

型号认可证数据单
THE VALIDATION DATA SHEET

编号/No: VTC0291A
版次/Revision:0
型号/Type:

UH-12E

批准人/Approved By:YANG ZHENMEI


职务/Title: Deputy Director General of
CAAC-AAD
日期/Date:2014-09-26

本数据单是型号认可证(编号: VTC0291A)的组成部分, 它规定了符合中国民用航空局的适航要求所颁发此型号认可证的产品状态和限制。

This data sheet, which is part of Validation of Type Certificate (No: VTC0291A) , prescribes condition and limitation under which the product for which the type certificate was issued meets the airworthiness requirements of the Chinese Civil Aviation Regulation.

型号认可证持有人/Validation of Type Certificate Holder:

Siam Hiller Holdings, Inc.

925 M. Street
Firebaugh, California , USA.93622-2234

有效页清单/List of effective pages :

Pages	1	2	3	4	5	6	7	8	9	10	11								
Revision	0	0	0	0	0	0	0	0	0	0	0								

第1部分 概述
SECTION 1 GENERAL

1. CAAC认可数据单和型号合格证数据单

CAAC Validation Data Sheet and Type Certificate Data Sheet

	中国民用航空局认可数据单号、版次和颁发日期 CAAC Validation Data Sheet Number, Revision and issuance date	型号合格证数据单号、版次和颁发日期 Type Certificate Data Sheet Number, Revision and issuance date
当前 Current	VTC0291A , Rev. 0, 2014-9-26	FAA TCDS 4H11, Revision 18, 06 January, 2010
历史 History		

2. 类别/Category: 正常类旋翼航空器/Normal Category Rotorcraft

3. 认可当局/Validation Authority: 中国民用航空局/CAAC

审定当局/Certifying Authority: 美国联邦航空局/FAA

4. 型号合格证持有人/Type Certificate Holder: Siam Hiller Holdings, Inc.
925 M. Street
Firebaugh, California , USA.93622-2234

5.型号合格证转让记录/Type Certification Ownership Records:

Hiller Aircraft Corporation transferred TC 4H11 to
Fairchild Industries, Inc. on 10/2/1959

Fairchild Industries, Inc. transferred TC 4H11 to
Heli-Parts, Inc. on 12/28/1972

Heli-Parts, Inc. transferred TC 4H11 to
Hiller Aviation on 12/28/1972

Hiller Aviation transferred TC 4H11 to
Rogerson Aircraft Corporation on 6/26/1984

Rogerson Aircraft Corporation transferred TC 4H11 to
Hiller Helicopters, a wholly owned subsidiary of Rogerson Aircraft Corporation on
6/29/1984

Hiller Helicopters, a wholly owned subsidiary of Rogerson Aircraft Corporation
transferred TC 4H11 to Rogerson Hiller Corporation on 11/14/1985

Rogerson Hiller Corporation transferred TC 4H11 to
Siam Hiller Holdings, Inc. on 7/14/1994

6. CAAC Special Requirements:

交付到中国的每一架 UH-12E 旋翼航空器须满足下述特殊要求/ Each of UH-12E rotorcraft delivered to China shall meet the following special requirements::

(1) 必须满足中国民用航空局的运行要求（例如标记标牌的中文要求）。/ The operational requirements of CAAC must be met (e.g. Chinese language requirements for markings and placards).

第2部分 SECTION 2

UH-12E

I. 型别/Model UH-12E, FAA 于 1959 年 1 月 6 日批准, CAAC 于 2014 年 9 月 26 日认可/Approved by FAA on January 6, 1959 ,Validated by CAAC on Sep 26,2014

型号设计定义/ Definition of Type design Hiller 公司的工程报告 2009-02 主图纸目录 (2009 年 11 月 10 日版) /Hiller Aircraft Corporation Engineering Report No.2009-02 Master Drawing List(Rev. Nov.10 2009)

发动机/Engine Lycoming VO-540-C2A. 关于发动机的详情参见发动机的型号认可数据单 VTCDS0280E /See Engine CAAC VTCDS0280E for more details.

