

中国民用航空总局
CIVIL AVIATION ADMINISTRATION OF CHINA

编号/No.: VTC124A
修订/Rev.: 0
型号/Type: 269C 269C-1
日期/Date: Apr. 1, 2002

型号认可证数据单编号: VTC124A
Validation Type Certificate Data Sheet No. VTC124A

本数据单, 作为型号认可证编号: VTC124A 的一部分, 规定了一些条件和限制, 在此条件和限制下, 获得本型号认可证的产品满足下述“认可审定基础”段落给出的适航要求。

This data sheet which is a part of Validation Type Certificate No. VTC124A prescribes conditions and limitations under which the product for which the Validation Type Certificate was issued meets the airworthiness requirements indicated in “Validation Basis” paragraph.

美国 FAA 型号合格证 No. 4H12、数据单第 25 版, 批准日期: 2002 年 1 月 22 日, 其中

269C (正常类): 1970 年 5 月 15 日 FAA 批准

269C (限制类): 1973 年 3 月 14 日 FAA 批准

269C-1 (正常类): 1995 年 7 月 31 日 FAA 批准

Type Certificate No. 4H12, Data Sheet Revision 25, Approved by FAA on January 22, 2002

Model 269C (Normal Category): Approved by FAA on May 15, 1970

Model 269C (Restricted Category): Approved by FAA on March 14, 1973

Model 269C-1 (Normal Category): Approved by FAA on July 31, 1995

型号合格证持有人 : 施维泽尔飞机公司
邮政信箱: 147
纽约州埃尔迈拉市 14902
Type Certificate Holder: Schweizer Aircraft Corporation
P.O. Box 147
Elmira, New York 14902

269C 直升机 (正常类) 和 269C 直升机 (限制类)

Model 269C Helicopter (Normal Category)

Model 269C Helicopter (Restricted Category)

发动机 Lycoming HIO-360-D1A

Engine Lycoming HIO-360-D1A

燃油 见旋翼航空器飞行手册规定

Fuel See Rotorcraft Flight Manual

发动机限制 Engine Limits	<u>马力</u> HP	<u>转速</u> RPM	<u>进气压力</u> (英寸汞柱) MP IN HG	<u>高度 (英尺)</u> Altitude Feet
对于 2050 磅总重: For 2050 lbs. :	190 190	3200 3200	26.0 24.7	海平面 S.L. 4200
旋翼限制转速和发动机使用转速 Rotor Limits and Engine Operating Speeds	<u>无动力 (旋翼转速表)</u> Power Off (Rotor Tach) 最大 Max. 504 rpm 最小 Min. 390 rpm		<u>有动力 (发动机转速表)</u> Power On (Engine Tach) 最大 Max. 3200 rpm 最小 Min. 3000 rpm	
空速限制 Airspeed Limits (IAS)	V _{ne} (不可超越) 海平面 109 英里/小时 (95 节) 对于 V _{ne} 随高度、随安装附件的减少以及限制类操作的 V _{ne} , 见旋翼航空器飞行手册。 V _{ne} (Never Exceed) S.L. 109 mph (95 knots) For reduction on V _{ne} with altitude with accessories installed, and for restricted category operation, see Rotorcraft Flight Manual.			
高度限制 Altitude Limits	对于 1700 磅或更小, 14600 英尺 (密度高度) 对于 1700 磅以上, 12000 英尺 (密度高度) 14600 feet density altitude for 1700 pounds or less 12000 feet density altitude for more than 1700 pounds			
重心范围 (纵向) C.G. Range (Longitudinal)	95.0 站位到 101.0 站位 对于附件安装的重心限制, 见旋翼航空器飞行手册 Sta. (95.0) to Sta. (101.0) For limits with accessories installed, see Rotorcraft Flight Manual.			
重心范围 (横向) C.G. Range (Lateral)	见旋翼航空器飞行手册装载说明 See Loading Instructions in Rotorcraft Flight Manual.			
调水平措施 Leveling Means	主旋翼桨毂顶部 Top Of Main Rotor Hub.			
最大重量 Maximum Weight	对于最大重量随附件安装的减少, 见旋翼航空器飞行手册 For reduction of maximum weight with accessories installed see Rotorcraft Flight Manual. 正常类: 2050 磅 (见注 1) Normal Category : 2050 lbs. (See NOTE 1.) 限制类: 2150 磅 (见注 1) Restricted Category : 2150 lbs. (See NOTE 1.) 起落架和支撑结构 (S/N-1796 及以后) 没有作过 2050 磅以上的结构验证。 如果在 2050 磅以上重量着陆, 必须检查起落架横梁。 The landing gear and support structure S/N -1796 and up are not structurally substantiated for operation above 2050 lbs. The cross beams must be inspected following landing above 2050 lbs.			
座位数 No. Seats	3 个, (83.2 站位 2 个, 80.0 站位 一个) 3, (2 at Station 83.2; 1 at Station 80.0)			

