

	2	2	2	2	2	2	2	2	2	2	2	2
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发动机限制 在所有使用情况下, 2400 转/分钟

但当经赛斯纳 MK172-72-01 号改装后(见注释 3),
在所有使用情况下, 2700 转/分钟

Engine Limits For all operations, 2400 RPM

When modified by Cessna Modification Kit MK172-72-01(See Note 3)
For all operations, 2,700RPM

螺旋桨 (a) 麦可利型号: IC235/LFA7570

(b) 桨毂: 图号 0550236

但当经赛斯纳 MK172-72-01 号改装后(见注释 3),

(a) 麦可利型号: 1A170E/JHA7660

(b) 桨毂: 图号 0550236

Propeller (a) McCauley Model: IC235/LFA7570

(b) Spinner: Drawing No.0550236.

When modified by Cessna Modification Kit MK172-72-01(See Note 3)

(a) McCauley Model: 1A170E/JHA7660

(b) Spinner: Drawing No.0550236.

螺旋桨限制 在满油门下的静态转速: 不超过 2165 转/分钟, 不低于 2065 转/分钟
不得有额外容差

直径: 不超过 75 英寸, 不低于 74 英寸

但当经赛斯纳 MK172-72-01 号改装后(见注释 3),

在满油门下的静态转速: 不超过 2400 转/分钟, 不低于 2300 转/分钟
不得有额外容差

直径: 不超过 76 英寸, 不低于 75 英寸

Propeller Limits: Static RPM at full throttle: Not over 2165;Not Under 2065
No Additional Tolerance Permitted
Diameter: not over 75 in., not under 74 in

When modified by Cessna Modification Kit MK172-72-01(See Note 3)

Static RPM at full throttle: Not over 2400;Not Under 2300

No Additional Tolerance Permitted

Diameter: not over 76 in., not under 75 in

空速限制

机动速度	99 节 (指示空速)	(97 节 (校正空速))
最大结构巡航速度	129 节 (指示空速)	(126 节 (校正空速))
不可超越速度	163 节 (指示空速)	(160 节 (校正空速))
襟翼展态速度	85 节 (指示空速)	(84 节 (校正空速))

但当经赛斯纳 MK172-72-01 号改装后(见注释 3),

机动速度	105 节 (指示空速)	(102 节 (校正空速))
最大结构巡航速度	129 节 (指示空速)	(126 节 (校正空速))
不可超越速度	163 节 (指示空速)	(160 节 (校正空速))
襟翼展态速度	85 节 (指示空速)	(84 节 (校正空速))

Airspeed Limits

Maneuvering	99 Knots IAS	(97 Knots CAS)
Max. Structural Cruising	129 Knots IAS	(126 Knots CAS)
Never Exceed	163 Knots IAS	(160 Knots CAS)
Flaps Extended	85 Knots IAS	(84Knots CAS)

When modified by Cessna Modification Kit MK172-72-01(See Note 3)

Maneuvering	105 Knots IAS	(102 Knots CAS)
Max. Structural Cruising	129 Knots IAS	(126 Knots CAS)
Never Exceed	163 Knots IAS	(160 Knots CAS)
Flaps Extended	85 Knots IAS	(84Knots CAS)

重心范围

正常类

- (1) 后限: 在不大于 2450 磅时, 基准之后的 47.3 英寸处
 (2) 前限: 在 2450 磅时, 基准之后的 40.0 英寸处; 在 1950 磅时, 基准之后的 35.0 英寸处; 在上述两者重量之间的重量时, 线性变化; 不大于 1950 磅时, 基准之后的 35.0 英寸处。

实用类

- (1) 后限: 在不大于 2100 磅时, 基准之后的 40.5 英寸处
 (2) 前限: 在 2100 磅时, 基准之后的 36.5 英寸处; 在 1950 磅时, 基准之后的 35.0 英寸处; 在上述两者重量之间的重量时, 线性变化; 不大于 1950 磅时, 基准之后的 35.0 英寸处。

但当经赛斯纳 MK172-72-01 号改装后(见注释 3),

正常类

- (1) 后限: 在不大于 2550 磅时, 基准之后的 47.3 英寸处
 (2) 前限: 在 2550 磅时, 基准之后的 41.0 英寸处; 在 1950 磅时, 基准之后的 35.0 英寸处; 在上述两者重量之间的重量时, 线性变化; 不大于 1950 磅时, 基准之后的 35.0 英寸处。

实用类

- (1) 后限: 在不大于 2200 磅时, 基准之后的 40.5 英寸处
 (2) 前限: 在 2200 磅时, 基准之后的 37.5 英寸处; 在 1950 磅时, 基准之后的 35.0 英寸处; 在上述两者重量之间的重量时, 线性变化; 不大于 1950 磅时, 基准之后的 35.0 英寸处。

CG Range:

Normal Category

- (1) Aft Limits: 47.3 inches aft of datum at 2450 pound or less.
 (2) Forward Limits: Linear variation from 40.0 inches aft of datum at 2450pounds to 35.0 inches aft of datum at 1950 pounds; 35.0 inches aft of datum at 1950 pounds or less.

Utility Category

- (1) Aft Limits: 40.5 inches aft of datum at 2100 pound or less.
 (2) Forward Limits: Linear variation from 36.5 inches aft of datum at 2100pounds to 35.0 inches aft of datum at 1950 pounds; 35.0 inches aft of datum at 1950 pounds or less.

When modified by Cessna Modification Kit MK172-72-01(See Note 3)

Normal Category

- (1) Aft Limits: 47.3 inches aft of datum at 2550 pound or less.
 (2) Forward Limits: Linear variation from 41.0 inches aft of datum at 2550pounds to 35.0 inches aft of datum at 1950 pounds; 35.0 inches aft of datum at 1950 pounds or less.

Utility Category

- (1) Aft Limits: 40.5 inches aft of datum at 2200 pound or less.
 (2) Forward Limits: Linear variation from 37.5 inches aft of datum at 2200pounds to 35.0 inches aft of datum at 1950 pounds; 35.0 inches aft of datum at 1950 pounds or less.