燃油/Fuel 100/100 低铅航空汽油/100/100 Low Lead minimum grade aviation gasoline

所有运行条件下的发动机关限制 最大转速/Maximum rpm: 3200 (305 hp).

/Engine Limits for all Operations 最大压力/Maximum manifold pressure:
25.2 in. Hg

旋翼转速限制和发动机运转速度/Rotor Limits and Engine Operating Speeds

无动力/Power Off (Rotor Tach.)	有动力/Power On (Engine Tach.)
最大 395rpm /Maximum 395 rpm	最大 3200 rpm /Maximum 3200 rpm
最小 314 rpm /Minimum 314 rpm	最小(海平面 5000 英尺以下)2900 rpm /Minimum (sea level to 5000 ft.) 2900 rpm 5000 英尺以上时, 高度每增加 1000 英尺, 最小转速增加 200rpm /(Above 5000 ft, increase minimum rpm by 20 rpm for each additional 1000 ft of altitude

(对于安装了件号为 53200-03 旋翼桨叶的 UH-12E 参考注 13)/
(See NOTE 13 for UH-12E with Main Rotor Blades P/N 53200-03)

空速限制/Airspeed Limits

构型/Configuration	不可超越速度/V _{NE} (IAS)
滑撬起落架/Skid Gear	96 mph (83 knots)
浮筒起落架/Float Gear	86 mph (74 knots)

上述空速限制适用于海平面 5000 英尺以下。高度大于 5000 英尺时, 每增加 1000 英尺高度, V_{NE} 减少 2.5mph (2.2 节)。对于安装了其它设备时的限制, 参见局方批准的飞行手册 (4 座构型见注 6) /The above airspeed applies from sea level to 5000 ft. Decrease V_{NE} 2.5 mph (2.2 knots) per 1000 ft of altitude above 5000 ft. For limits with accessories installed, see the FAA-Approved Rotorcraft Flight Manual.
(See NOTE 6 for 4-place configurations)

高度限制/Altitude Limits

见批准的飞行手册
/See FAA-Approved Rotorcraft Flight Manual.

重心范围 (纵向)
/C.G. Range (Longitudinal)

构型/Configuration	
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滑撬起落架/Skid Gear	Sta. (79.5) to (84.8)
浮筒起落架/Float Gear	Sta. (81.0) to (84.8)

对于带有其它设备的重心范围见局方批准的飞行手册 /For range with accessories installed see the FAA-Approved Rotorcraft Flight Manual.
(4 座构型情况见注 6) /(See NOTE 6 for 4-place configuration)
(装有件号为 53200-03 的旋翼桨叶的 UH-12E 见注 13) /(See NOTE 13 for UH-12E with Main Rotor Blades P/N 53200-03)