最大货物 Maximum Cargo	见旋翼航空器飞行手册装载说明和限制 See Loading Instructions and Limitations in Rotorcraft Flight Manual.																	
燃油容量 Fuel Capacity	序列号: 1796 到 1803 SN: 1796 thru 1803. 30.0 加仑 站位 108.5 (30.0 gal Sta. 108.5) 65.2 加仑带选装燃油箱 站位 108.5 (65.2 gal with Optional Tank Sta. 108.5) 序列号: 1804 及以后 SN: 1804 and Subseq. 33.0 加仑 站位 108.5 (33.0 gal Sta. 108.5) 66.0 加 仑带选装燃油箱 站位 108.5 (66.0 gal with Optional Tank Sta. 108.5)																	
滑油容量 Oil Capacity	2 加仑 站位 91 2 gal. Sta. (91)																	
起落架油气减震器压力 Landing Gear Oleo pressure	<table><tr><td>269A3240 “标准”起落架</td><td>269A3240 “Standard” Landing Gear</td></tr><tr><td>前: 350 磅/平方英寸</td><td>350 psi front (P/N 269A3150-5,-15,-19)</td></tr><tr><td>后: 560 磅/平方英寸</td><td>560 psi rear (P/N 269A3150-7)</td></tr><tr><td>后: 725 磅/平方英寸</td><td>725 psi rear (P/N 269A3150-9, -17, -21)</td></tr><tr><td>269A3260 增高起落架</td><td>269A3260 “Extended Height” Landing Gear</td></tr><tr><td>前: 350 磅/平方英寸</td><td>350 psi front (P/N 269A3150-23)</td></tr><tr><td>后: 560 磅/平方英寸</td><td>560 psi rear (P/N 269A3150-27)</td></tr></table>				269A3240 “标准”起落架	269A3240 “Standard” Landing Gear	前: 350 磅/平方英寸	350 psi front (P/N 269A3150-5,-15,-19)	后: 560 磅/平方英寸	560 psi rear (P/N 269A3150-7)	后: 725 磅/平方英寸	725 psi rear (P/N 269A3150-9, -17, -21)	269A3260 增高起落架	269A3260 “Extended Height” Landing Gear	前: 350 磅/平方英寸	350 psi front (P/N 269A3150-23)	后: 560 磅/平方英寸	560 psi rear (P/N 269A3150-27)
269A3240 “标准”起落架	269A3240 “Standard” Landing Gear																	
前: 350 磅/平方英寸	350 psi front (P/N 269A3150-5,-15,-19)																	
后: 560 磅/平方英寸	560 psi rear (P/N 269A3150-7)																	
后: 725 磅/平方英寸	725 psi rear (P/N 269A3150-9, -17, -21)																	
269A3260 增高起落架	269A3260 “Extended Height” Landing Gear																	
前: 350 磅/平方英寸	350 psi front (P/N 269A3150-23)																	
后: 560 磅/平方英寸	560 psi rear (P/N 269A3150-27)																	
有效序列号: Serial Nos. Eligible	—1796 及以后 (序列号编码含义见注 5) --1796 and up (See NOTE 5 for Serial umber coding).																	
基准 Datum	主旋翼中心线前 100 英寸 100 inches forward of main rotor centerline																	
其它使用限制 Other Operating Limitations	见旋翼航空器飞行手册 要求的标牌见注 2, 维修信息见注 4。 See Rotorcraft Flight Manual. See NOTE 2 for required placards. See NOTE 4 for maintenance.																	
主旋翼桨叶运动 Main Rotor Blade Movements	(相对于装配位置) 总距行程 $12^{\circ} \pm 1^{\circ}$ (按照 HMI 建立低桨距止动器以获得适当的自转 RPM) (Relative to Rigging Position) Collective travel $12^{\circ} \pm 1^{\circ}$ (low pitch stop to be established in accordance with HMI to obtain proper auto rotation RPM). <table><tr><td>周期变距 Cyclic,</td><td>向前 Forward</td><td>8.5° to 9.75°</td></tr><tr><td></td><td>向后 Aft</td><td>6.5° to 7.5°</td></tr><tr><td></td><td>向左 Left</td><td>6.5° to 7.5°</td></tr></table>				周期变距 Cyclic,	向前 Forward	8.5° to 9.75°		向后 Aft	6.5° to 7.5°		向左 Left	6.5° to 7.5°					
周期变距 Cyclic,	向前 Forward	8.5° to 9.75°																
	向后 Aft	6.5° to 7.5°																
	向左 Left	6.5° to 7.5°																