空机重心:

无

Empty Wt. C.G. Range:

None

基准:

防火墙前表面的较低部位

Reference Datum: Lower portion of front face of Firewall

平均气动弦:

58.8 英寸; 平均气动弦前缘为基准之后的 25.9 英寸处

MAC:

58.8 inches; Leading edge of MAC 25.9 inches aft of datum

调水平措施

尾锥左侧、基准之后的 108.0 英寸和 142.0 英寸处

Leveling Means: Left side of Tailcone at 108.0 inches and 142.0 inches aft of datum

最大重量:

正常类

最大停机坪重量	2457 磅
最大起飞重量	2450 磅
最大着陆重量	2450 磅

特技类	
最大停机坪重量	2207 磅
最大起飞重量	2200 磅
最大着陆重量	2200 磅

但当经赛斯纳 MK172-72-01 号改装后(见注释 3),

正常类	
最大停机坪重量	2558 磅
最大起飞重量	2550 磅
最大着陆重量	2550 磅

特技类	
最大停机坪重量	2208 磅
最大起飞重量	2200 磅
最大着陆重量	2200 磅

Maximum Weights:

Normal Category	
Maximum Ramp:	2457 pounds
Maximum Takeoff:	2450 pounds
Maximum Landing:	2450 pounds

Utility Category	
Maximum Ramp:	2207 pounds
Maximum Takeoff:	2200 pounds
Maximum Landing:	2200 pounds

When modified by Cessna Modification Kit MK172-72-01(See Note 3)

Normal Category	
Maximum Ramp:	2558 pounds
Maximum Takeoff:	2550 pounds
Maximum Landing:	2550 pounds

Utility Category	
Maximum Ramp:	2208 pounds
Maximum Takeoff:	2200 pounds
Maximum Landing:	2200 pounds

座位数量	4 (2 个座位在基准之后的 34.0 英寸至 46.0 英寸之间; 2 个座椅在基准之后的 73.0 英寸)
No. of Seats:	4 (2 at 34.0 to 46.0 inches aft of datum; 2 at 73.0 inches aft of datum)

最大行李重量:	120 磅, 在基准之后的 95.0 英寸处
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但当经赛斯纳 MK172-72-01 号改装后(见注释 3),

120 磅在基准之后的 82.0 至 108.0 英寸之间

50 磅在基准之后的 108.0 至 142.0 英寸之间

(行李舱位置的最大累积承重量为 120 磅)

Maximum Baggage:	120 pounds at 95.0 inchs aft of datum
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When modified by Cessna Modification Kit MK172-72-01(See Note 3)

120 pounds at 82.0 to 108.0 inchs aft of datum

50 pounds at 108.0 to 142.0 inches aft of datum

(Max. combined weight capacity for baggage area is 120 pounds)

燃油容量	总计 56 加仑; 可用燃油 53 加仑 (两个 28 加仑的机翼油箱, 位于基准之后的 48 英寸处) 关于不可用燃油, 见注释 1
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Fuel Capacity (Gal.):	56 gal. total; 53 gal. Usable (Two 28 gallon tanks in wings at 48 inches aft of datum)
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See NOTE 1 for data on unusable fuel.

滑油容量

2.0 加仑, 基准之前的 13.1 英寸处; 可用滑油 3.5 夸脱

但当经赛斯纳 MK172-72-01 号改装后(见注释 3),

2.0 加仑, 基准之前的 13.1 英寸处; 可用滑油 3.0 夸脱

Oil Capacity:

2.0 gallons at 13.1 inches forward of datum; 3.5 quarts usable

When modified by Cessna Modification Kit MK172-72-01(See Note 3)

2.0 gallons at 13.1 inches forward of datum; 3.0 quarts usable

操纵面位移

机翼襟翼: 起飞 $0^{\circ}-10^{\circ}$
 着陆 $0^{\circ}-30^{\circ}+0^{\circ}/-2^{\circ}$
 升降舵调整片: 向上 $22^{\circ}+1^{\circ}/-0^{\circ}$ 向下 $19^{\circ}+1^{\circ}/-0^{\circ}$
 副翼: 向上 $20^{\circ}\pm 1^{\circ}$ 向下 $15^{\circ}\pm 1^{\circ}$
 升降舵: 向上 $28^{\circ}+1^{\circ}/-0^{\circ}$ 向下 $23^{\circ}+1^{\circ}/-0^{\circ}$
 (中立位置为升降舵平衡区底面与水平安定面齐平的位置)
 方向舵(平行于水线): 向右 $16^{\circ}10'\pm 1^{\circ}$ 向左 $16^{\circ}10'\pm 1^{\circ}$
 方向舵(垂直于铰链线): 向右 $17^{\circ}44'\pm 1^{\circ}$ 向左 $17^{\circ}44'\pm 1^{\circ}$

Control Surface Movements:

Wing Flaps: Takeoff $0^{\circ}-10^{\circ}$
 Landing $0^{\circ}-30^{\circ}+0^{\circ}/-2^{\circ}$
 Elevator Tab: Up $22^{\circ}+1^{\circ}/-0^{\circ}$ Down $19^{\circ}+1^{\circ}/-0^{\circ}$
 Ailerons: Up $20^{\circ}\pm 1^{\circ}$ Down $15^{\circ}\pm 1^{\circ}$
 Elevator: Up $28^{\circ}+1^{\circ}/-0^{\circ}$ Down $23^{\circ}+1^{\circ}/-0^{\circ}$
 (Neutral position is with bottom of balance area flush with bottom of stabilizer)
 Rudder (Parallel to 0.00 W.L.): Right: $16^{\circ}10'\pm 1^{\circ}$ Left: $16^{\circ}10'\pm 1^{\circ}$
 Rudder (Perpendicular to hinge line): Right: $17^{\circ}44'\pm 1^{\circ}$ Left: $17^{\circ}44'\pm 1^{\circ}$

FAA 审定基础

1965 年 2 月 1 日生效的联邦航空条例, 修正案为 23-1 至 23-6, 但以下条款例外:

FAR 23.423; 23.611; 23.619; 23.623; 23.689; 23.775; 23.871; 23.1323 和 23.1563 为修正案 23-7; FAR 23.807 和 23.1524 为修正案 23-10; FAR 23.507; 23.771; 23.853(a),(b),(c) 和 23.1365 为修正案 23-14; FAR 23.951 为修正案 23-15; FAR 23.607; 23.675; 23.685; 23.733; 23.787; 23.1309 和 23.1322 为修正案 23-17; FAR 23.1301 为修正案 23-20; FAR 23.1353 和 23.1559 为修正案 23-21; FAR 23.603; 23.605; 23.613; 23.1329 和 23.1545 为修正案 23-23; FAR 23.441 和 23.1549 为修正案 23-28; FAR 23.779 和 23.781 为修正案 23-33; FAR 23.1; 23.51 和 23.561 为修正案 23-34; FAR 23.301; 23.331; 23.351; 23.427; 23.677; 23.701; 23.735 和 23.831 为修正案 23-42; FAR 23.961; 23.1093; 23.1143(g); 23.1147(b); 23.1303; 23.1357; 23.1361 和 23.1385 为修正案 23-43; FAR 23.562(a), 23.562(b)2, 23.562(c)1, 23.562(c)2, 23.562(c)3 和 23.562(c)4 为修正案 23-44; FAR 23.33; 23.53; 23.305; 23.321; 23.485; 23.621; 23.655 和 23.731 为修正案 23-45。

1969 年 12 月 1 日生效的 FAR 36, 修正案为 36-1 至 36-21。

FAA Certification Basis

Part 23 of the Federal Aviation Regulations effective February 1, 1965, as amended by 23-1 through 23-6, except as follows:

FAR 23.423; 23.611; 23.619; 23.623; 23.689; 23.775; 23.871; 23.1323; and 23.1563 as amended by Amendment 23-7. FAR 23.807 and 23.1524 as amended by Amendment 23-10. FAR 23.507; 23.771; 23.853(a),(b) and (c); and 23.1365 as amended by Amendment 23-14. FAR 23.951 as amended by Amendment 23-15. FAR 23.607; 23.675; 23.685; 23.733; 23.787; 23.1309 and 23.1322 as amended by Amendment 23-17. FAR 23.1301 as amended by Amendment 23-20. FAR 23.1353; and 23.1559 as amended by Amendment 23-21. FAR 23.603; 23.605; 23.613; 23.1329 and 23.1545 as amended by Amendment 23-23. FAR 23.441 and 23.1549 as amended by Amendment 23-28. FAR 23.779 and 23.781 as amended by Amendment 23-33. FAR 23.1; 23.51 and 23.561 as amended by Amendment 23-34. FAR 23.301; 23.331; 23.351; 23.427; 23.677; 23.701; 23.735; and 23.831 as amended by Amendment 23-42. FAR 23.961; 23.1093; 23.1143(g); 23.1147(b); 23.1303; 23.1357; 23.1361 and 23.1385 as amended by Amendment 23-43. FAR 23.562(a), 23.562(b)2, 23.562(c)1, 23.562(c)2, 23.562(c)3, and 23.562(c)4 as amended by Amendment 23-44. FAR 23.33; 23.53; 23.305; 23.321; 23.485; 23.621; 23.655 and 23.731 as amended by Amendment 23-45.

FAR 36 dated December 1, 1969, as amended by Amendments 36-1 through 36-21.

仅对 Garmin G1000 综合驾驶舱系统 (ICS) 的附加条款:

14 CFR 23.303、23.307、23.601、23.1163(a)、23.1367和23.1381, 修正案为原版; 14 CFR 23.1589, 修正案23-13; 14 CFR 23.771(a)修正案23-14; 14 CFR 23.607和(电气系统)、23.1309(a)(1)(2)、(c), 修正案23-17; 14 CFR 23.1301、23.1327和23.1547(e), 修正案23-20; 14 CFR 23.1501和23.1541(a)(1), (a)(2), (b)(1), (b)(2), 修正案23-21; 14 CFR 23.603和23.605, 修正案23-23; 14 CFR 23.1529, 修正案23-26; 14 CFR 23.561(e)、23.1523、23.1581(a)(2)、23.1583(a), (b),(d),(f)和23.1585(a)(b)(d), 修正案23-34; 14 CFR 23.301, 修正案23-42; 14 CFR 23.1322、23.1331和23.1357(a)(b)(c)(d)修正案23-43; 14 CFR 23.305、23.773(a)(1), (a)(2)、23.1525和23.1549修正案23-45; 14 CFR 23.1303(a)(b)(c)(f)、23.1309(a)(1)(i), (a)(1)(ii), (a)(2), (b)(1), (b)(2)(i), (b)(2)(ii), (b)(3), (b)(4)(i), (b)(4)(ii), (b)(4)(iii), (b)(4)(iv), (c)(1), (c)(2)(iii), (c)(3), (d), (e), (f)(1)、23.1311; 23.1321(a)(c)(d)(e)、23.1323(a), (b)(1), (b)(2), (c)、23.1329(g)(h)、23.1351(a)(1), (a)(2)(i), (b)(1)(iii), (b)(2)(3), (c)(4), (d)(1)、23.1353(a)(b)(c)(d)(e)、23.1359(c)、23.1361、23.1365(a)(b)(d)(e)(f)和23.1431(a)(b)(d)(e), 修正案23-49; 14 CFR 23.1325(a), (b)(1), (b)(2)(i), (b)(3), (c)(d)(e)、23.1543(b)(c)、23.1545(a), (b)(1), (b)(2), (b)(3), (b)(4)、23.1553、23.1555(a)(b)、23.1563(a)和23.1567(a), 修正案23-50; 14 CFR 23.777(a)(b)、23.955(a)(2)、23.1337(a)(1), (a)(2), (b)(1), (c)修正案23-51; 14 CFR 23.1305(a)(1), (a)(2), (a)(3), (b)(2), (b)(3)(i), (b)(4)(i), (b)(5), (b)(6)(i), 修正案23-52; 14 CFR 23.901(a)(b), 修正案23-53。

Additions for the Garmin G1000 Integrated Cockpit System (ICS) Only:

14 CFR 23.303; 23.307; 23.601; 23.1163(a); 23.1367 and 23.1381 as amended by Amendment 23- N/C. 14 CFR 23.1589 as amended by Amendment 23-13. 14 CFR23.771(a) as amended by Amendment 23-14. 14 CFR 23.607 and (Electrical System)23.1309(a)(1)(2), (c) as amended by Amendment 23-17. 14 CFR 23.1301; 23.1327 and23.1547(e) as amended by Amendment 23-20. 14 CFR 23.1501 and 23.1541(a)(1),(a)(2), (b)(1), (b)(2) as amended by Amendment 23-21. 14 CFR 23.603 and 23.605 as amended by Amendment 23-23. 14 CFR 23.1529 as amended by Amendment 23-26. 14CFR 23.561(e); 23.1523; 23.1581(a)(2); and 23.1583(a), (c), (d), (f) as amended by Amendment 23-34. 14 CFR 23.301 as amended by Amendment 23-42. 14 CFR23.1322; 23.1331 and 23.1357(a)(b)(c)(d) as amended by Amendment 23-43. 14 CFR23.305; 23.773(a)(1), (a)(2); 23.1525 and 23.1549 as amended by Amendment 23-45. 14CFR 23.1303(a)(b)(c)(f); 23.1309(a)(1)(i), (a)(1)(ii), (a)(2), (b)(1), (b)(2)(i), (b)(2)(ii),(b)(3), (b)(4)(i), (b)(4)(iii), (b)(4)(iv), (c)(1), (c)(2)(iii), (c)(3), (d), (e), (f)(1);23.1311; 23.1321 (a)(c)(d)(e); 23.1323(a), (b)(1), (b)(2), (c); 23.1329 (g)(h);23.1351(a)(1), (a)(2)(i), (b)(1)(iii), (b)(2)(3), (c)(4), (d)(1); 23.1353(a)(b)(c)(d)(e);23.1359(c); 23.1361; 23.1365(a)(b)(d)(e)(f) and 23.1431(a)(b)(d)(e) as amended by Amendment 23-49. 14 CFR 23.1325(a), (b)(1), (b)(2)(i), (b)(3), (c)(d)(e); 23.1543(b)(c);23.1545(a), (b)(1), (b)(2), (b)(3), (b)(4); 23.1553; 23.1555(a)(b); 23.1563(a) and 23.1567(a) as amended by Amendment 23-50. 14 CFR 23.777(a)(b); 23.955(a)(2);23.1337(a)(1), (a)(2), (b)(1), (c) as amended by Amendment 23-51. 14 CFR23.1305(a)(1), (a)(2), (a)(3), (b)(2), (b)(3)(i), (b)(4)(i), (b)(5), (b)(6)(i) as amended by Amendment 23-52. 14 CFR 23.901(a)(b) as amended by Amendment 23-53.

仅对 Garmin GFC-700 自动飞行控制系统 (AFCS) 的附加条款:

14CFR23.1335, 修正案 23-20; 14CFR 23.1329(a)(c)(d)(e)(f), 修正案 23-49。

Additions for the Garmin GFC-700 Automatic Flight Control System (AFCS) only:

14CFR23.1335 as amended by Amendment 23-20, 14CFR 23.1329(a)(c)(d)(e)(f) as amended by Amendment 23-49.

等效安全条款:

- | | |
|------------|---|
| (1) 进气系统防冰 | FAR §1093; 参见 1996 年 5 月 3 日的 FAA 信函 |
| (2) 油门控制 | FAR § 23.1143(g); 参见 1996 年 3 月 22 日的 FAA 信函 |
| (3) 油气混合控制 | FAR § 23.1147(b); 参见 1996 年 3 月 22 日的 FAA 信函 |
| (4) 防撞灯光系统 | 14 CFR § 23.1401(d); 参见 ACE-07-09, 2007 年 10 月 12 日的 FAA 信函 |
| (5) 航行白色要求 | 14 CFR § 23.1397(c); 参见 ACE-07-10, 2007 年 11 月 29 日的 FAA 信函 |

Equivalent Safety Items:

- | | |
|---------------------------------------|--|
| (1) Induction System Icing Protection | FAR §1093; Refer to FAA letter dated 5/3/96 |
| (2) Throttle Control | FAR § 23.1143(g); Refer to FAA letter dated 3/22/96 |
| (3) Mixture Control | FAR § 23.1147(b); Refer to FAA letter dated 3/22/96 |
| (4)Anti-Collision Light System | 14CFR §23. 1401(d); Refer to ACE-07-09, FAA letter dated 10/12/07 |
| (5)Aviation White Color Reqmt | 14CFR §23. 1397(c) ; Refer to ACE-07-10, FAA letter dated 11/29/07 |

制造人序列号 17280001 及其以后

Serial Nos. Eligible 17280001 and On

专用条件如下: :

No. 23-159-SC, “专用条件: 赛斯纳飞机公司, 赛斯纳172R飞机, 安装电子飞行仪表系统和防护此系统受高

强度辐射场 (HIRF) 的影响。

Special Conditions as follows:

No. 23-159-SC, "Special Conditions: Cessna Aircraft Company; Cessna Model 172R Airplane; Installation of Electronic Flight Instrument System and the Protection of the System From High Intensity Radiated Fields (HIRF)

CAAC 认可审定基础 CAAC 接受 FAA 的审定基础为认可审定基础
CAAC Validation Basis CAAC accept FAA's certification basis as validation basis

生产依据 生产许可证编号 PC-4
Production Basis Production Certificate No. PC-4

设备 认可审定基础规定的必要的设备必须安装在飞机上供适航审定
Equipment The basic required equipment as prescribed in the Validation Basis must be installed in the airplane for certification.

注释 1: 重量和平衡

制造人序列号: 17280001 和及其之后的

审定的空重及其相应的重心位置必须包括不可用燃油 18 磅 (基准之后的 46.0 英寸处) 和全部滑油 15.0 磅 (基准之前的 13.1 英寸处)。

NOTE 1: Weight and Balance:

Serial Nos. 17280001 and On

The certificated empty weight and corresponding center of gravity location must include unusable fuel of 18lbs. at 46.0 inches aft of datum, and full oil of 15.0 pound at 13.1 inches forward of datum.

