

型号认可证数据单
THE VALIDATION DATA SHEET

编号/No.: VTC0249E

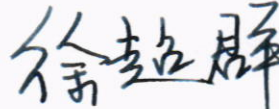
版次/Revision: 3

型号/Type:

AS907-1-1A, AS907-2-1G,

AS907-2-1A, AS907-3-1E

批准人/Approved by: Xu Chaoqun



职务/Title: Director General of CAAC-AAD

日期/Date: 2017-12-19

本数据单是型号认可证（编号 VTC0249E）的组成部分，它规定了符合中国民用航空局的适航要求所颁发此型号认可证的产品状态和限制。

This data sheet, which is a part of Validation Type Certificate (No: VTC0249E), prescribes conditions and limitations under which the product for which the Type Certificate was issued meets the airworthiness requirements of the Chinese Civil Aviation Regulation.

型号认可证持有人/Validation of Type Certificate Holder:

Honeywell International Inc.

111 South 34th Street

Phoenix, AZ, 85034

有效页清单/List of effective pages :

页数/Pages	1	2	3	4	5	6	7	8	9	10	11
版次/Revision	3	3	3	3	3	3	3	3	3	3	3

第1部分 概述

SECTION 1 GENERAL

1. CAAC认可数据单和型号合格证数据单/CAAC Validation Data Sheet and Type Certificate Data Sheet

	中国民用航空局认可数据单号、版次和颁发日期 CAAC Validation Data Sheet Number, Revision and issuance date	型号合格证数据单号、版次和颁发日期 Type Certificate Data Sheet Number, Revision and issuance date
当前 Current	VTC0249E, 版次 3, 2017 年 12 月 19 日 VTC0249E, Revision 3, December 19, 2017	FAA TCDS No. E00010LA, 第 7 版, 2016 年 3 月 22 日 FAA TCDS No. E00010LA, Revision 7, March 22, 2016
历史 History	VTC0249E, 版次 0, 2010 年 12 月 2 日 VTC0249E, Revision 0, December 2, 2010	FAA TCDS No. E00010LA, 第 2 版, 2004 年 4 月 14 日 FAA TCDS No. E00010LA, Revision 2, April 14, 2004
	VTC0249E, 版次 1, 2013 年 5 月 15 日 VTC0249E, Revision 1, May 15, 2013	FAA TCDS No. E00010LA, 第 3 版, 2011 年 3 月 18 日 FAA TCDS No. E00010LA, Revision 3, March 18, 2011
	VTC0249E, 版次 2, 2014 年 12 月 29 日 VTC0249E, Revision 2, December 29, 2014	FAA TCDS No. E00010LA, 第 6 版, 2014 年 9 月 15 日 FAA TCDS No. E00010LA, Revision 6, September 15, 2014

2. 类别/Category:

航空发动机/Aeroengine

3. 认可当局/Validation Authority:

中国民用航空局/CAAC

审定当局/Certifying Authority:

美国联邦航空管理局/FAA

4. 型号合格证持有人/Type Certificate Holder:

Honeywell International Inc.
111 South 34th Street
Phoenix, AZ, 85034
USA

5. ETOPS:

不适用/NA

6. CAAC 专用要求/Special Requirements:

满足中国 3 号燃油 (GB6537-2006) 的要求。
Compliance with requirement of P.R. China JET Fuel
No.3 (GB6537-2006)