	向右 Right	4.5° to 6.5°
主旋翼桨叶 阻尼器设定 Main Rotor Blade Damper Setting	摩擦阻尼器 (269A1222, 269A1927, 269A1927-3) 扭矩值: 最小 200 英寸-磅, 最大 230 英寸-磅 Friction Dampers (269A1222, 269A1927, 269A1927-3). Torque to move the damper shaft through the low load stage: 200 in-lb. minimum, 230 in-lb. maximum.	
尾桨总距 Tail Rotor Blade Collective Pitch	269C 机 269A6034-7 尾桨组件: +26°±1° (推力向右); -12°±1° (推力向左) Model 269C with 269A6034-7 Tail Rotor Assembly +26°±1° (thrust to right); -12°±1° (thrust to left)	

FAA/DOA 批准的直升机飞行手册
FAA/DOA APPROVED HELICOPTER FLIGHT MANUALS

FAA2001 年 10 月 24 日修订的旋翼航空器飞行手册 (No.CSP-C-1) 及 FAA 以后批准的新的版次。
The Rotorcraft Flight Manual (No.CSP-C-1) approved by FAA revised on 24 OCT 2001 and new revisions approved by FAA afterwards.

使用寿命限制 Service Life Limits	寿命限制件清单见注 3。 See NOTE 3 for list of life limited components.
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认可基础 Validation Basis	认可基础由两部分组成: The validation basis is made up of two parts:
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- (1) 原 FAA 审定基础:
(1) FAA certification basis:

269C 正常类: 1951 年 1 月 15 日生效的 CAR 6, 再加上修正案 6-1 到 6-7 和 6-8, 不包括 CAR 6.604(c)。另外, 还要求符合 1958 年 5 月 17 日生效的 CAR 6.401(b), 1957 年 4 月 1 日生效的 CAR 6.637 和 1968 年 2 月 25 日生效的 FAR 27.1323(修正案 27-2、替代 CAR 6.612(a))。

269C 限制类: 1965 年 2 月生效的 FAR 21.25。

Model 269C (Normal Category): CAR Part 6 dated 15 January 1951, including Amendments 6-1 through 6-7 and 6-8, except CAR 6.604(c). In addition, compliance with CAR 6.401(b) effective 17 May 1958, CAR 6.637 effective 1 April 1957 and FAR 27.1323 of Amendment 27-2 effective 25 February 1968 in lieu of CAR 6.612(a) has been required.

Model 269C (Restricted Category): FAR 21.25 dated February 1, 1965.

(2) 中国民用航空总局 (CAAC) 的附加要求: CCAR 27.33(e)、CCAR 27.399

(2) CAAC additional requirements: CCAR 27.33(e), CCAR 27.399

生产基础 Production Basis	生产许可证 编号: 101 Production Certificate No. 101.
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设计定义 Design definition	见编号为 269A0053-1 的总装配图纸 See Top Assembly Drawings No. 269A0053-1
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设备	适用的适航条例 (见认可基础) 里规定的基本要求设备必须安装在直升机上。 所有必须安装的要求设备以及选装设备都列在下面的 Hughes/Schweizer 公司报告: SA-269C-22-4 “设备清单, 269C 型直升机, 序列号: 1796 及以后。”
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Equipment The basic required equipment as prescribed in the applicable airworthiness regulations (See Validation Basis) must be installed in the helicopter for certification. All required equipment that must be installed as well as optional equipment installations are listed in the following Hughes/Schweizer reports:
SA-269C-22-4 "Equipment List. Model 269C Helicopter S/N 1796 – Subseq."