注释 2: 飞机必须依照相应的《驾驶员操作手册和 FAA 批准的飞机飞行手册》(POH/AFM)运行。编号为 172R PHUS00 的 POH/AFM (或 FAA 后续批准的修订) 适用于 172R 型飞机; 编号为 172R180 PH00 的 POH/AFM (或 FAA 后续批准的修订) 适用于经赛斯纳改装 KIT MK172-72-01 改装的 172R 型飞机。除非特别说明, 172RPHUS00 手册的所有 POH/AFM 增补本也适用于 172R180 PH00 手册。FAA 要求的所有标牌内容在《驾驶员操作手册和 FAA 批准的飞机飞行手册》(POH/AFM)第二章。标牌内容也可以在编号为 172RMM00 《维修手册》(或其后续修订版)第 11 章“标牌和标记”中找到。

FAA 批准的飞行手册(AFM): 编号 172RPHAS-00(或后续修订版)适用于装有 Garmin G1000 驾驶舱综合系统的 172R 型飞机。该飞机必须依照相应飞行手册运行。要求的标牌内容在飞机飞行手册中。

FAA 批准的飞行手册(AFM): 编号 172RPHBUS-00(或后续修订版)适用于装有 Garmin G1000 驾驶舱综合系统和 Garmin GFC-700 自动飞行控制系统的 172R 型飞机。该飞机必须依照相应飞行手册运行。要求的标牌内容在飞机飞行手册中。

NOTE 2: The airplane must be operated according to the appropriate Pilot's Operating Handbook and FAA Approved Airplane Flight Manual (POH/AFM). POH/AFM part number 172RPHUS00 (or later approved revision) is applicable to Production Model 172R. POH/AFM part number 172R180PH00 (or later approved revision) is applicable to Production Model 172R airplanes when modified by Cessna Modification Kit MK172-72-01. All POH/AFM Supplements approved for part number 172RPHUS00, are also applicable to part number 172R180PH00, unless specifically noted otherwise in the Supplement. All FAA required placards are included in Section 2 of the applicable POH/AFM. Placards may also be found in the Maintenance Manual, part number 172RMM00 (or later revision), Chapter Eleven (11), "Placards and Markings."

FAA Approved Airplane Flight Manual (AFM): Part Number 172RPHAS-00 (or later FAA approved revisions) is applicable to the Model 172R equipped with Garmin G1000 Integrated Cockpit System. The airplane must be operated according to the appropriate AFM. Required placards are included in the AFM.

FAA Approved Airplane Flight Manual (AFM): Part Number 172RPHBUS-00 (or later FAA approved revisions) is applicable to the Model 172R equipped with Garmin G1000 Integrated Cockpit System and Garmin GFC-700 AFCS. The airplane must be operated according to the appropriate AFM. Required placards are included in the AFM.

注释 3 只有某些序列号的 172R 型飞机适宜于赛斯纳改装 KIT MK172-72-01。适用的序列号如下:

17280159 17280242 17280251 17280253 17280257
17280262 17280281 17280292 17280301 17280305
17280426 17280488 17280606 17280607 17280608

17280609 17280610 17280613 17280614 17280616
 17280621 17280622 17280623 17280624 17280631
 17280632 17280633 17280634 17280638 17280639
 17280640 17280646 17280648 17280652 17280659
 17280660 17280661 17280662 17280664 17280667
 17280668 17280669 17280670 17280672 17280673
 17280674 17280675 17280701 17280707

Only certain Model 172R airplane serial numbers are eligible for modification by Cessna Modification Kit MK172-72-01. Applicable serial numbers are as follows:

17280159 17280242 17280251 17280253 17280257
 17280262 17280281 17280292 17280301 17280305
 17280426 17280488 17280606 17280607 17280608
 17280609 17280610 17280613 17280614 17280616
 17280621 17280622 17280623 17280624 17280631
 17280632 17280633 17280634 17280638 17280639
 17280640 17280646 17280648 17280652 17280659
 17280660 17280661 17280662 17280664 17280667
 17280668 17280669 17280670 17280672 17280673
 17280674 17280675 17280701 17280707

II-172S型飞机, Skyhawk SP, 4座、封闭座舱、陆上型单翼飞机(正常类), 2座、封闭座舱、陆上型单翼飞机(实用类)

II - Model 172S, Skyhawk SP, 4PCLM (Normal Category), 2PCLM (Utility Category)

发动机 莱康明公司制造的 IO-360-L2A, 额定 180 马力
 Engine Lycoming IO-360-L2A, Rated 180 Horsepower

燃油 最低航空汽油等级 100/100LL
 Fuel 100/100LL minimum grade aviation gasoline

发动机限制 在所有使用情况下, 2700 转/分钟
 Engine Limits For all operations, 2700 RPM

螺旋桨 (a) 麦可利 型号: 1A170E/JHA7660
 (b) 桨毂: 图号 0550236

Propeller (a) McCauley Model: 1A170E/JHA7660
 (b) McCauley Spinner: Drawing No.0550236

螺旋桨限制 在全推力下的静态转速: 不超过 2400 转/分钟, 不低于 2300 转/分钟
 直径: 不超过 76 英寸, 不低于 75 英寸

Propeller Limits: Static RPM at full throttle: Not over 2400; Not Under 2300
 Diameter: not over 76 in., not under 75 in

空速限制

机动速度	105 节 (指示空速)	(102 节 (校正空速))
最大结构巡航速度	129 节 (指示空速)	(126 节 (校正空速))
不可超越速度	163 节 (指示空速)	(160 节 (校正空速))
襟翼展态速度	85 节 (指示空速)	(85 节 (校正空速))

Airspeed Limits

Maneuvering	105 Knots IAS	(102 Knots CAS)
Max. Structural Cruising	129 Knots IAS	(126 Knots CAS)
Never Exceed	163 Knots IAS	(160 Knots CAS)
Flaps Extended	85 Knots IAS	(85 Knots CAS)

重心范围 正常类

(1) 后限: 在不大于 2550 磅时, 基准之后的 47.3 英寸处

(2) 前限: 在 2550 磅时, 基准之后的 41.0 英寸处; 在 1950 磅时, 基准之后的 35.0 英寸处; 在上述两者重量之间的重量时, 线性变化; 不大于 1950 磅时, 基准之后的 35.0 英寸处。

特技类

(1) 后限: 在不大于 2200 磅时, 基准之后的 40.5 英寸处

(2) 前限: 在 2200 磅时, 基准之后的 37.5 英寸处; 在 1950 磅时, 基准之后的 35.0 英寸处; 在上述两者重量之间的重量时, 线性变化; 不大于 1950 磅时, 基准之后的 35.0 英寸处。

CG Range:

Normal Category

(1) Aft Limits: 47.3 inches aft of datum at 2550 pound or less.

(2) Forward Limits: Linear variation from 41.0 inches aft of datum at 2550pounds to 35.0 inches aft of datum at 1950 pounds; 35.0 inches aft of datum at 1950 pounds or less.

Utility Category

(1) Aft Limits: 40.5 inches aft of datum at 2200 pound or less.