基准/Datum	尾梁和机身连接处上固定面前方 107.25 英寸 /107.25 in. fwd of tail boom-fuselage upper mounting face
重心范围(水平) /C.G. Range (Lateral)	直升机中心线左侧 4.83 英寸 /Left of helicopter center line, 4.83 in. 直升机中心线右侧 1.85 英寸 /Right of helicopter center line, 1.85 in. (对于安装有件号为 53200-03 的旋翼桨叶的 UH-12E 直升机参考注 13) /(See NOTE 13 for UH-12E with Main Rotor Blades P/N 53200-03)
定水平/Leveling Means	座椅下凸缘上表面/Top face of flanges under seat
最大重量/Maximum Weight	2750 lb (对于增加最大重量情况参考注 7 和注 13) /(See NOTE 7 or NOTE 13 for increased maximum weight)
座椅数量/Number of Seats	3 (sta.53) (四座构型参考注 6)/(See NOTE 6 for 4-place configuration)
最大行李/Maximum Baggage	参考局方批准的飞行手册中装载说明/See loading instructions in FAA-Approved Rotorcraft Flight Manual
燃油量/Fuel Capacity	总共 46、66 或 86 加仑。/Total 46, 66, or 86 gal. 一个主燃油箱 46 加仑(站位 82.9)。/One main tank 46 gal. (sta.82.9). 参考 Hiller 服务通告 2008 安装 1 个或 2 个辅助油箱, 每个 20 加仑 (站位 85.8) /One or two auxiliary tanks installed per Hiller Service Bulletin 2008, 20 gal. each (85.8). 参考注 1 关于不可用的燃油数据/See NOTE 1 for unusable fuel data.
滑油量/Oil Capacity	9.2 夸托/9.2 qt (sta.94), 或者安装有辅助燃油箱时 12.3 夸托/or 12.3 qt (sta.94) with auxiliary fuel tanks installed.
其它运行限制/Other Limitations	见局方批准的飞行手册/See FAA-Approved Rotorcraft Flight Operating Manual
旋翼桨叶和操纵 /Rotor Blade and Control	相对于主轴测量/Measured with respect to the mast 注: 当主轴垂直时, 直升机抬头 1°/Note: When the mast is vertical, the helicopter is 1° nose up)
主旋翼桨叶/Main Blades: 总距/Collective Pitch	最低设定为+1.0°, 总行程 10.5° (以桨根固定板为准测量)。(最低设定是按防止自转时发生超转确定的) /Low setting +1.0°, total travel 10.5° (Measured at Retention Plate) (Low setting is determined as the lowest which will preclude overspeeding in autorotation) 操纵旋翼/Control rotor $\pm 12^\circ$, 主旋翼/Main Rotor $\pm 9^\circ$

翘板式/Teetering

旋转斜盘/Wobble Plate:

横向/Lateral

7.7° to 8.3°

纵向/Longitudinal

前/Fwd. 8.0° to 8.5°, 后/Aft 9.0° to 9.2°

控制桨叶/Control Blades:

中立/Neutral

迎角 $9^{\circ} \pm 0.2^{\circ}$ (旋翼桨毂和旋转斜盘垂直于主轴) / $9^{\circ} \pm 0.2^{\circ}$ incidence (Rotor hub and wobble plate perpendicular to mast)

尾桨桨叶挥舞角

$+17^{\circ}$ to -17°

/Tail Rotor Blade

Flapping:

尾桨桨距调节/Tail Rotor Pitch Rigging

尾桨变速箱件号/T.R. Gearbox P/N	行程(度)/Travel (degrees)
25200	+15 to -3
25200-3 or -5	+16 to -4
25300-5	+20 to -4

水平安定面设定/Horizontal Stabilizer Setting

<u>Configuration(With helicopter level)</u>	
滑撬起落架/Skid Gear	0°
浮筒起落架/Float Gear	-10°

浮筒压差/Float Pressure differential

最大值/8.0 psi maximum

(对于4座构型参考注6)/(See NOTE 6 for 4-place configuration)

序列号/ Serial Nos. Eligible

UH-12E 型号/Model UH-12E; 942, 954, 2001 through 2166, 2172 through 2229, 2233 through 2241, 2246 through 2248, 2253 through 2255, 2257 through 2282, 2286 through 2292, 2294 through 2306, 2309 through 2499, 2518, 5001 and up.

FAA 审定基础/FAA Certification basis

1956年12月20日的CAR 6, 包含6-1和6-2修正案, 申请人确认自愿符合修正案6-3。/CAR 6 dated December 20, 1956, including Amendments 6-1 and 6-2, Voluntary compliance with Amendment 6-3 has been established. Type Certificate 4H11 issued January 6, 1959. Date of Application for Type Certificate December 13, 1957.