269C-1 直升机（正常类），FAA 1995 年 7 月 31 日批准

Model 269C-1 Helicopter (Normal Category) Approved July 31, 1995

发动机 Textron Lycoming HO-360-C1A, FAA 型号合格证 E-286
Engine Textron Lycoming HO-360-C1A, FAA Type Certificate E-286

燃油 见旋翼航空器飞行手册
Fuel See Rotorcraft Flight Manual

发动机限制	马力	转速	进气压力(英寸汞柱)	高度(英尺)
Engine Limits	<u>HP</u>	<u>RPM</u>	<u>MP IN HG</u>	<u>Altitude Feet</u>
最大连续功率	180	2700	全油门	海平面
Max. Continuous	180	2700	Full Throttle	S.L.

旋翼限制转速和发动机转速:

Rotor Limits and Engine Operating Speeds:

<u>无动力（旋翼转速表）</u>	<u>有动力（发动机转速表）</u>
<u>Power Off (Rotor Tach)</u>	<u>Power On (Engine Tach)</u>
最大 Max. 504 rpm	最大 Max. 2700 rpm
最小 Min. 390 rpm	最小 Min. 2534 rpm

空速限制 Vne(不可超越速度); 海平面 94 节 (108 英里/小时)
Vne (门在关上位置); 90 节 (104 英里/小时)
对于 Vne 随高度的减少, 见旋翼航空器飞行手册

Airspeed Limits Vne(Never Exceed); S.L. 94 knots (108 mph)
Vne with Doors-Off; 90 knots (104 mph)
For reduction of Vne with altitude see Rotorcraft Flight Manual

高度限制 起飞/着陆 - 8,000 英尺密度高度
巡航 - 10,000 英尺密度高度
Altitude Limits Takeoff/Landing - 8,000 ft density altitude
Enroute - 10,000 ft density altitude

重心范围 (纵向)

C.G. Range Station (95.0) to Station (101.0)
(Longitudinal)

重心范围 (横向) 见旋翼航空器飞行手册装载说明

C.G. Range (Lateral) See Loading instructions in Rotorcraft Flight Manual

调水平措施 主旋翼桨毂顶部
Leveling Means Top of Main Rotor Hub

基准 Datum	主旋翼中心线前 100 英寸 100 inches forward of Main Rotor centerline
最大重量 Maximum Weight	1750 磅 1750 lbs
座位数 座位数 No. Seats	2 个, (83.2 站位 2 个) 右侧主驾驶 3 个, (83.2 站位 2 个, 80.0 站位一个) 左侧主驾驶 2. (2 at station 83.2) Right Hand PIC 3. (2 at station 83.2, 1 at station 80.0) Left Hand PIC
最大货物 Maximum Cargo	见旋翼航空器飞行手册装载说明和限制 See Loading Instructions and Limitations in Rotorcraft Flight Manual.
燃油容量 Fuel Capacity	序列号: 0001 到 0105 S/N 0001 thru 0105 标准燃油箱 Standard 总容量 35.2 美加仑, 站位 108.5 Total Capacity 35.2 U.S. gallons, station 108.5 可用容量 35.0 美加仑, 站位 108.5 Usable Capacity 35.0 U.S. gallons, station 108.5 标准+辅助燃油箱 (选装) Standard + Auxiliary (optional) 总容量 65.2 美加仑, 站位 108.5 Total Capacity 65.2 U.S. gallons, station 108.5 可用容量 63.0 美加仑, 站位 108.5 Usable Capacity 63.0 U.S. gallons, station 108.5 序列号 0106 及以后 S/N 0106 and Subsequent 标准燃油箱 Standard 总容量 33.0 美加仑, 站位 108.5 Total Capacity 33.0 U.S. gallons, station 108.5 可用容量 32.5 美加仑, 站位 108.5 Usable Capacity 32.5 U.S. gallons, station 108.5 标准+辅助燃油箱 (选装) Standard + Auxiliary (optional) 总容量 66.0 美加仑, 站位 108.5 Total Capacity 66.0 U.S. gallons, station 108.5 可用容量 64.0 美加仑, 站位 108.5 Usable Capacity 64.0 U.S. gallons, station 108.5
滑油容量 Oil Capacity	2.0 加仑, 站位 91 2.0 gallons, station 91