第2部分 数据单

SECTION 2 DATA SHEET

型号/MODELS	AS907-1-1A	AS907-2-1A	AS907-2-1G	AS907-3-1E
类型/TYPE	涡扇：一级风扇，四级轴流高压压气机和一级离心高压压气机，环形燃烧室，两级高压涡轮，三级低压涡轮。 Turbofan: One stage fan, four stage axial flow high pressure compressor, one stage centrifugal high pressure compressor, annular combustor, two stage high pressure turbine, and three stage low pressure turbine.			
静推力额定值（见注 4 和注 12） Static Thrust Ratings (see NOTE 4 and NOTE 12)	海平面、外界温度 ISA+15℃、最大连续： 6929 磅 Max Continuous at Ambient Temperature, at Sea Level, lbs 6929 at ISA +15℃	海平面、外界温度 ISA+14℃、最大连续： 7530 磅 Max Continuous at Ambient Temperature, at Sea Level, lbs 7530 at ISA +14℃	海平面、外界温度 ISA +17℃、最大连续：7337 磅 Max Continuous at Ambient Temperature, at Sea Level, lbs 7337 at ISA +17℃	海平面、外界温度 ISA +22.5℃、最大连续： 7335 磅 Max Continuous at Ambient Temperature, at Sea Level, lbs 7335 at ISA +22.5℃
	海平面、外界温度 ISA+20℃、起飞：6944 磅（见注 10）Takeoff, at Ambient Temperature, at Sea Level, lbs (see NOTE 10) 6944 at ISA +20℃	海平面、外界温度 ISA+20℃、起飞：7530 磅（见注 10）Takeoff, at Ambient Temperature, at Sea Level, lbs (see NOTE 10) 7530 at ISA +20℃	海平面、外界温度 ISA+17℃、起飞：7765 磅（见注 10）Takeoff, at Ambient Temperature, at Sea Level, lbs (see NOTE 10) 7765 at ISA +17℃	海平面、外界温度 ISA+18℃、起飞：7638 磅（见注 10）Takeoff, at Ambient Temperature, at Sea Level, lbs (see NOTE 10) 7638 at ISA +18℃
控制/Controls	燃油控制和功率管理由全权限数字电子控制系统完成，该系统带有两个电子控制单元，双通道控制。系统的硬件、软件以及带有整体燃油泵的液压机械组件由批准的发动机设备清单控制（见注 13） Fuel control and power management are controlled by a Full Authority Digital Electronic Control (FADEC) system which features a dual-channel electronic control in the form of two electronic control units (ECUs). The hardware and software configurations of this system and the associated engine hydromechanical unit with integral fuel pump are controlled by an approved engine equipment list. (See NOTE 13).			
主要尺寸/Principal Dimensions	尺寸和重心位置参见发动机安装图纸 Refer to the engine installation drawing for dimensions and center of gravity location.			
干重(最大), 磅 Weight, Dry, Pounds (maximum)	1534	1534	1534	1514
	发动机重量包含批准的发动机设备清单列出的所有基本零件。安装在发动机上、经 14 CFR 25 部审定批准的零件，并不包含在基本重量中。 The engine weight includes all components of the basic engine as defined by the approved Engine Equipment List. Components that are certified as part of the aircraft under Title 14 CFR part 25, which are mounted on the engine, are not included in the basic weight.			

型号/MODELS	AS907-1-1A	AS907-2-1A	AS907-2-1G	AS907-3-1E
燃油/Fuel	符合 Honeywell 国际公司 EMS53111 (A 型燃油)、EMS53112(A-1 和 JP-8 型燃油)、EMS53113 (Jet B 和 JP-4 型燃油)和 EMS53116 (JP-5 型燃油)规范。 Fuels conforming to Honeywell International Inc. Specifications EMS53111 (Jet A Type), EMS53112 (Jet A-1 and JP-8 Types), EMS53113 (Jet B and JP-4 Types), and EMS53116 (JP-5 Types). 将主燃油转换到替换燃油时，不需要燃油控制调节。 No fuel control adjustment is required when changing from primary to alternate fuels. 批准的燃油牌号和燃油添加剂参见发动机安装手册（见注 12） Refer to engine installation manual for approved fuel types and fuel additives (see NOTE 12)			
滑油/Oil	符合霍尼韦尔国际公司 EMS53110 规范，II 类。批准的发动机滑油参见发动机安装手册（见注 12）。 Oil conforming to Honeywell International Inc. Specification EMS53110, Type II. Refer to engine installation manual for approved engine oils (see NOTE 12)			

审定基础/Certification Basis:

AS907-1-1A 和 AS907-2-1G	1965 年 2 月 1 日生效的 14 CFR 第 33 部，包含 2000 年 12 月 13 日生效的第 1 到 20 修正案。14 CFR Part 33, effective February 1, 1965, as amended by 33-1 through 33-20, effective December 13, 2000. 1990 年 9 月 10 日生效的 14 CFR 第 34 部，包含 2013 年 10 月 23 日生效的第 1 到 5A 修正案。燃油排泄和排气排出物规定；见注 15。
AS907-2-1A	1965 年 2 月 1 日生效的 14 CFR 第 33 部，包含 2012 年 9 月 20 日生效的第 1 到 33 修正案，不包含第 22 和 31 修正案。/ 14 CFR Part 33, effective February 1, 1965, as amended by 33-1 through 33-33, effective September 20, 2012, excluding Amendments 33-22 and 33-31. 1990 年 9 月 10 日生效的 14 CFR 第 34 部，包含 2013 年 10 月 23 日生效的第 1 到 5A 修正案。燃油排泄和排气排出物规定；见注 15。
AS907-3-1E	1965 年 2 月 1 日生效的 14 CFR 第 33 部，包含 2012 年 9 月 20 日生效的第 1 到 33 修正案，不包含第 22 修正案。 1990 年 9 月 10 日生效的 14 CFR 第 34 部，包含 2013 年 10 月 23 日生效的第 1 到 5A 修正案。燃油排泄和排气排出物规定；见注 15。

发动机型号列表/Engine Table of Models

型号/Model	申请日期 Date of Application	型号合格证颁发日期 Type Certificate Issued	首次生产日期 1st Production Date
AS907-1-1A	1998 年 11 月 4 日 November 4, 1998	2002 年 6 月 25 日 June 25, 2002	2003 年 5 月 29 日 May 29, 2003
AS907-2-1G	2006 年 4 月 20 日 April 20, 2006	2011 年 3 月 18 日 March 18, 2011	2011 年 3 月 25 日 March 25, 2011
AS907-2-1A	2012 年 8 月 31 日 August 31, 2012	2014 年 5 月 13 日 May 13, 2014	2014 年 5 月 14 日 May 14, 2014
AS907-3-1E	2009 年 2 月 26 日 February 26, 2009	2014 年 7 月 7 日 July 7, 2014	2014 年 7 月 12 日 July 12, 2014

生产基础/

PRODUCTION BASIS

1965 年 3 月 4 日颁发的 413 号生产许可证, 2000 年 1 月 25 日重新颁发给 Honeywell 国际公司 413NM 号生产许可证

Production Certificate No. 413, issued March 4, 1965. Reissued Production Certificate No. 413NM to Honeywell International Inc. on January 25, 2000.

注释/NOTE

注 1/NOTE 1.

发动机高压转子和低压转子的最大允许转速

Maximum permissible operating speeds for the low- and high-pressure rotors of the engine are as follows:

型号 Model	低压转子转速 Low-Pressure Rotor Speed (N_1), RPM			高压转子转速 High-Pressure Rotor Speed (N_2), RPM		
	最大连续 Maximum Continuous	起飞(见注 10) Takeoff (see NOTE 10)	最大瞬态 (最大 20 秒) Maximum Transient (20 Seconds Maximum)	最大连续 Maximum Continuous	起飞(见注 10) Takeoff (see NOTE 10)	最大瞬态 (最大 20 秒) Maximum Transient (20 Seconds Maximum)
AS907-1-1A	9723	9812	9957	27319	27568	28075
AS907-2-1A	9800	9830	9957	27530	27714	28075
AS907-2-1G	9800	9830	9957	27599	27714	28075
AS907-3-1E	9830	9830	9957	27530	27686	28075

注 2/NOTE 2.

温度限制/Temperature Limits:

最大指示涡轮级间温度 (ITT) 限制/Maximum Indicated Interturbine Temperature (ITT) Limits: °F (°C)

型号 Model	最大连续 Max. Continuous	起飞 (见注 10) Takeoff (see NOTE 10)	起动 (地面/空中) Starting (Ground/Air)	瞬态 Transient
AS907-1-1A	1702 (928)	1735 (946)	*	1764 (962) **
AS907-2-1A	1742 (950)	1751 (955)	*	1778 (970) **
AS907-2-1G	1742 (950)	1751 (955)	*	1778 (970) **
AS907-3-1E	1742 (950)	1751 (955)	*	1778 (970) ***

* 随 N2 转速变化, 参见发动机安装手册 (见注 12)

Varies with N2 speed, refer to engine installation manual (see NOTE 12)

** 最大 20 秒/20 seconds maximum

*** 最大 2 分钟/2 minutes maximum

最大滑油进口温度限制/Maximum Oil Inlet Temperature Limits:

连续/Continuous, °F (°C)	瞬态 (最大 2 分钟) /Transient (2 Minutes Max.), °F (°C)
280 (138)	310 (154)

外部发动机零件最大温度 (某些零件的温度限制) 已在发动机安装手册中指定 (见注 12)

External engine components maximum temperature (limiting temperature of specific components) is as specified in the engine installation manual (see NOTE 12).