中国民用航空局认可审定基础/
CAAC Validation Basis:

1. 美国联邦航空局的合格审定基础: / CAAC accept FAA Certification Basis as the Validation Basis.
2. CAAC 的附加技术条件/ CAAC Additional Technical Conditions: 参见问题纪要 G001 / Refer to Issue Paper G001 (I) 中文标识要求 / As per CCAR 21.29(e), All literal markings and placards on an aircraft used for advising, warning, or notifying passengers must be at least in Chinese; all literal markings and placards on an airplane used for indicating passengers or outboard rescue personnel of the

location of emergency exits and doors and their means of opening must be at least in Chinese; and all the operation instructions of airplane emergency equipment that may be used by passengers must be at least in Chinese.

生产基础/Production Basis

无/None. 2003 年 4 月 23 日后在首次颁发每架直升机的适航证前, FAA 代表必须对工艺、材料、与批准设计资料的符合性进行详细检查并且核实飞行特性。如果申请标准适航证或者有申请人想要依据 14 CFR § 21.183 (d)生产新航空器, 申请人正在依据它人的型号合格证制造、组装新航空器, 其必须获得型号合格证持有人的书面许可。/Before original airworthiness certification of each aircraft manufactured subsequent to April 23, 2003, an FAA representative must perform a detailed inspection for workmanship, materials, conformity with the approved technical data, and a check of the flight characteristics. In the event of an application for a standard airworthiness certificate or, if an applicant intends to produce new aircraft under 14 CFR § 21.183 (d), and the applicant is manufacturing, building, or assembling to another person's type certificate, the applicant must provide written evidence of permission from the type certificate holder. Conduct of such activity without written evidence of permission may be a violation of 49 U.S.C. § 44704(a)(3).

设备/Equipment

适航规章(见审定基础)中要求的基本设备必须安装到直升机上。Hiller 报告 59-30“UH-12E 主设备清单”包含了所有必需设备和可选安装设备的清单, 并且已获得局方的批准。(4 座构型参考注 6)
/The basic required equipment as prescribed in the applicable airworthiness regulations (See Certification basis) must be installed in the helicopter for certification. Hiller Report 59-30, "Model UH-12E Master Equipment List," contains a list of all required equipment that must be installed as well as optional equipment installations approved by FAA. (See NOTE 6 for 4-place configuration.)

注 1/NOTE 1. 当前的载重平衡报告包含了含在审定空重中的设备清单以及装载说明, 因此必须放置在每架直升机上(除非营运人另有经批准的载重控制系统)。必要时的配重必须安装局方批准的飞行手册中装载说明来进行。/Current weight and balance report, including list of equipment included in certificated empty weight, and loading instructions must be in each helicopter at the time of original certification and at all times thereafter (except in the case of operators having an approved weight control system). Ballast, when necessary, must be carried in accordance with Loading Instructions in the Approved Rotorcraft Flight Manual.

显示的燃油量和滑油量是高于可使用燃油量和滑油量的总量。主燃油箱包含了 0.3 加仑的不可用燃油量, 在所有飞行姿态均不能使用, 必须包含在空重中。可选辅助燃油箱中的不可用燃油量可以忽略不计。/Fuel and Oil capacities as indicated are total tank capacities over and above "Trapped Fuel and Oil." The main fuel tank capacity includes "Unusable" fuel of 0.3 gallon, which cannot be used safely in all flight attitudes, and which must be included in the empty weight. Unusable fuel in the optional auxiliary fuel tanks is negligible.

注 2/NOTE 2. 下列标牌必须放置在飞行员的检查单上: “该直升机必须在符合局方批准飞行手册中规定的限制条件下运行。”其它标牌, 见飞行手册。/The following placard must be installed on the Pilot's Checklist:

"This Helicopter must be operated in compliance with the operating limitations specified in the FAA-Approved Rotorcraft Flight Manual."

For additional placards, see the Approved Flight Manual.