起落架油气减震器压力

Landing Gear Oleo pressure

前: 350 磅/平方英寸(P/N 269A3150-19) 350 psi front (P/N 269A3150-19)
 后: 725 磅/平方英寸(P/N 269A3150-21) 725 psi rear (P/N 269A3150-21)

有效序列号:

—0001 及以后 (除序列号 0013 外)

Serial Nos. Eligible

-- 0001 and subsequent except Serial No. 0013

主旋翼桨叶运动

(相对于装配位置)

总距行程 $12^{\circ} \pm 1^{\circ}$ (按照 HMI 建立低桨距止动器以获得适当的自转 RPM)Main Rotor Blade
Movements

(Relative to rigging position)

Collective Travel $12^{\circ} \pm 1^{\circ}$ (low pitch stop to be established in accordance with HMI to obtain proper auto rotation RPM).

周期变距	向前	7.5° to 9.4°
Cyclic,	Forward	
	向后	6.0° to 7.5°
	Aft	
	向左	6.5° to 7.5°
	Left	
	向右	4.5° to 6.5°
	Right	

尾桨总距

 $+26^{\circ} \pm 1^{\circ}$ (推力向右); $-12^{\circ} \pm 1^{\circ}$ (推力向左)

Tail Rotor Blade

 $+26^{\circ} \pm 1^{\circ}$ (thrust to right); $-12^{\circ} \pm 1^{\circ}$ (thrust to left)

使用寿命限制

见注 6。

Service Life Limits

See Note 6.

认可基础

认可基础由两部分组成:

Validation Basis

The validation basis is made up of two parts:

(1) 原 FAA 审定基础:

(1)FAA certification basis:

正常类: 1951 年 1 月 15 日生效的 CAR 6, 再加上修正案 6-1 到 6-7 和 6-8, 不包括 CAR 6.604(c)。另外, 还要求符合 1958 年 5 月 17 日生效的 CAR 6.401(b), 1957 年 4 月 1 日生效的 CAR 6.637 和 1968 年 2 月 25 日生效的 FAR 27.1323(修正案 27-2、替代 CAR 6.612(a)), 按照 FAR36 部附录 J (第 20 修正案) 的噪声标准 (Normal Category): CAR Part 6 dated 15 January 1951, including Amendments 6-1 through 6-7 and 6-8, except CAR 6.604(c). In addition, compliance with CAR 6.401(b) effective 17 May 1958, CAR 6.637 effective 1 April 1957, FAR 27.1323 of Amendment 27-2 effective 25 February 1968 in lieu of CAR 6.612(a), and noise standards per FAR Part 36 Appendix J, Amendment 20 has been shown.

(2)中国民用航空总局 (CAAC) 的附加要求: CCAR 27.33(e)、CCAR 27.399

(2)CAAC additional requirements : CCAR 27.33(e).CCAR 27.399

生产基础

生产许可证 编号: 101

Production Basis

Production Certificate No. 101

其它使用限制

见旋翼航空器飞行手册。要求的标牌见注 2, 维修信息见注 4。

Other Operating Limitations	See Rotorcraft Flight Manual. See NOTE 2 for required placards. See NOTE 4 for maintenance.
设计定义 Design definition	见编号为 269A0053-3 的总装配图纸 See Top Assembly Drawings No.269A0053-3
设备	适用的适航条例（见认可基础）里规定的基本要求设备必须安装在直升机上。 所有必须安装的要求设备以及选装设备都列在下面的 Schweizer 公司报告： SA-269C-22-5。
Equipment	The basic required equipment as prescribed in the applicable airworthiness regulations (See Validation Basis) must be installed in the helicopter for certification. All required equipment that must be installed as well as optional equipment installations are listed in the Schweizer Report SA-269C-22-5.

FAA 批准的直升机飞行手册

FAA APPROVED HELICOPTER FLIGHT MANUAL

FAA2000 年 3 月 23 日修订的旋翼航空器飞行手册（No.CSP-C1-1）及 FAA 以后批准的新的版次。

The Rotorcraft Flight Manual (No.CSP-C1-1) approved by FAA and revised on 23 MAR 2000 and new revisions approved by FAA afterwards.