起动期间, 在蒸汽体积与液体体积比 (V/L) 为 0.45 时, AS907-1-1A、AS907-2-1A 和 AS907-2-1G 型发动机可以在发动机燃油进口温度最高为 185°F (85°C) 条件下工作, AS907-3-1E 型发动机可以在发动机燃油进口温度最高为 131°F (55°C) 条件下工作; 起动期间, 发动机可以在燃油粘度不大于 12.0 厘斯时, 在发动机燃油进口温度最低为 -65°F (-54°C) 条件下工作; 以上工作信息均在发动机安装手册中得到批准 (见注 12)。

Operation at an engine fuel inlet temperature as high as 185°F (85°C) for the AS907-1-1A, AS907-2-1A and AS907-2-1G engines and 131°F (55°C) for the AS907-3-1E engine with a vapor volume to liquid volume ratio (V/L) equal to 0.45, and as low as -65°F (-54°C) with fuel at a viscosity of 12.0 centistokes or less during starting is approved as specified in the engine installation manual (see NOTE 12).

注 3/NOTE 3.

燃油和滑油压力限制/Fuel and Oil Pressure limits:

燃油泵进口压力/Fuel pump inlet pressure:

最小值-下列数值中的最大值/Minimum – Whichever is highest of the following:

- 大于真蒸汽压力 5 磅/平方英寸/5 psi above the true vapor pressure of the fuel;
- 对应蒸汽体积与液体体积比为 0.45 的压力/Pressure corresponding to a V/L ratio of 0.45;
- 35% 大气压力/35% of atmospheric pressure;
- 2 磅/平方英寸/psia.

最大值/Maximum:

- 35 磅/平方英寸 (表压) /psig

滑油压力/Oil pressure:

滑油压力不受调节, 压力限制随 N₂ 转速变化。参见发动机安装手册 (见注 12)

Oil pressure is not regulated and pressure limits vary with N₂. Refer to engine installation manual (NOTE 12).

注 4/NOTE 4.

额定值基于静态试验, 并在下述条件下得到/The ratings are based on static test stand operation and under the following conditions:

- a) 无航空器附件传动载荷/ No loading of aircraft accessory drives;
- b) 无航空器用压气机引气提取/ No aircraft compressor bleed air extraction;
- c) 风扇排气和涡轮排气喷管符合霍尼韦尔国际公司 N10780-1 和 N10781-1 图纸/Fan exhaust and turbine exhaust nozzles conforming to Honeywell International Inc. Drawings N10780-1 and N10781-1;
- d) 钟形进口遵照霍尼韦尔国际公司 5837800-1 图纸/Bellmouth inlet conforming to Honeywell International Inc. Drawing 5837800-1;
- e) 干进口空气/Dry inlet air;
- f) 无排气喷管背压/No exhaust nozzle back pressure;
- g) 转点处的静态推力额定值外界温度/Static Thrust Ratings ambient temperature at breakpoint.

注 5/NOTE 5.

附件传动规定/Accessory Drive Provisions:

附件传动 Accessory Drive	传动类型(每种一类) Drive Type (One Each)	内花键构造 Internal Spline Configuration	转速 (转每分钟) 和面向驱动末端的转向 /RPM and Rotation Facing Drive End	注 (b) /Note (b) 附件最大扭矩 (磅-英寸) /Accessory Max Torque (lb-in)			最大重量 (磅) Maximum Weight (pounds)	最大悬挂力矩 (磅-英寸) Maximum Overhang Moment (lb-in)**
				Tc	To	Ts		
发电机/交流发电机 D30* (AS907-1-1A, AS907-2-1A 和 AS907-2-1G) Generator/Alternator D30* (AS907-1-1A, AS907-2-1A, and AS907-2-1G)	在以下几方面 修改的 AS468B-AV1: 转速、扭矩、附件重量和力矩/ AS468B-AV1 modified as follows: RPM, torques, accessory weight and moment as shown	AS468B	13665 注 (a) /Note (a) CW	242	363	1600	34.7	128.5
发电机/交流发电机 D30* (AS907-3-1E) Generator/Alternator D30*(AS907-3-1E)	在以下几方面 修改的 AS468B-AV1: 转速、扭矩、附件重量和力矩/ AS468B-AV1 modified as	AS468B	13665 注/Note (a) CW	242	363	1600	40.7	190

	follows: RPM, torques, accessory weight and moment as shown							
液压泵 D10* (AS907-1-1A,AS907-2-1A 和 AS907-2-1G)/Hydraulic Pump D10* (AS907-1-1A, AS907-2-1A, and AS907-2-1G)	在以下几方面 修改的 AS468B-AV1: 转速、扭矩、附 件重量和力矩/ AS468B-AV1 modified as follows: RPM, torques, accessory weight and moment as shown	AS468B	5974 注/Note (a) CW	250	375	1544	22.3	103.9
液压泵 D10* (AS907-3-1E) Hydraulic Pump D10* (AS907-3-1E)	在以下几方面 修改的 AS468B-AV1: 转速、扭矩、附 件重量和力矩/ AS468B-AV1 modified as follows: RPM, torques, accessory weight and moment as shown	AS468B	5974 注/Note (a) CW	250	388	1544	22.3	103.9

CW=顺时针/Clockwise

Tc=连续扭矩/Continuous Torque

To=扭矩过载（每 4 小时 5 分钟）/Torque Overload (5 min. per 4 hr. period)

T_S=静止扭矩/Static Torque

* 安装图纸上使用这些符号来标记附件安装座/Accessory pads are identified by these symbols on the installation drawing.

** 在快速安装/拆卸界面上/At quick attach/detach (QAD) interface

注/Notes:

(a) 驱动转速基于 100%设计 N2 转速 28100 转每分钟 /Drive speeds are based on a 100% design N2 of 28,100 rpm.

(b) 总的附件功率提取限制在发动机安装手册中作了规定（见注 12）/Total combined accessory power extraction limits are specified in the engine installation manual (see NOTE 12).

注 6/NOTE 6.

航空器用引气流量限制参见发动机安装手册（见注 12）

For aircraft bleed airflow limits, refer to the engine installation manual (see NOTE 12).

注 7/NOTE 7.

对于 AS907-1-1A, AS907-2-1A 和 AS907-2-1G 型发动机装机, 来自发动机泵的燃油被提取用来驱动飞机燃油系统(动力流)引射泵或涡轮泵。详见发动机安装手册(见注 12)

For the AS907-1-1A, AS907-2-1A, and AS907-2-1G engine aircraft installations, fuel from the engine pump is extracted to drive jet or turbine pumps in the aircraft fuel system (motive flow). Refer to the engine installation manual (see NOTE 12).

对于 AS907-3-1E 发动机, 发动机燃油泵不驱动飞机燃油系统动力流

The AS907-3-1E engine fuel pump does not drive aircraft system motive flow.

注 8/NOTE 8.

发动机满足 FAR 25 部附件 C 定义包线结冰条件下的运行要求

The engine meets FAA requirements for operation in icing conditions within the envelope defined in Title 14 CFR part 25, Appendix C.

注 9/NOTE 9.

某些发动机零件具有寿命限制, 这些限制在发动机轻维护手册的适航限制部分作了规定

Certain engine parts are life-limited. These limits are included in the engine Light Maintenance Manual, Airworthiness Limitations Section.

注 10/NOTE 10.

当出现发动机意外停车时, 正常的 5 分钟起飞时间限制可以延长至 10 分钟

The normal 5-minute takeoff time limit may be extended to 10 minutes for engine out contingency.

注 11/NOTE 11.

所有工况下的功率设定、功率检查和发动机推力输出控制基于低压转子转速 N_1 。因此, 发动机构型中包含转速传感器

Power setting, power checks and control of engine thrust output in all operations are based on N_1 . Speed sensors are included in the engine assembly for this purpose.