注 3/NOTE 3. 关键件的时寿时间列在下表中。在未获得局方工程批准时不得擅自增加时寿时间。/The retirement times of critical parts are listed in the following tables. These values of retirement or service life cannot be increased without FAA Engineering approval.

UH-12E 系列直升机各型号或构型间互换时寿件时，必须将其限制到最低的使用时间内。如果此前的使用经历没有标明其安装型号或构型，必须使用所有型号或构型的最低使用时间。已到时寿限制被拆下的组件必须销毁或者作上标记阻止其再次被使用。/Life limited components interchanged between UH-12 series models or configurations must be restricted to the lowest service life indicated for the models or configurations that the individual part has been used on. If previous service history does not identify the model or configuration, the lowest service life of all possible models or configurations must be used. Life limited components removed at retirement are to be destroyed or conspicuously marked to prevent inadvertent return to service.

名称/NOMENCLATURE	UH-12E 时寿件/ MODEL UH-12E FINITE LIFE COMPONENT	
	件号/PART NO.	更换小时要求 /REPLACEMENT PERIOD HOURS
<u>Rotors and Drives</u>		
Blade Assy., main rotor (Parsons)	2253-1101-04	2500 NOTE A
Blade Assy., main rotor (Parsons)	2253-1101-03	2500
Blade Assy., main rotor	53200-03	6670 NOTE H
Shaft Assy., output speed decreaser	25202-5	5790
Plate set, main rotor tension-torsion (component of 51430-1 and 51430-3 assembly)	51430-1, -3,-5	3350
Hub Assy., main rotor (Service Bulletin No. 2014 not complied with)	51437-9	1540
Hub Assy., main rotor (Service Bulletin No. 2014 complied with)	51437-9	2500 NOTE B
Hub Assy., main rotor	51437-11,-14 -901,- 911	2500
Pin, main rotor outboard tension torsion	51452,-1	643
Fork, main rotor blade root	52110-3	2500
Drag strut, main rotor blade	52120, -5	2500
Drag strut, main rotor blade	52120-7	2500 NOTE H
Bar Assy., tail rotor tension-torsion	55054, -1	12500
Blade Assy., tail rotor (3-place model)	55073 (All Dash Numbers)	5400
Blade Assy., tail rotor (4-place model)	55073 (All Dash Numbers)	3240
Yoke, Tail Rotor	55046 (All Dash Numbers)	2500 NOTE J
<u>Power Plant</u>		
Snubber Assy., engine, longitudinal	63192-7, -11	4650
Snubber Assy., engine, lateral	63192-5, -9	4650
Bracket, engine snubber	63197	6160
Bracket, engine snubber	63197-5	9850
Attaching bolts, engine snubber (used to attach P/N 63197 bracket to engine deck and	AN3	600

snubber)		
Gimbal outer engine mounting	63309,-3,-9,-11	6300
Gimbal outer engine mounting	63309,-3,-9,-11	5480 Note H
One way clutch (Borg-Warner)	X132591	375 Note C
One way clutch (Borg-Warner)	X132815	375 Note C

Collective and Cyclic Controls

Push rod Assy., Collective pitch incidence	31113-13 or 31333	13400 Note H
Flyweight collective pitch ballast	31197-3	2500
Bellcrank, collective control	31318	15000
Tube, collective control	31319,-3	23500
Shaft, dual collective control	31321	10400
Block Assy., collective stick	31344, -3, -5,-7,-9	6400 NOTE I
Push rod, collective control	31365	8150 Note H
Arm, collective pitch blade incidence	31389	14500 Note H
Incidence Arm	31113-13	13400 Note H
(Old style-Incs.Fafnir or Shafer Rod Ends)		
Incidence Arm	31333 WA7-75083-S1.	13400 Note H
(Heim Rod Ends on Condition)		
Arm Assy., collective and throttle controls (4-place model)	31403	21500
Scissor, cyclic control, lower	34141, -5	275
Spar and Blade Assy., Control Rotor(all faired Assys.)		Note D
Fairing Blade Assy., Control Rotor	36003, 36129, 36129-25	2500 Note F
Trunion, Control Rotor Hub	36116-4	5150 Note E
Cuff Assy., Control Rotor (used with faired Assy.)	36124	5150 Note G
Cuff Assy., Control Rotor (used with unfaired assy.)	36124	6860 Note G
Cuff Assy., Control Rotor (used with unfaired assy.)	36124	5550 Note H,G