注1. 每架直升机必须在原始适航审定之时和以后的任何时刻（除非运营者具有适当的重量控制系统）配有当前的重量和平衡报告，包括设备清单、包括审定空重和装载说明。必要时，必须按照旋翼航空器飞行手册装载说明携带配重。

NOTE 1. Current weight and balance report, including list of equipment including certificated empty weight and loading instructions, must be provided for each helicopter at the time of original airworthiness certification and at all times thereafter (except in the case of operators having an appropriate weight control system). Ballast, when necessary, must be carried in accordance with the loading instructions in the Rotorcraft Flight Manual.

注2. 下述标牌必须安装在驾驶员的清晰的视野内：

NOTE 2. The following placard must be installed in clear view of the pilot:

"本直升机的操作必须符合相应的旋翼航空器飞行手册中规定的使用限制。"

"This Helicopter must be operated in compliance with the operating limitations specified in the pertinent Rotorcraft Flight Manual."

对于其它标牌，见相应的旋翼航空器飞行手册。

For additional placards, see the pertinent Rotorcraft Flight Manual.

注2. (a) 269C 型机关键件更换时间列于下表。没有 FAA 工程批准这些更换时间值或使用寿命不能增加。

NOTE 3. (a) The retirement times of critical parts for Model 269C are listed in the following table. These values of retirement or service life cannot be increased without approval by FAA Engineering.(See Note 6 for Model 269C-1):

269C 型机 序列号 1796 及以后 Model 269C S/N 1796& Subs.		
描述 Description	件号 Part Number	小时 Hours
桨叶组件—主旋翼 Blade Assembly - M/R	269A1160	5500
	269A1185-1,-7	5500
	269A1185-9	3050
桨距轴承轴—主旋翼 Pitch Brg. Shaft - M/R	269A1240-7	3600
弹性阻尼器—主旋翼 见注 3(e) Dampers-Elastomeric-M/R see note 3(e)	269A1290-1, -3	6000
旋翼轴轴筒—主旋翼 Mast -M/R	269A2010-5, -15	13590
止推轴承—主旋翼 Thrust Bearing - M/R	269A5050-63, -95	3000
尾梁组件 Tail Boom Assy	269A2320-7 with 269A2324-11 center attach fitting installed	2100
	269A2320-7 with 269A2324-7 center attach fitting installed	500
	269A2320-11	2100
	269A2320-17,-21	4200
	269A2320-19	2100
尾梁斜支柱 Tail Boom Struts	269A2015-11, -13, -15, -17	10700
安定面组件—垂直 Stab. Assy - Vert.	269A2419-3	20540
安定面组件—水平 Stab. Assy - Horiz.	269A2516-9	2500
	269A2516-21	4200
主齿轮箱小齿轮组件 Main Gear Box Pinion Assy	269A5103	6000
	269A5103-9	6000
	269A5103-21	6000
	269A5103-31, -41, -51	6000
主旋翼驱动轴 Main Rotor Drive Shaft	269A5305-11, -111	1900
主旋翼驱动轴(花键连接) Main Rotor Drive Shaft (splined)	269A5326-1	3200
主旋翼桨毂(花键连接) Main Rotor Hub(splined)	269A5325-1	8000
联轴器组件—冠齿轮, 见 3f Carrier Assembly-Ring Gear, see item 3f	269A5194	6000
下皮带轮联结轴(269A5504-5 组件) Lower Pulley Coupling Shaft (269A5504-5 Assy)	269A5504-3	1500
下部皮带轮联结轴 (269A5559 组件) Lower Pulley Coupling Shaft (269A5559 Assy)	269A5559-3	6000

惰性轮轴承 Idler Pulley Bearings	269A5050-62	600
尾桨齿轮箱输入轴 Shaft - Input T/R GB	369A5406	8600
	369A5425, -3, -5	8600
	269A5626-3, -5	8600
尾桨驱动轴组件 Shaft Assy - T/R Drive	269A6040-7,-9,-9M	6000
花键适配接头 Spline Adapter Fitting	269A6035-9, -19, -23	9000
桨叶组件—尾桨 Blade Assy - T/R		
拉扭条—尾桨 Retention Straps - T/R	369A1706	3540
	269A6065	3540
	269A6065-507	5100
	369A1706-505, -507	5100