(2) Forward Limits: Linear variation from 37.5 inches aft of datum at 2200pounds to 35.0 inches aft of datum at 1950 pounds; 35.0 inches aft of datum at 1950 pounds or less.

空机重心: 无

Empty Wt. C.G. Range: None

基准: 防火墙前表面的较低部位

Reference Datum: Lower portion of front face of Firewall

平均气动弦: 58.8 英寸; 平均气动弦前缘为基准之后的 25.9 英寸处

MAC: 58.8 inches; Leading edge of MAC 25.9 inches aft of datum

调水平措施 尾锥左侧、基准之后的 108.0 英寸和 142.0 英寸处

Leveling Means: Left side of Tailcone at 108.0 inches and 142.0 inches aft of datum

最大重量:

正常类

最大停机坪重量 2558 磅

最大起飞重量 2550 磅

最大着陆重量 2550 磅

特技类

最大停机坪重量 2208 磅

最大起飞重量 2200 磅

最大着陆重量 2200 磅

Maximum Weights:

Normal Category

Maximum Ramp: 2558 pounds

Maximum Takeoff: 2550 pounds

Maximum Landing: 2550 pounds

Utility Category

Maximum Ramp: 2208 pounds

Maximum Takeoff: 2200 pounds

Maximum Landing: 2200 pounds

座位数量 4 (2 个座位在基准之后的 34.0 英寸至 46.0 英寸之间; 2 个座椅在基准之后的 73.0 英寸)

No. of Seats: 4 (2 at 34.0 to 46.0 inches aft of datum; 2 at 73.0 inches aft of datum)

最大行李重量:

120 磅在基准之后的 82.0 至 108.0 英寸之间

50 磅在基准之后的 108.0 至 142.0 英寸之间

(行李舱位置的最大累积承重量为 120 磅)

Maximum Baggage:

120 pounds at 82.0 to 108.0 inches aft of datum

50 pounds at 108.0 to 142.0 inches aft of datum

(Max. combined weight capacity for baggage area is 120 pounds)

燃油容量

总计 56 加仑; 可用燃油 53 加仑

(两个 28 加仑的机翼油箱, 位于基准之后的 48 英寸处)

Fuel Capacity (Gal.):	关于不可用燃油, 见注释 1 56 gal. total; 53 gal. Usable (Two 28 gallon tanks in wings at 48 inches aft of datum) See NOTE 1 for data on unusable fuel.		
滑油容量	8.0 夸脱, 基准之前的 13.1 英寸处; 可用滑油 3.0 夸脱		
Oil Capacity:	8.0 quarts at 13.1 inches forward of datum; 3.0 quarts usable		
操纵面位移			
机翼襟翼:	起飞	0° -10°	
	着陆	0°-30° + 0°/-2°	
升降舵调整片:	向上	22°+1°/- 0°	向下 19°+1°/- 0°
副翼:	向上	20°± 1°	向下 15° ± 1°
升降舵:	向上	28°+1°/- 0°	向下 23°+1°/- 0°
	(中立位置为升降舵平衡区底面与水平安定面齐平的位置)		
方向舵 (平行于水线):		向右 16°10'± 1°	向左 16°10'± 1°
方向舵 (垂直于铰链线):		向右 17°44'± 1°	向左 17°44'± 1°
Control Surface Movements:			
Wing Flaps:	Takeoff	0° -10°	
	Landing	0° -30° + 0°/-2°	
Elevator Tab:	Up	22°+1°/- 0°	Down 19°+1°/- 0°
Ailerons:	Up	20°± 1°	Down 15° ± 1°
Elevator:	Up	28°+1°/- 0°	Down 23°+1°/- 0°
	(Neutral position is with bottom of balance area flush with bottom of stabilizer)		
Rudder (Parallel to 0.00 W.L.):		Right: 16°10'± 1°	Left: 16°10'± 1°
Rudder (Perpendicular to hinge line):		Right: 17°44'± 1°	Left: 17°44'± 1°

FAA 审定基础

1965 年 2 月 1 日生效的联邦航空条例, 修正案为 23-1 至 23-6, 但以下条款例外:

FAR 23.423; 23.611; 23.619; 23.623; 23.689; 23.775; 23.871; 23.1323 和 23.1563 为修正案 23-7; FAR 23.807 和 23.1524 为修正案 23-10; FAR 23.507; 23.771; 23.853(a),(b),(c) 和 23.1365 为修正案 23-14; FAR 23.951 为修正案 23-15; FAR 23.607; 23.675; 23.685; 23.733; 23.787; 23.1309 和 23.1322 为修正案 23-17; FAR 23.1301 为修正案 23-20; FAR 23.1353 和 23.1559 为修正案 23-21; FAR 23.603; 23.605; 23.613; 23.1329 和 23.1545 为修正案 23-23; FAR 23.441 和 23.1549 为修正案 23-28; FAR 23.779 和 23.781 为修正案 23-33; FAR 23.1; 23.51 和 23.561 为修正案 23-34; FAR 23.301; 23.331; 23.351; 23.427; 23.677; 23.701; 23.735 和 23.831 为修正案 23-42; FAR 23.961; 23.1093; 23.1143(g); 23.1147(b); 23.1303; 23.1357; 23.1361 和 23.1385 为修正案 23-43; FAR 23.562(a), 23.562(b)2, 23.562(c)1, 23.562(c)2, 23.562(c)3 和 23.562(c)4 为修正案 23-44; FAR 23.33; 23.53; 23.305; 23.321; 23.485; 23.621; 23.655 和 23.731 为修正案 23-45。

1969 年 12 月 1 日生效的 FAR 36, 修正案为 36-1 至 36-21。

FAA Certification Basis

Part 23 of the Federal Aviation Regulations effective February 1, 1965, as amended by 23-1 through 23-6, except as follows:

FAR 23.423; 23.611; 23.619; 23.623; 23.689; 23.775; 23.871; 23.1323; and 23.1563 as amended by Amendment 23-7. FAR 23.807 and 23.1524 as amended by Amendment 23-10. FAR 23.507; 23.771; 23.853(a),(b) and (c); and 23.1365 as amended by Amendment 23-14. FAR 23.951 as amended by Amendment 23-15. FAR 23.607; 23.675; 23.685; 23.733; 23.787; 23.1309 and 23.1322 as amended by Amendment 23-17. FAR 23.1301 as amended by Amendment 23-20. FAR 23.1353; and 23.1559 as amended by Amendment 23-21. FAR 23.603; 23.605; 23.613; 23.1329 and 23.1545 as amended by Amendment 23-23. FAR 23.441 and 23.1549 as amended by Amendment 23-28. FAR 23.779 and 23.781 as amended by Amendment 23-33. FAR 23.1; 23.51 and 23.561 as amended by Amendment 23-34. FAR 23.301; 23.331; 23.351; 23.427; 23.677; 23.701; 23.735; and 23.831 as amended by Amendment 23-42. FAR 23.961; 23.1093; 23.1143(g); 23.1147(b); 23.1303; 23.1357; 23.1361 and 23.1385 as amended by Amendment 23-43. FAR 23.562(a), 23.562(b)2, 23.562(c)1, 23.562(c)2, 23.562(c)3, and 23.562(c)4 as amended by Amendment 23-44. FAR 23.33; 23.53; 23.305; 23.321; 23.485; 23.621; 23.655 and 23.731 as amended by Amendment 23-45.

FAR 36 dated December 1, 1969, as amended by Amendments 36-1 through 36-21.

仅对 Garmin G1000 综合驾驶舱系统 (ICS) 的附加条款:

14 CFR 23.303、23.307、23.601、23.1163(a)、23.1367 和 23.1381, 修正案为原版; 14 CFR 23.1589, 修正案 23-13; 14 CFR 23.771(a) 修正案 23-14; 14 CFR 23.607 和 (电气系统)、23.1309(a)(1)(2)、(c), 修正案 23-17; 14 CFR 23.1301、23.1327 和 23.1547(e), 修正案 23-20; 14 CFR 23.1501 和 23.1541(a)(1)、(a)(2)、(b)(1)、(b)(2), 修

修正案23-21; 14 CFR 23.603和23.605, 修正案23-23; 14 CFR 23.1529, 修正案23-26; 14 CFR 23.561(e), 23.1523, 23.1581(a)(2), 23.1583(a), (b),(d),(f)和23.1585(a)(b)(d), 修正案23-34; 14 CFR 23.301, 修正案23-42; 14 CFR 23.1322, 23.1331和23.1357(a)(b)(c)(d)修正案23-43; 14 CFR 23.305, 23.773(a)(1), (a)(2), 23.1525和23.1549修正案23-45; 14 CFR 23.1303(a)(b)(c)(f), 23.1309(a)(1)(i), (a)(1)(ii), (a)(2), (b)(1), (b)(2)(i), (b)(2)(ii), (b)(3), (b)(4)(i), (b)(4)(ii), (b)(4)(iii), (b)(4)(iv), (c)(1), (c)(2)(iii), (c)(3), (d), (e), (f)(1), 23.1311; 23.1321(a)(c)(d)(e), 23.1323(a), (b)(1), (b)(2), (c), 23.1329(g)(h), 23.1351(a)(1), (a)(2)(i), (b)(1)(iii), (b)(2)(3), (c)(4), (d)(1), 23.1353(a)(b)(c)(d)(e), 23.1359(c), 23.1361, 23.1365(a)(b)(d)(e)(f)和23.1431(a)(b)(d)(e), 修正案23-49; 14 CFR 23.1325(a), (b)(1), (b)(2)(i), (b)(3), (c)(d)(e), 23.1543(b)(c), 23.1545(a), (b)(1), (b)(2), (b)(3), (b)(4), 23.1553, 23.1555(a)(b), 23.1563(a)和23.1567(a), 修正案23-50; 14 CFR 23.777(a)(b), 23.955(a)(2), 23.1337(a)(1), (a)(2), (b)(1), (c)修正案23-51; 14 CFR 23.1305(a)(1), (a)(2), (a)(3), (b)(2), (b)(3)(i), (b)(4)(i), (b)(5), (b)(6)(i), 修正案23-52; 14 CFR 23.901(a)(b), 修正案23-53。

Additions for the Garmin G1000 Integrated Cockpit System (ICS) Only:

14 CFR 23.303; 23.307; 23.601; 23.1163(a); 23.1367 and 23.1381 as amended by Amendment 23- N/C. 14 CFR 23.1589 as amended by Amendment 23-13. 14 CFR23.771(a) as amended by Amendment 23-14. 14 CFR 23.607 and (Electrical System)23.1309(a)(1)(2), (c) as amended by Amendment 23-17. 14 CFR 23.1301; 23.1327 and23.1547(e) as amended by Amendment 23-20. 14 CFR 23.1501 and 23.1541(a)(1),(a)(2), (b)(1), (b)(2) as amended by Amendment 23-21. 14 CFR 23.603 and 23.605 as amended by Amendment 23-23. 14 CFR 23.1529 as amended by Amendment 23-26. 14CFR 23.561(e); 23.1523; 23.1581(a)(2); and 23.1583(a), (c), (d), (f) as amended by Amendment 23-34. 14 CFR 23.301 as amended by Amendment 23-42. 14 CFR23.1322; 23.1331 and 23.1357(a)(b)(c)(d) as amended by Amendment 23-43. 14 CFR23.305; 23.773(a)(1), (a)(2); 23.1525 and 23.1549 as amended by Amendment 23-45. 14CFR 23.1303(a)(b)(c)(f); 23.1309(a)(1)(i), (a)(1)(ii), (a)(2), (b)(1), (b)(2)(i), (b)(2)(ii),(b)(3), (b)(4)(i), (b)(4)(ii), (b)(4)(iii), (b)(4)(iv), (c)(1), (c)(2)(iii), (c)(3), (d), (e), (f)(1);23.1311; 23.1321 (a)(c)(d)(e); 23.1323(a), (b)(1), (b)(2), (c); 23.1329 (g)(h);23.1351(a)(1), (a)(2)(i), (b)(1)(iii), (b)(2)(3), (c)(4), (d)(1); 23.1353(a)(b)(c)(d)(e);23.1359(c); 23.1361; 23.1365(a)(b)(d)(e)(f) and 23.1431(a)(b)(d)(e) as amended by Amendment 23-49. 14 CFR 23.1325(a), (b)(1), (b)(2)(i), (b)(3), (c)(d)(e); 23.1543(b)(c);23.1545(a), (b)(1), (b)(2), (b)(3), (b)(4); 23.1553; 23.1555(a)(b); 23.1563(a) and 23.1567(a) as amended by Amendment 23-50. 14 CFR 23.777(a)(b); 23.955(a)(2);23.1337(a)(1), (a)(2), (b)(1), (c) as amended by Amendment 23-51. 14 CFR23.1305(a)(1), (a)(2), (a)(3), (b)(2), (b)(3)(i), (b)(4)(i), (b)(5), (b)(6)(i) as amended by Amendment 23-52. 14 CFR 23.901(a)(b) as amended by Amendment 23-53.