NOTE A - The replacement period for blade assemblies S/N 4261 and subsequent is 6670 hours. The replacement period for blade assemblies S/N 3396 through S/N 4260 is extended to 6670 hours if these blades are returned to the manufacturer for installation of rolled-thread anti-node bar assembly P/N 2253-1124 before 2500 hours time in service.

NOTE B - The 2500 hour service life total includes service time prior to compliance with Hiller Aviation Service Bulletin No. 2014.

NOTE C - Unlimited life after incorporation of Hiller Aviation Service Bulletin No. 2027A.

NOTE D - Any unfaired control rotor blade assembly with an "R" after the part number (reworked per Hiller Aviation Service Bulletin 36-1, Revision 2) must be replaced upon attaining 2500 additional operating hours after rework.

NOTE E - 5150 hour limitation applicable to trunions used with control rotor fairing and blade assy. P/N 36003, 36129, or 36129-25.

NOTE F - If any of these part numbers are reworked in accordance with Hiller Aviation Service Bulletin 36-1, Revision 3, the part must still be replaced upon attaining a total time of 2500 hours regardless of the replacement period shown in Hiller Aviation Service Bulletin 36-1, Revision 3.

NOTE G - All control rotor cuff assemblies installed with control rotor blade assemblies (both faired and unfaired) with an "R" after the part number (reworked per Hiller Aviation Service Bulletin 36-1, Revision 3) must be replaced upon attaining the original retirement life, but not to exceed 2500 additional hours after rework.

NOTE H - These new or revised replacement periods are for the "Hiller Model UH-12E with new main rotor blades 53200-03." (See NOTE 13 for installation of main rotor blade 53200-03.)

NOTE I - Part number 31344 Collective Stick block assembly is a matched set consisting of 31344-3 and 31344-5. Part number 31344-7 Collective Stick block assembly is a matched set consisting of 31344-5 and 31344-9. Serial number for both lower and upper blocks must be the same.

NOTE J - Life limited applies only when increased pitch style tail rotor gearbox part number 25300-5 tail rotor speed decrease is used.

注 4/NOTE 4. 直升机必须按照当前 Hiller Aircraft Corporation 持续适航说明文件（检查向导）、修理手册、使用和大修手册进行维护。/These helicopters must be serviced and maintained in conformance with current Hiller Aircraft Corporation Instructions for Continued Airworthiness (Inspection Guide), repair handbook, and service and overhaul manuals for pertinent model.

注 5/NOTE 5. 保留/Reserved.

注 6/NOTE 6. UH-12E 型号 4 座构型和 4 座改装包，图纸号 10044，按照 Hiller 服务通告 2010 改装。下列限制适用：/Model UH-12E 4-Place Configuration and 4-Place Modification Kit, Hiller Dwg. No. 10044, installed per Hiller Service Bulletin No. 2010. The following additional limitations apply: (For limits with accessories installed see FAA-Approved Rotorcraft Flight Manual.)