注 3. (b) 禁止在不同系列的直升机（即 369/269）上互换寿命限制部件。在不同型号、构型之间互换的或者以前在不同系列之间互换的寿命限制部件必须限制到所涉及的型号或构型的最小使用寿命。零部件仅适用于列出使用寿命的型号。具有已知的使用小时数但没有型号适用性确认的交换部件不得超过任何适用型号的最低寿命。如果使用小时数未知，不管型号适用性如何，该部件都不得交换到将该部件列为寿命限制件的型号上。

NOTE 3. (b) It is prohibited to interchange life limited components between different series of helicopters (i.e. 369/269). Life limited components interchanged between Models, configurations, or previously between series must be restricted to the lowest service life indicated for the Models or configurations affected. Parts are applicable only on Models under which a service life is listed. Interchanged components with known service hours but without Model application identification may not exceed the lowest life listed for any applicable Model. If the service hours are not known, regardless of Model application, the component cannot be interchanged to Models that list the component as limited life.

(c) 当寿命限制达到后必须将拆掉的寿命限制件破坏或作上永久性标记，以防返回使用。

(c) Life limited components removed when life limit has been reached must be destroyed or permanently marked to prevent return to service.

(d) （弹性阻尼器）强制检查必须按照 269 系列“直升机维修手册”（HMI）要求进行，在 4200 小时以前检查间隔为 600 小时，4200 小时以后检查间隔为 300 小时，直到阻尼器的总使用寿命达到 6000 小时为止。

(d) (Elastomeric Dampers) Mandatory inspection required in accordance with the 269 Series "Helicopter Maintenance Instruction" (HMI) requirements at 600-hour intervals for operation up to 4200 hours and at 300-hour intervals thereafter to a total damper operational service time of 6000 hours.

(e) 在件号上附加的字母和/或数字后缀表示特殊的制造或处理工艺，并不改变零部件的更换时间。例如，269A5305-11 和 269A5305-11M2 维修要求相同。

(e) Alpha and/or numeric suffixes added to part numbers denote special manufacturing or handling procedures and do not alter the replacement requirements of the part. For example, 269A5305-11 and 269A5305-11M2 are subject to the same requirements.

(f) 269A5193 联轴器是 269A5194 联轴器组件的一部分。

(f) 269A5193 Carrier is part of 269A5194 Carrier Assembly.

注 4. 直升机正确维护的必要信息包容在制造商的维修手册中，维修手册随每架直升机提供。
NOTE 4. Information essential to the proper maintenance of these helicopters is contained in the Manufacturer's Handbook of Maintenance Instructions, which is provided with each helicopter.

注 5. 航空器序列号编码表示制造顺序的年月。
NOTE 5. Aircraft serial numbers are coded to show the month and year of manufacture sequence.

例如: 640103, 1150015

EXAMPLES: 640103,
1150015

6

4

0103

11

5

0015

连续的序列号，每一型号都从
0001 开始

Serial number in consecutive
order from 0001 for each model

制造的年:

Year of Manufacture:

4 - 1964

5 - 1965

制造的月:

6 - 6 月/June

Month of Manufacture:

11 - 11 月/November

S/N 1065, S/N 1075 及以后的 269C 直升机交付时序列号中不再有制造日期编码。从 S/N S1166 开始，序列号前加一字母“S”。

Model 269C Helicopters, S/N 1065, S/N 1075 and subsequent will be delivered without the manufacturing date coding as part of the serial number. Serial numbers are prefixed by the letter "S" starting with S/N S1166 and up.

注 6. (a) 269C-1 型直升机关键件的更换时间列于下表。这些更换时间或使用寿命没有 FAA 工程批准不能增加。

NOTE 6. (a) The retirement times of critical parts for Model 269C-1 are listed in the following table. These values of retirement or service life cannot be increased without approval by FAA Engineering.