注 12/NOTE 12.

其它性能、批准的运行和安装信息参见 FAA 批准的下述发动机安装手册:

For additional performance, authorized operation and installation detailed information, refer to FAA approved sections of the engine installation manual as follows:

型号/Model AS907-1-1A: 24-IM-8014

型号/Model AS907-2-1A: 24-IM-8014

型号/Model AS907-2-1G: 24-IM-8029

型号/Model AS907-3-1E: 24-IM-8030

注 13/NOTE 13.

已批准发动机控制系统限时派遣(TLD)运行, 关于最大批准放行间隔和发动机控制系统维护要求的适航限制在发动机轻维护手册中作了规定

The engine control systems has been approved for Time Limited Dispatch (TLD) operations, and the airworthiness limitations pertaining to the maximum approved dispatch intervals and maintenance requirements of the engine control system are specified in the engine Light Maintenance Manual, Airworthiness Limitations Section.

注 14/NOTE 14.

推荐的发动机检查间隔参见发动机轻维护手册第 5 章

Recommended engine inspection intervals are included in the engine Light Maintenance Manual, Chapter 5.

注 15/NOTE 15.

发动机型号列表中所列的所有型号均满足 2013 年 10 月 23 日生效的 14 CFR 34 部第 34-5A 修正

案和 2012 年 7 月 18 日生效的 40 CFR 87 部所颁布的排放标准。

The following emissions standards promulgated in 14 CFR part 34, Amendment 34-5A, effective October 23, 2013, and 40 CFR part 87, effective July 18, 2012, have been complied with for all models listed in Engine Table of Models.

- 燃油排泄标准: 14 CFR 34.10(b)和 34.11, 以及修订的 40 CFR 87.10(b)和 87.11
Fuel Venting Emission Standards: 14 CFR §§34.10(b) and 34.11; in addition, 40 CFR §§87.10(b) and 87.11 as amended.
- 发烟指数 (SN) 排放标准: 14 CFR §34.21(e)(2), 以及 40 CFR §87.23(c)(1)
Smoke Number (SN) Emission Standards: 14 CFR §34.21(e)(2); addition, 40 CFR §87.23(c)(1).
- 一氧化碳 (CO) 排放标准: 14 CFR 34.21(d)(1)(ii)条款, 以及 40 CFR 87.23(c)(1)
Carbon Monoxide (CO) Emission Standards: 14 CFR §34.21(d)(1)(ii); in addition, 40 CFR §87.23(c)(1).
- 碳氢化合物 (HC) 排放标准: 14 CFR 34.21(d)(1)(i)条款, 以及 40 CFR 87.23(c)(1)
Hydrocarbons (HC) Emission Standards: 14 CFR §34.21(d)(1)(i); in addition, 40 CFR §87.23(c)(1).
- 氮氧化物(NO_x)排放标准: 14 CFR 34.23(a)(2)和(b)(1)条款, 以及 40 CFR §87.23(c)(2)和(c)(3)
Oxides of Nitrogen (NO_x) Emission Standards: 14 CFR §34.23(a)(2) and (b)(1); in addition, 40 CFR §87.23(c)(2) and (c)(3).

除满足型号合格证数据单上的审定要求外, 发动机制造商声明发动机对以下标准的符合性也已得到验证: ICAO 附件 16 第 II 卷第三版第 III 部分第 2 章 2.2.2 节有关烟数、2.3.2 节有关一氧化碳和碳氢化合物、2.3.2.d.ii 节有关氮氧化物 (也称为 CAEP/6)、2.3.2.e 节有关氮氧化物 (也称为 CAEP/8), 以及第 II 部分第 2 章有关燃油排泄的排放标准。

In addition to the FAA's finding of compliance based on the certification requirements defined in this TCDS, the engine manufacturer has declared that the ICAO emissions standards identified in Annex 16, Volume II, Third Edition, Part III, Chapter 2, Section 2.2.2 for SN, Section 2.3.2 for CO and HC, Section 2.3.2.d.ii for NO_x (also known as CAEP/6) and Section 2.3.2.e for NO_x (also known as CAEP/8) and Part II Chapter 2 for fuel venting have also been demonstrated.