仅对 Garmin GFC-700 自动飞行控制系统 (AFCS) 的附加条款:

14CFR23.1335, 修正案 23-20; 14CFR 23.1329(a)(c)(d)(e)(f), 修正案 23-49。

Additions for the Garmin GFC-700 Automatic Flight Control System (AFCS) only:

14CFR23.1335 as amended by Amendment 23-20, 14CFR 23.1329(a)(c)(d)(e)(f) as amended by Amendment 23-49.

等效安全条款:

- | | |
|------------|---|
| (1) 进气系统防冰 | FAR §1093; 参见 1998 年 5 月 1 日的 FAA 信函 |
| (2) 油门控制 | FAR § 23.1143(g); 参见 1998 年 5 月 1 日的 FAA 信函 |
| (3) 油气混合控制 | FAR § 23.1147(b); 参见 1998 年 5 月 1 日的 FAA 信函 |
| (4) 防撞灯光系统 | 14 CFR § 23.1401(d); 参见 ACE-07-09, 2007 年 10 月 12 日的 FAA 信函 |
| (5) 航行白色要求 | 14 CFR § 23.1397(c); 参见 ACE-07-10, 2007 年 11 月 29 日的 FAA 信函 |

Equivalent Safety Items:

- | | |
|---------------------------------------|--|
| (1) Induction System Icing Protection | FAR §1093; Refer to FAA letter dated 5/1/98 |
| (2) Throttle Control | FAR § 23.1143(g); Refer to FAA letter dated 5/1/98 |
| (3) Mixture Control | FAR § 23.1147(b); Refer to FAA letter dated 5/1/98 |
| (4) Anti-Collision Light System | 14CFR §23. 1401(d); Refer to ACE-07-09, FAA letter dated 10/12/07 |
| (5) Aviation White Color Reqmt | 14CFR §23. 1397(c) ; Refer to ACE-07-10, FAA letter dated 11/29/07 |

制造人序列号 172S8001 及其以后
Serial Nos. Eligible 172S8001 and On

专用条件如下:

No. 23-159-SC, “专用条件: 赛斯纳飞机公司, 赛斯纳172S飞机, 安装电子飞行仪表系统和防护此系统受高强度辐射场 (HIRF) 的影响。

Special Conditions as follows:

No. 23-159-SC, "Special Conditions: Cessna Aircraft Company; Cessna Model 172S Airplane; Installation of Electronic Flight Instrument System and the Protection of the System From High Intensity Radiated Fields (HIRF)

CAAC 认可审定基础 CAAC 接受 FAA 的审定基础为认可审定基础
CAAC Validation Basis CAAC accept FAA's certification basis as validation basis

生产依据 生产许可证编号 PC-4
Production Basis Production Certificate No. PC-4

设备 认可审定基础规定的必要的设备必须安装在飞机上供适航审定
Equipment The basic required equipment as prescribed in the Validation Basis must be installed in the airplane for certification.

注释 1: 重量和平衡

制造人序列号: 172S8001 和及其之后的
审定的空重及其相应的重心位置必须包括不可用燃油 18 磅 (基准之后的 46.0 英寸处) 和全部滑油 15.0 磅 (基准之前的 13.1 英寸处)。

NOTE 1: Weight and Balance:

Serial Nos. 172S8001 and On: (Model 172S)

The certificated empty weight and corresponding center of gravity location must include unusable fuel of 18lbs. at 46.0 inches aft of datum, and full oil of 15.0 pound at 13.1 inches forward of datum.

注释 2: 《驾驶员操作手册和 FAA 批准的飞机飞行手册》(POH/AFM): 编号 172S PHUS00 (或 FAA 后续批准的修订) 适用于 172S 型飞机。172S 型飞机必须依照相应的《驾驶员操作手册和 FAA 批准的飞机飞行手册》(POH/AFM)运行。FAA 要求的所有标牌内容在《驾驶员操作手册和 FAA 批准的飞机飞行手册》(POH/AFM)第二章。对于 172S 型飞机, 标牌内容也可以在编号为 172RMM02 《维修手册》(或其后续修订版)第 11 章“标牌和标记”中找到。

FAA 批准的飞行手册(AFM): 编号 172SPHAUS-00(或后续修订版)适用于装有 Garmin G1000 驾驶舱综合系统的 172S 型飞机。该飞机必须依照相应飞行手册运行。要求的标牌内容在飞机飞行手册中。

FAA 批准的飞行手册(AFM): 编号 172SPHBUS-00(或后续修订版)适用于装有 Garmin G1000 驾驶舱综合系统和 Garmin GFC-700 自动飞行控制系统的 172S 型飞机。该飞机必须依照相应飞行手册运行。要求的标牌内容在飞机飞行手册中。

NOTE 2: Pilot's Operating Handbook and FAA Approved Airplane Flight Manual (AFM) (POH/AFM): Part number 172SPHUS00 (or later FAA approved revision)is applicable to the Model 172S.The Airplane must be operated according to the appropriate POH/AFM. All FAA required placards are included in Section 2 of the POH/AFM. Placards may also be found in the Maintenance Manual, part number 172RMM02 (or later revision) for the Model 172S, Chapter11, Placards and Markings.

FAA Approved Airplane Flight Manual (AFM): Part Number 172SPHAUS-00 (or later FAA approved revisions) is applicable to the Model 172S equipped with Garmin G1000 Integrated Cockpit System. The airplane must be operated according to the appropriate AFM. Required placards are included in the AFM.

FAA Approved Airplane Flight Manual (AFM): Part Number 172SPHBUS-00 (or later FAA approved revisions) is applicable to the Model 172S equipped with Garmin G1000 Integrated Cockpit System and Garmin GFC-700 AFCS. The airplane must be operated according to the appropriate AFM. Required placards are included in the AFM.

.....结束.....

.....END.....