(b) 描述 Description	件号 Part Number	序列号 0001 和以后 S/N 0001 & Subq 小时/Hours
主旋翼桨叶 Main Rotor Blade	269A1185-1,-7	5,500
桨距轴承轴 Pitch Bearing Shaft	269A1185-9	3,050
弹性阻尼器 Elastometric Dampers	269A1240-7	4,000
主旋翼输入小齿轮 M/R Input Pinion	269A1290-3	6,000
主旋翼驱动轴(螺栓连接) M/R Drive Shaft(bolted)	269A5103-51	8,000
主旋翼驱动轴(花键连接) M/R Drive Shaft(splined)	269A5305-111	1,900
主旋翼桨毂 M/R Hub (Splined)	269A5326-1	4,000
尾桨驱动轴 T/R Drive Shaft	269A5325-1	8,000
	269A6040-7,-9,- 9M	6,000

尾桨齿轮箱输入轴 Shaft-Input T/R GB	269A5626-5	8,600
尾桨叶 T/R Blade	269A6035-23	9,000
尾桨拉-扭条 T/R T-T Straps	269A6065-507	5,100
旋翼主轴轴筒 Main Rotor Mast	269A2010-5, -15	13,590
尾梁组件 Tail Boom Assy.	269A2320-13	2,100
尾梁斜支柱 Tail Boom Strut	269A2320-15	4,200
水平安定面 Horizontal Stab.	269A2015-11, -13, -15, -17	10,700
下皮带轮联结轴 Lower Pulley Coupling Shaft	269A2516-21	4,200
止推轴承-主旋翼 Thrust Bearing-M/R	269A5559-3	6,000
联轴器组件-冠齿轮 (见 6(h)) Carrier Assy-Ring Gear see item 6(h)	269A5050-63, -95	4,200
	269A5194	8,000

(c) 禁止在不同系列的直升机 (即 369/269) 上互换寿命限制部件。在不同型号、构型之间互换的或者以前在不同系列之间互换的寿命限制部件必须限制到所涉及的型号或构型的最小使用寿命。零部件仅适用于列出使用寿命的型号。具有已知的使用小时数但没有型号适用性确认的交换部件不得超过任何适用型号的最低寿命。如果使用小时数未知, 不管型号适用性如何, 该部件都不得交换到将该部件列为寿命限制件的型号上。

(c) It is prohibited to interchange life limited components between different series of helicopters (i.e. 369/269). Life limited components interchanged between Models, configurations, or previously between series must be restricted to the lowest service life indicated for the Models or configurations affected. Parts are applicable only on Models under which a service life is listed. Interchanged components with known service hours but without Model application identification may not exceed the lowest life listed for any applicable Model. If the service hours are not known, regardless of Model application, the component cannot be interchanged to Models that list the component as limited life.

(d) 当寿命限制达到后必须将拆下来的寿命限制件破坏或作上永久标记, 以免返回使用。

(d) Life limited components removed when life limit has been reached must be destroyed or permanently marked to prevent return to service.

(e) 269A2402 垂直安定面是 269A2320-13 尾梁组件的一部分。垂直安定面和尾梁有相同的使用寿命 (2100 小时), 所以垂直安定面应该与尾梁组件一起更换。

(e) The 269A2402 Vertical Stabilizer is part of the 269A2320-13 Tail Boom Assembly. The vertical Stabilizer has the same service life (2,100 hours) as does the Tail Boom and therefore the vertical stabilizer shall be retired with the Tail Boom Assembly.

(f) 一些零部件看起来可以在 269C-1 和其它 269 系列直升机上互换。但是由于维修计划的不同, 只有注 6(b) 所规定的 “-” 后最新尾数相应的零部件适用于安装在 269C-1 上。

(f) Some Parts may appear to be interchangeable between the Model 269C-1 and other 269 series helicopters. However due to differences in maintenance schedules, only the most current dash numbers as defined in Note 6(b) are applicable for installation on the Model 269C-1.

(g) 在件号上附加的字母和/或数字后缀表示特殊的制造或处理工艺, 并不改变零部件的更换时间。例如, 269A5305-11 和 269A5305-11M2 维修要求相同。

(g) Alpha and/or numeric suffixes added to part numbers denote special manufacturing or handling procedures and do not alter the replacement requirements of the part. For example, 269A5305-11 and 269A5305-11M2 are subject to the same requirements.

(h) 269A5193 联轴器是 269A5194 联轴器组件的一部分。
(h) 269A5193 Carrier is part of 269A5194 Carrier Assembly.

注 7. 噪声验证:
NOTE 7. Noise Substantiation:

尽管不是 FAA 审定基础的一部分, 269C 直升机符合 FAR36 部附录 J 第 20 修正案的要求
Although not part of the Certification Basis, the Model 269C Helicopter is compliant with the requirements of FAR Part 36 Appendix J, Amendment 20.

— 结束 —
- END -