.)。
- NOTE8 At the time of original certification, the following must be provided for each aircraft: current weight and balance data; loading information; list of equipment included in the empty weight.
The certification Empty Weight must include unusable fuel and unusable (or unusable (undrainable)) oil as show below:
The Basic Empty Weight must include unusable fuel and engine oil (include capacity oil and undrainable, where applicable) as shown below:
Fuel 96 lbs.(+168 in.) and oil 81 lbs.(+131 in.) for Model 65-B80(LD-468 and up). Fuel 24 lbs.(+140 in.) and oil 16 lbs.(+101 in.) for Model C90, C90A, C90GT, H90(T-44A) and C90GTi. Fuel 40 lbs.(+140 in.) and oil 56 lbs.(+101 in.) for Model E90.
- 注9 飞行慢车螺旋桨低桨矩止动点设置在 1800rpm，发动机扭矩设置在海平面标准条件的 545-0/+40ft.lb.

- NOTE9 Flight idle propeller low pitch stop is set so that at 1800rpm the engine torque is 545 -0/+40 ft. lb. torque corrected to sea level standard day conditions.
- 注 10 公司更名的有效期是从 1996 年 4 月 15 日。在 Raytheon Aircraft Company 名字下生产的 C90A 和 C90GT: 系列号是从 LJ-1437 到 LJ-1826。
- NOTE10 Company name change effective April 15, 1996. The following Serial Numbers are manufactured under the name of Raytheon Aircraft Company: C90A and G90GT: LJ-1437 through LJ-1826.
- 注 11 公司更名的有效期是从 2007 年 3 月 26 日。下面系列号是在 Hawker Beechcraft Corporation 名字下生产的: LJ-1827 和以后的。
- NOTE11 Company name change effective 3-26-07. The following serial numbers are manufactured under the name of Hawker Beechcraft Corporation: LJ-1827 and after.