(a)	空速限制/Airspeed limits	构型/Configuration	不可超越速度/ V_{NE} (IAS)
		滑橇起落架/Skid Gear	95 mph (82 knots)
		浮筒起落架/Float Gear	86 mph (74 knots)
			Decrease V_{NE} 2.5 mph (2.2 knots) per 1000 ft above 5000 ft
(b)	重心范围/C.G. range	滑橇起落架/Skid Gear	Sta. (79.5) to (84.8)
	纵向/(Longitudinal)	浮筒起落架/Float Gear	Sta. (81.0) to (84.8)
(c)	座椅数量/No. seats	1 (25), 3 (53)	
(d)	Horizontal stabilizer P/N 37027	+3° (Measured with helicopter level)	
(e)	设备 Equipment	The basic required equipment as prescribed in the applicable airworthiness regulations (See Certification Basis) must be installed in the helicopter for certification. Hiller Report 60-60 "Model UH-12E (4-place) Master Equipment List" contains a list of all required equipment that must be installed as well as optional equipment installations approved by the FAA.	

注 7/NOTE 7. UH-12E 3 座和 4 座构型在安装了 Hiller 服务通告 2031 中规定的设备后，最大重量 2800 磅。运行限制见局方批准的飞行手册。/UH-12E, 3- and 4-place configurations are eligible at

2800 pounds maximum weight when items specified by Hiller Service Bulletin No. 2031 are installed. See FAA-Approved Rotorcraft Flight Manual for operation limitations.

注 8/NOTE 8. 保留/Reserved.

注 9/NOTE 9. 保留/Reserved.

注 10/NOTE 10 保留/Reserved.

注 11/Note 11 保留/Reserved.

注 12/Note 12 保留/Reserved.

注 13/NOTE 13. 可按照 Hiller 服务通告 10-2 和 51-3 完成主旋翼桨叶 53200-03 的安装。改装后需要局方批准的针对安装有件号为 53200-03 旋翼桨叶的旋翼航空器飞行手册。下列限制适用。/The main Rotor Blades P/N 53200-03 may be installed on Hiller Model UH-12E by accomplishment of Hiller Service Bulletins No. 10-2 and No. 51-3. FAA-Approved Rotorcraft Flight Manual "Hiller Model UH-12E Helicopter with 53200-03 Main Rotor Blades" is required. The following limitations apply. (For limits with accessories installed, see FAA-Approved Rotorcraft Flight Manual "Hiller UH-12E Helicopter with 53200-03 Main Rotor Blades".)

(a)	旋翼限制和发动机 运转速度/Rotor limits and engine operating speeds	<u>无动力/Power Off (Rotor Tach)</u>	<u>有动力/Power On (Engine Tach)</u>
		Maximum 395 rpm Maximum 314 rpm	Maximum 3200 rpm Minimum (S.L. to 10,000 ft.) 3000 rpm (Above 10,000 ft increase minimum rpm by 20 rpm for each additional 1000 ft of altitude)
(b)	空速限制/Airspeed limits	<u>构型/Configurations</u>	<u>不可超越速度/V_{NE} (IAS)</u>
		Skid Gear 2801 to 3100 pounds	75 mph (65 knots) decrease V _{NE} 2.5 mph (2.2 knots) per 1000 ft above 5000 ft
(c)	高度限制/Altitude limits	<u>构型 Configuration</u>	
		Skid Gear - 2800 lb Skid Gear-2801 to 3100 lb	S.L. to 15,000 feet S.L. to 7400 feet
(d)	重心范围/C.G. Range	<u>构型 Configuration</u>	
		Skid Gear 2800 pounds	Left of helicopter center line, 3.75 in. Right of helicopter center line, 1.85 in.
		3100 pounds	Left of helicopter center line, 3.40 in. Right of helicopter center line, 1.85 in.
		The variation in left lateral C.G. is a straight line interpolation between 2801 pounds and 3100 pounds.	
(e)	重心范围/C.G. Range 纵向/(Longitudinal)	<u>构型 Configuration</u>	
		Skid Gear 10/11	Forward: Sta. (79.5) to 7000 feet;

Sta. (80.1) above 7000 feet.
Aft: Sta. (84.8)

(f) 最大重量/Maximum 3100 pounds
Weight

注 14/NOTE 14.保留/Reserved.

...END...