注 16/NOTE 16.

模型描述-相似性、差异和特殊特性/Model description – similarities, differences, and special characteristics:

型号/Model	型号描述/Model Description
AS907-1-1A	一级风扇, 四级轴流高压压气机, 一级离心高压压气机, 环形燃烧室, 两级高压涡轮, 三级低压涡轮。 One stage fan, four stage axial flow high-pressure compressor, one stage centrifugal high-pressure compressor, annular combustor, two stage high-pressure turbine, and three stage low-pressure turbine.
AS907-2-1G	除以下方面外, 与 AS907-1-1A 相同: 提高了推力额定值、涡轮级间温度和转子速度限制值, 更改了离心压气机和高/低压涡轮导向叶片, 改进了燃烧系统, 飞机用风扇引气的规定, 采用了新的排气混合喷管系统, 更改了 FADEC 系统软件。 Same as AS907-1-1A except increased thrust ratings and ITT and rotor speed limits. Modified centrifugal compressor and high- and low-pressure turbine nozzles, improved combustion system, provisions for customer fan bleed, a new exhaust mixer nozzle system, and modified FADEC system software.
AS907-2-1A	除以下方面外, 与 AS907-2-1G 相同: 提高了最大连续推力额定值和起飞 N2 转速限制值, 可选安装界面零件,

	升级了 ECU 及其软件。 Same as AS907-2-1G except increased maximum continuous thrust rating and takeoff N2 rotor speed limit, select installation interface parts, and updated ECUs and software.
AS907-3-1E	除以下方面外，与 AS907-2-1A 相同：更改了推力额定值、涡轮级间温度瞬态时间及转子转速限制值，更改风扇进口罩，更改了风扇外涵道通道和后安装系统，不可调的第一级压气机静子叶片，更改了压气机和燃烧室机匣、增加了附件传动限制值，更改了发电机快速安装/拆卸，更改了空气涡轮起动机，更改了 FADEC 系统软件。 Same as AS907-2-1A except modified thrust ratings and ITT transient time and rotor speed limits, modified fan inlet housing, modified fan outer bypass duct and aft mounting system, non-variable stage 1 compressor vanes, modified compressor and combustor cases, increased accessory drive limits, modified generator quick attach-detach (QAD), modified air turbine starter, and modified FADEC system software.

注 17/NOTE 17.

发动机型号设计不包括反推装置（T/R）。反推装置的安装参见发动机安装手册（见注 12）。发动机与下列反推装置兼容：

The engine type design does not include a thrust reverser (T/R). Considerations for the installation of a thrust reverser are contained in the engine installation manual (see NOTE 12). The engines have demonstrated compatibility with the following thrust reversers:

型号/Model	反推装置制造商 T/R Manufacturer	反推装置件号/T/R Part number	
		左边/Left Hand	右边/Right Hand
AS907-1-1A	Aircelle（以前的 Hurel-Hispano） Aircell (formerly Hurel-Hispano)	按照反推装置安装图纸 13A025-03-0G Per T/R installation drawing 13A025-03-0G	按照反推装置安装图纸 13A026-02-0G Per T/R installation drawing 13A026-02-0G
		按照反推装置安装图纸 13A016-00-0G Per T/R installation drawing 13A016-00-0G	按照反推装置安装图纸 13A017-00-0G Per T/R installation drawing 13A017-00-0G
AS907-2-1G	Aircell	按照反推装置安装图纸 31A516-01-0G Per T/R installation drawing 31A516-01-0G	按照反推装置安装图纸 31A517-01-0G Per T/R installation drawing 31A517-01-0G
AS907-2-1A	Aircell	按照反推装置安装图纸 33A016-01-0G Per T/R installation drawing 33A016-01-0G	按照反推装置安装图纸 33A017-01-0G Per T/R installation drawing 33A017-01-0G
AS907-3-1E	Aircell	按照反推装置安装图纸 32A716-02-0G Per T/R installation drawing 32A716-02-0G	按照反推装置安装图纸 32A717-02-0G Per T/R installation drawing 32A717-02-0G

注 18/NOTE 18

删除/DELETED

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